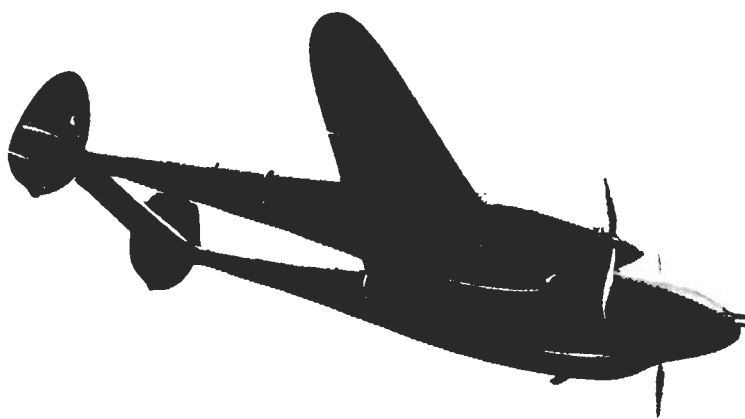


FOR OFFICIAL USE ONLY

*PILOT'S FLIGHT OPERATING
INSTRUCTIONS*
FOR
ARMY MODELS
P-38H Series, P-38J Series, P-38L-1
L-5 and F-5B AIRPLANES

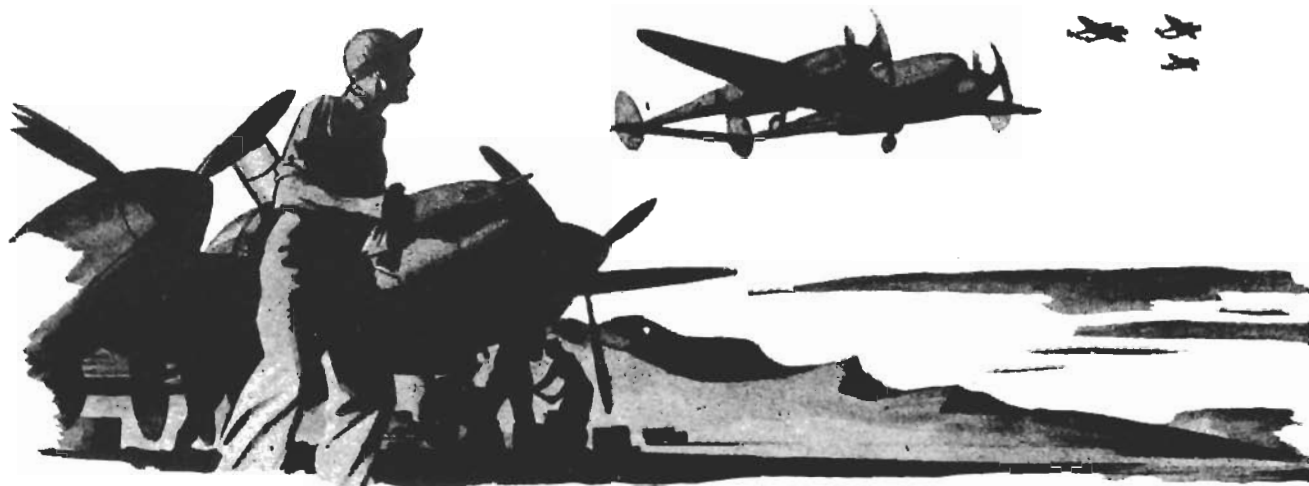


This publication contains specific instructions for pilots and should be available for Transition Flying Training as contemplated in AAF Reg. 50-16.

Appendix I of this publication shall not be carried in aircraft on combat missions or when there is a reasonable chance of its falling into the hands of the enemy.

Published under joint authority of the Commanding General, Army Air Forces, the Chief of the Bureau of Aeronautics, and the Air Council of the United Kingdom.

NOTICE: *This document contains information affecting the national defense of the United States within the meaning of the Espionage Act, 50 U. S. C., 31 and 32, as amended. Its transmission or the revelation of its contents in any manner to an unauthorized person is prohibited by law.*



Description

1. AIRPLANE.

a. GENERAL.

(1) The P-38H, P-38J, P-38L, and F-5B airplanes are twin boomed, single place monoplanes manufactured by the Lockheed Aircraft Corporation. P-38H, P-38J, and P-38L are fighter airplanes. The F-5B is a photographic airplane. Hydraulically operated landing gear, flaps, brakes, and coolant shutters are provided. (Late airplanes are equipped with hydraulically boosted aileron control.) The approximate overall dimensions are as follows:

Length	37 feet 10 inches
Height (top of prop rad.) ..	12 feet 10 inches
Span	52 feet 0 inches

(2) The armament is mounted in the nose of the fuselage, and armor protection is provided as shown in figure 2. Photographic airplanes are protected by armor the same as the fighters, but all armament is replaced by cameras.

b. FLIGHT CONTROLS.

(1) **TRIM TABS.**—Trim tabs are mounted on all the movable surfaces and are controllable from the cockpit during flight. Airplanes equipped with aileron control booster do not have aileron trim tabs.

(2) **FLAPS.**—The flaps are a Lockheed modified Fowler type.

(a) **CONTROLS.**—Flap action is controlled by the lever (figure 7-2) on the right-hand side of the cockpit. When the lever is placed to UP, DOWN, or MANEUVER, the flaps will automatically stop at the desired position. The lever should be returned to CLOSED as soon as the end position is reached. The control will not go to the DOWN position until the trigger on the lever is lifted through the notch just forward of the CLOSED position.

Note

When using maneuvering flaps, the flap lever must be left in the MANEUVER position. If it is moved even slightly forward and then returned to MANEUVER, the flaps will extend completely. (This condition is corrected on late airplanes.)

(b) **POSITION INDICATOR.**—On the early airplanes, flap position is indicated by the flap and landing gear position indicator on the instrument panel. On late airplanes the flap position is indicated by a small pop-up lever (figure 3) on the trailing edge of the left center section just inside the boom. This indicator projects above the wing whenever the flaps are not full up.

(3) **SURFACE CONTROL LOCK.**—The surface control lock (figure 7-10) is stowed on the right-hand window sill and extends across the center of the cockpit when in use. On the late airplanes this lock does not lock the rudders.

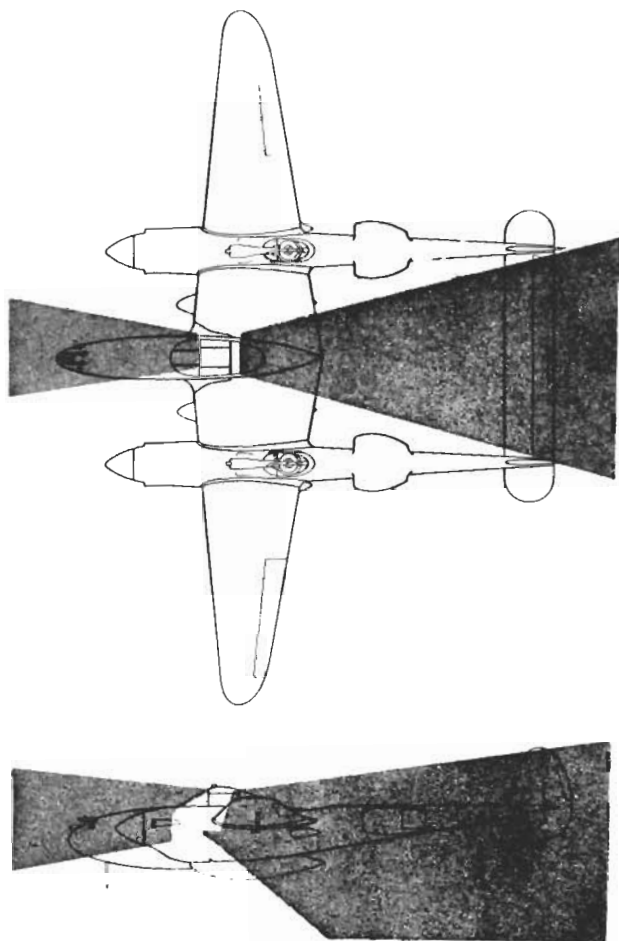


Figure 2 — Armor Protection

(4) **AUTOMATIC PILOT.**—A type A-4 automatic pilot is installed in the F-5B airplane. The automatic pilot control unit is on the instrument panel; the shut-off valve (figure 24-10) is below the instrument panel; and the automatic pilot oil pressure gage (figure 33-10) is on the center control stand.

(5) **DIVE RECOVERY FLAP.**—P-38L and late P-38J airplanes are equipped with electrically operated dive recovery flaps which are controlled by a switch (figure 9-2) on the pilot's control wheel. The dive recovery flaps will extend or retract within two seconds.

(6) **AILERON CONTROL HYDRAULIC BOOSTER.**—P-38L and late P-38J airplanes are equipped with hydraulically boosted aileron control. On

these airplanes most of the aileron control force is provided by hydraulic boost; the remainder is applied by the pilot. Figure 15 illustrates the control linkage. Control cables which control the boost mechanism are mechanically connected to the control surfaces, allowing manual flight control in an emergency. The aileron boost shut-off valve (figure 9-5), is located on the right side of the cockpit near the pilot's control column. In addition to this valve an automatic by-pass valve is incorporated in the mechanism to allow free movement of the ailerons in case the hydraulic pressure should fail.

IMPORTANT

The ailerons ride approximately one inch higher with boost OFF. This change in trim is normal.

c. LANDING GEAR. (See figure 16.)

(1) The landing gear lever (figures 4-32) controls the extension and retraction of all three wheels. A lock on the lever prevents the lever from being moved out of the DOWN position when the airplane is on the ground (when the left main shock strut is compressed). If this lock fails, or if it is necessary to retract the landing gear on the ground because of engine failure at take-off, the lock may be released by rotating the landing gear control release-knob (figure 4-33) in a counter-clockwise direction.

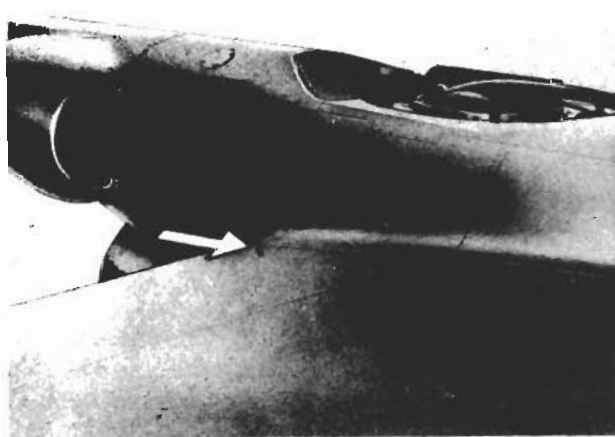
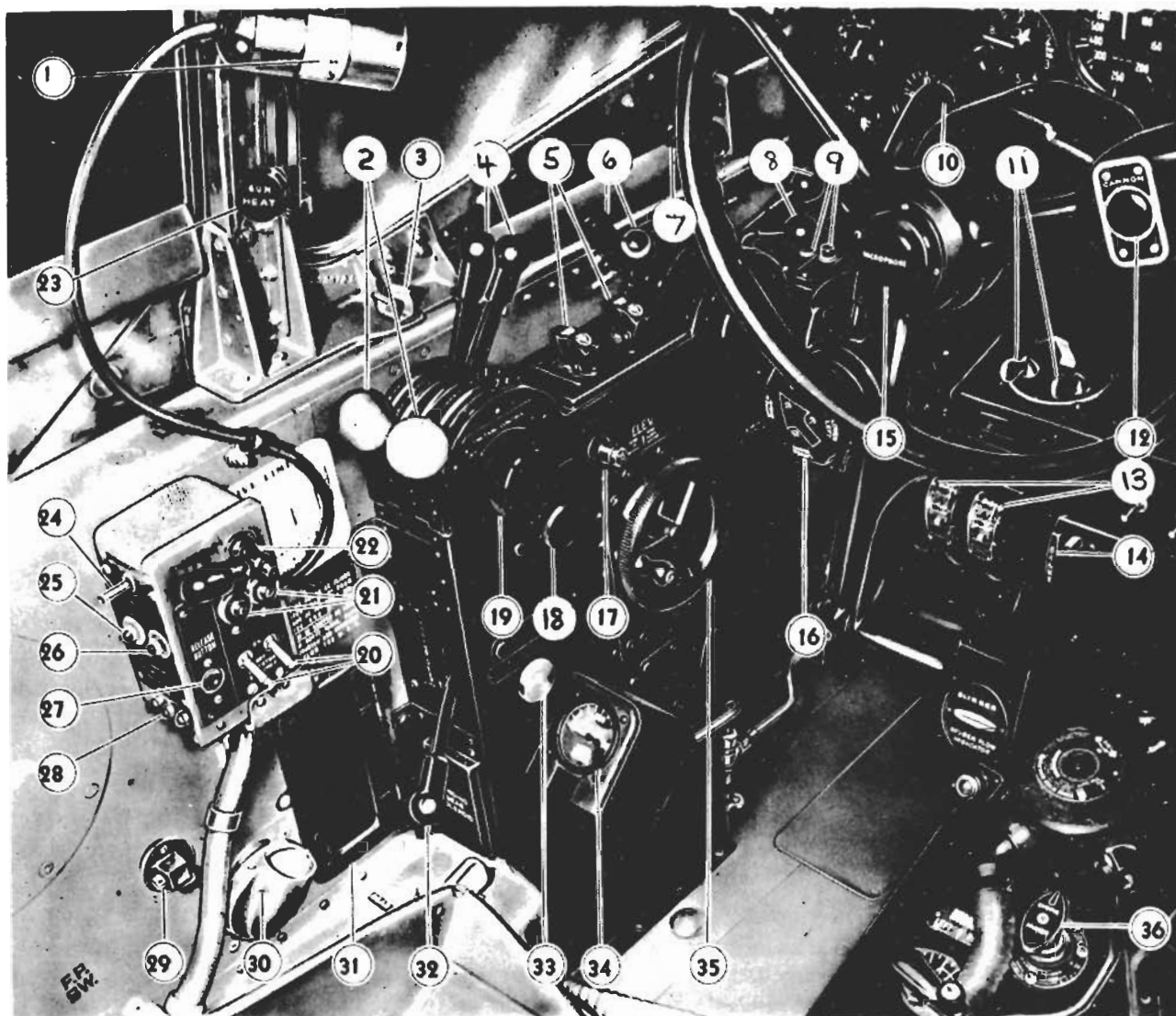


Figure 3 — Flap Pop-up Indicator



1. Spotlight (normal position).
2. Throttles.
3. Surface controls lock 'clip.
4. Propeller controls.
5. Propeller selector switches.
6. Mixture controls.
7. Propeller warning lights (P-38H only).
8. Carburetor air filter control. (Late airplanes.) (Control not used on early airplanes.)
9. Propeller circuit breaker buttons.
10. Gun charger handle (on early airplanes only).
11. Ignition switches.
12. Cannon trigger button. (Machine gun button on forward side of wheel.) (Switch arrangements vary with different airplane models.)
13. Propeller feathering switches.
14. Parking brake handle.
15. Microphone button. (Location varies with airplane model.)
16. Gun charging selector knob (P-38H only).
17. Landing gear warning light. (Early airplanes only.)
18. Propeller lever vernier knob.
19. Friction control.
20. Bomb or tank release selector switches.
21. Bomb or tank release indicator lights.
22. Cockpit light.
23. Gun (or camera) compartment heat control. (Cockpit heat on later airplanes.)
24. Arm-safe switch. (Bombs.)
25. Arming indicator light.
26. Safe indicator light.
27. Bomb or droppable tank release button.
28. Spare indicator lights.
29. Spotlight alternate position socket.
30. Cockpit ventilator control.
31. Gun sight dark glass stowage. (Early airplanes only.)
32. Landing gear control handle.
33. Landing gear control release.
34. Oxygen pressure gage.
35. Elevator tab control.
36. Engine primer.

Figure 4 — Cockpit — Left-hand Side

(2) POSITION INDICATOR.

On P-38H, early P-38J and F-5B airplanes, the landing gear position is indicated on the instrument panel by the flap and landing gear position indicator. A light on the control stand, and a warning horn operate when either throttle is closed if the landing gear is not locked down. On later P-38J and F-5B, and all P-38L airplanes, the position indicator and warning light on the engine control stand have been replaced by a warning light (figure 8-24) on the instrument panel. The warning light glows whenever the landing gear is in transit - not locked up or down. With either throttle closed the warning light glows unless the landing gear is down and locked.

(3) BRAKES.—The brake system is not connected to the main hydraulic system. No emergency braking system is provided. See figure 18 for brake system diagram.

d. HYDRAULIC SYSTEM. (See figures 14 and 16.)

(1) Normal system pressure (figure 8-22) is between 1100 to 1400 psi, surge to 1600 psi permissible. See Section IV for emergency operating instructions.

(2) There are three separate systems of operation for the hydraulic equipment in this airplane.

(a) The normal system operates all the hydraulic equipment (except brakes) using hydraulic pressure from the engine driven hydraulic pumps and hydraulic fluid from the top half of the main hydraulic fluid reservoir.

(b) The auxiliary system operates the same equipment and uses the same lines as the normal system except that the hand hydraulic pump furnishes the hydraulic pressure and the fluid comes from the bottom of the main hydraulic fluid reservoir. When the auxiliary system is in use the hand pump source selector valve (figure 17-4) is UP and the bypass valve (figure 17-3) is OPEN.

Note

It will be impossible to build up pressure with the hand pump unless the aileron boost valve is OFF and the coolant override switches are OFF. The coolant override valves have a fixed bleed in the system when the switches are in the override position.

(c) The emergency system is used to extend the landing gear in case of complete failure of the normal and auxiliary systems. The emergency system is equipped

with a separate reservoir and separate lines. Pressure is supplied by the hand pump. When the emergency system is in use the hand pump source selector valve is DOWN and the bypass valve is CLOSED. (Late airplanes have the bypass valve incorporated in the source selector valve.)

e. ELECTRICAL SYSTEM.

(1) GENERAL.—On P-38H and early P-38J airplanes, the 24 volt electrical system is powered by a generator on the left engine and by a battery. F-5B, P-38L, and late P-38J airplanes have a generator on each engine and a battery. On fighter airplanes the battery is in the left boom, and on photographic airplanes the battery is in the nose compartment. The battery switch (figure 5-15) cuts out the battery, leaving the rest of the system operating on the generator(s). The generator switch (figure 5-16) (two switches on late airplanes, figure 8-11) turns off the power from the generator, allowing the system to draw power from the battery only. The ignition master switch (figure 5-1) is not connected to the airplane's electrical system, and turns off the ignition only to both engines.

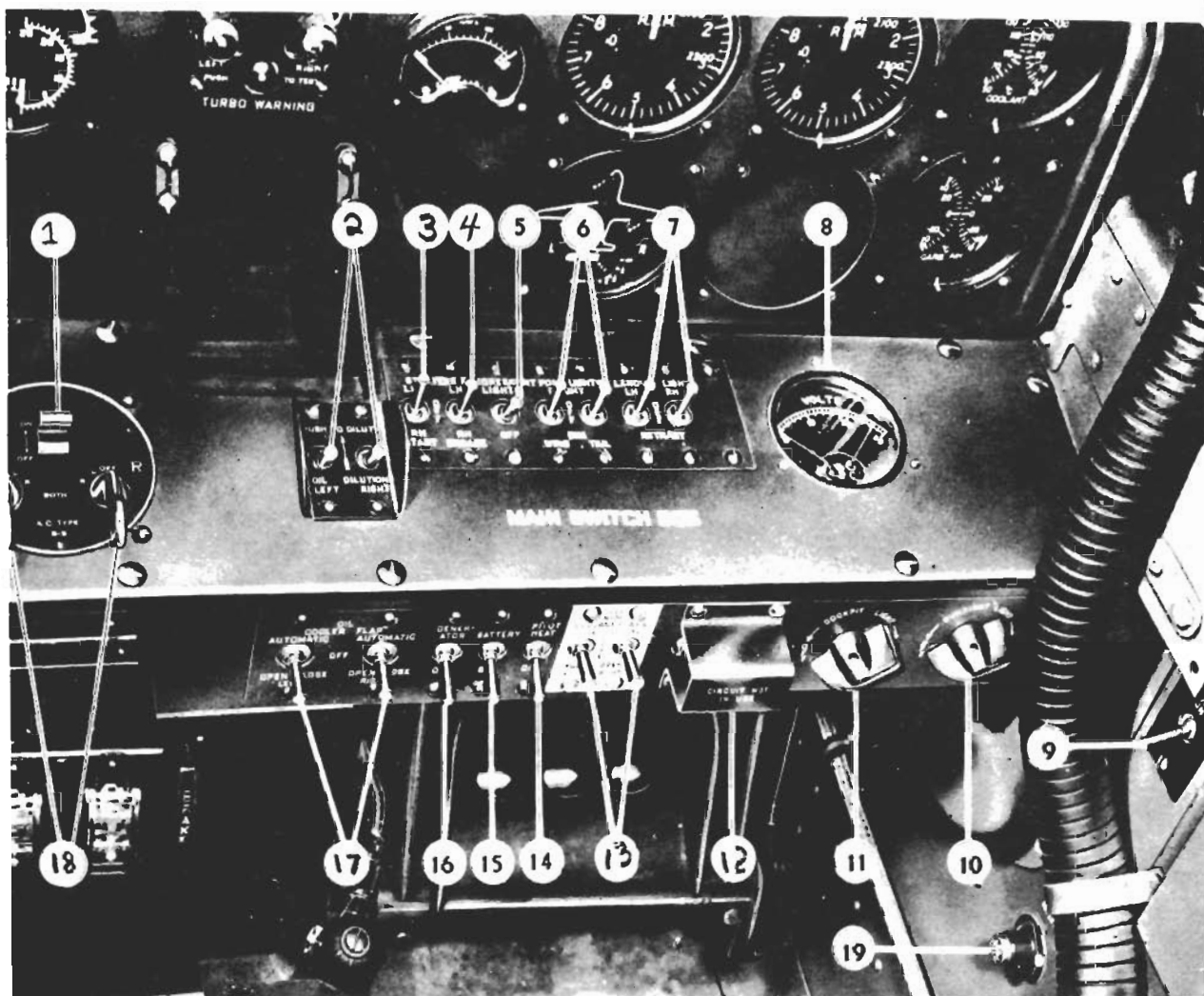
(2) CIRCUIT BREAKERS AND FUSES.—On late airplanes, circuit breakers are used to replace the fuses previously located in the nose gear wheel well. These circuit breakers are mounted along the right forward side of the cockpit (figure 6) and act as fuses to automatically break the circuit whenever an overload occurs. They may be reset by allowing a short interval for cooling and then pushing the button.

Note

It is not possible to tell by visual inspection whether or not a circuit breaker is open.

(3) LIGHTS.

(a) LANDING LIGHTS.—The landing light is located under the left wing and controlled by a switch (figure 5-7) on the main switch box. P-38H airplanes have a landing light under each wing, with individual switches. With the switch ON the light turns on and extends. On P-38H airplanes, when the switches are turned OFF the lights turn off, but remain extended. On F-5B and early P-38J airplanes when the switch is turned OFF the lights remain ON and extended. Later P-38J and P-38L airplanes have the landing light recessed in the left wing leading edge. The landing lights should not be used unnecessarily, as the life of the lamps is approximately 25 hours.



1. Ignition master switch.
2. Oil dilution switches
3. Starter switch.
4. Engage switch.
5. Flourescent light switch.
6. Position light switches.
7. Landing light switches (left-hand only on P-38J, P-38L, and F-5B).
8. Voltmeter.
9. Inverter switch (P-38H).
10. Gun sight light rheostat (on control column on later airplanes).
11. Cockpit light rheostat.
12. Intercooler flap switches. (P-38J, P-38L, and F-5B. Circuit not in use on P-38H.)
13. Coolant flap override switches.
14. Pitot heat switch.
15. Battery switch.
16. Generator switch.
17. Oil cooler flap switches.
18. Ignition switches.
19. Inverter warning light (P-38H).

Figure 5 — Main Switch Box



- | | |
|--|--------------------------------------|
| 1. Fluorescent light rheostat (late airplanes). | 14. SCR-695 radio. |
| 2. Landing light. | 15. Range receiver. |
| 3. Starter and booster. | 16. SCR-274N and SCR-522A Receivers. |
| 4. Cross suction and oil dilution. | 17. Radio dynamotor. |
| 5. R.H. fuel pump. | 18. Fuel gages. |
| 6. L.H. fuel pump. | 19. Temperature instruments. |
| 7. Cockpit lights. | 20. Oxygen and landing gear warning. |
| 8. Position lights. | 21. Coolant override. |
| 9. Recognition lights. | 22. Right intercooler flap motor. |
| 10. Bomb and droppable tank release. | 23. Left intercooler flap motor. |
| 11. Camera and armament firing and solenoids and relays. | 24. Right oil cooler flap motor. |
| 12. Camera and armament firing and solenoids and relays. | 25. Left oil cooler flap motor. |
| 13. Contactor heater switch. | |

Figure 6 — Circuit Breakers

(b) **RECOGNITION LIGHTS.**—These airplanes have three downward lights, red, green, and amber. White lights are on some airplanes.

CAUTION

It is possible to burn the plastic lenses of the downward recognition lights by operating them for more than thirty seconds while on the ground.

(c) **POSITION LIGHTS.**—The position lights are controlled by switches (figure 5-6) on the main switch box.

(d) **COCKPIT LIGHTS.** (Figures 7-12 and 4-22).—These lights are controlled by a rheostat (figure 5-11) on the main switch box and by a switch on the lights themselves.

(e) **FLUORESCENT INSTRUMENT LIGHTS.**—The fluorescent lights are mounted on the forward side of the pilot's control column and are turned ON by a switch (figure 5-5) on the main switch box. Light intensity is regulated by twisting the ends of the lighting unit.

(f) **SPOTLIGHT.** (Figure 4-1).—The spotlight is normally on the left windshield support. An alternate position (figure 4-29) is provided over the fuel tank selector valves. The spotlight switch is located on the light. The beam may be focused by sliding the screw head forward and aft in its slot.

(4) **INVERTER.**—On P-38H airplanes the inverter is turned on by the switch (figure 5-9) under the flap control lever. It supplies alternating current to operate the fluorescent lights and the remote indicating compass. A warning light (figure 5-19) glows when the inverter is not operating, indicating failure of the inverter and consequent failure of the compass and fluorescent lights.

(a) On P-38J, P-38L, and F-5B airplanes the inverter operates only the compass and is turned on by a switch (labeled COMPASS) on the main switch box. No warning light is installed.

f. PILOT COMFORT.

(1) The rudder pedals are adjusted by pushing or pulling the small lever (figure 7-18) on the outboard corner of each pedal and moving the pedals to suit. Care must be taken to insure that both pedals are adjusted equally.

(2) The seat is adjusted by lifting the small lever (figure 17-1) on the right side of the seat and raising or lowering the seat as required. After the lever is released, check to make sure the seat is firmly locked in the new position.

(3) Shoulder harness should be worn at all times. It will be impossible to lean forward unless the harness lever (figure 13-7) on the left side of the pilot's seat is raised. The harness lock will re-engage as soon as an upright position is resumed.

(4) **PILOT'S RELIEF TUBE.** (Figure 7-16.)

2. POWER PLANT.

a. **ENGINES.**—P-38H, P-38J, and F-5B airplanes are powered by one V-1710-89 right-hand rotating engine and one V-1710-91 left-hand rotating engine. P-38L airplanes are powered by one V-1710-111 right-hand rotating engine and one V-1710-113 left-hand rotating engine. These are 12-cylinder V-type liquid-cooled Allison engines. The engines drive three-bladed, constant-speed, full-feathering Curtiss electric propellers.

b. FUEL, OIL, AND COOLANT SPECIFICATIONS.

Fuel.....Specification AN-7-28, Grade 100/150 or Specification AN-7-26, Grade 91.

Oil.....Specification AN-VV-O-446, Grade 1120 (for cold weather operation, Grade 1100A).

Coolant..Specification AN-E-2 (Ethylene Glycol—Inhibited with NaMBT).

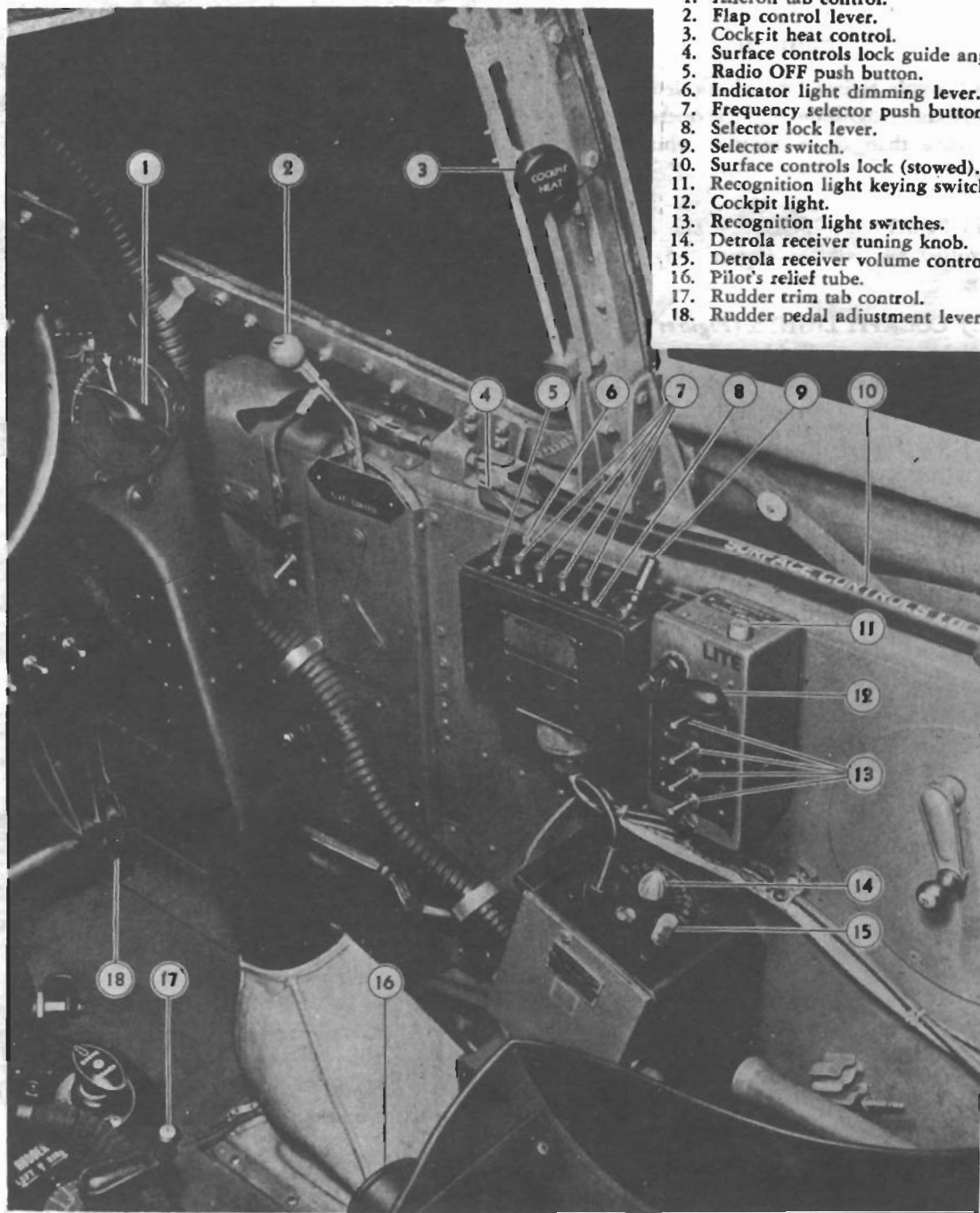
c. FUEL SYSTEM. (See figures 10, 10A, and 11.)

A separate fuel system is provided for each engine. Fuel may also be cross-fed to the opposite engine during single engine operation or when operating on the droppable tanks.

The booster pump switches and the tank selector valves are on the left beside the pilot's seat. On airplanes equipped with outer wing tanks, the outer wing tank low level warning lights are in the forward left-hand corner of the cockpit. On unmodified airplanes the engine priming pump is on the floor in front of the seat. On modified airplanes the priming-oil dilution switch is on the main switch box.

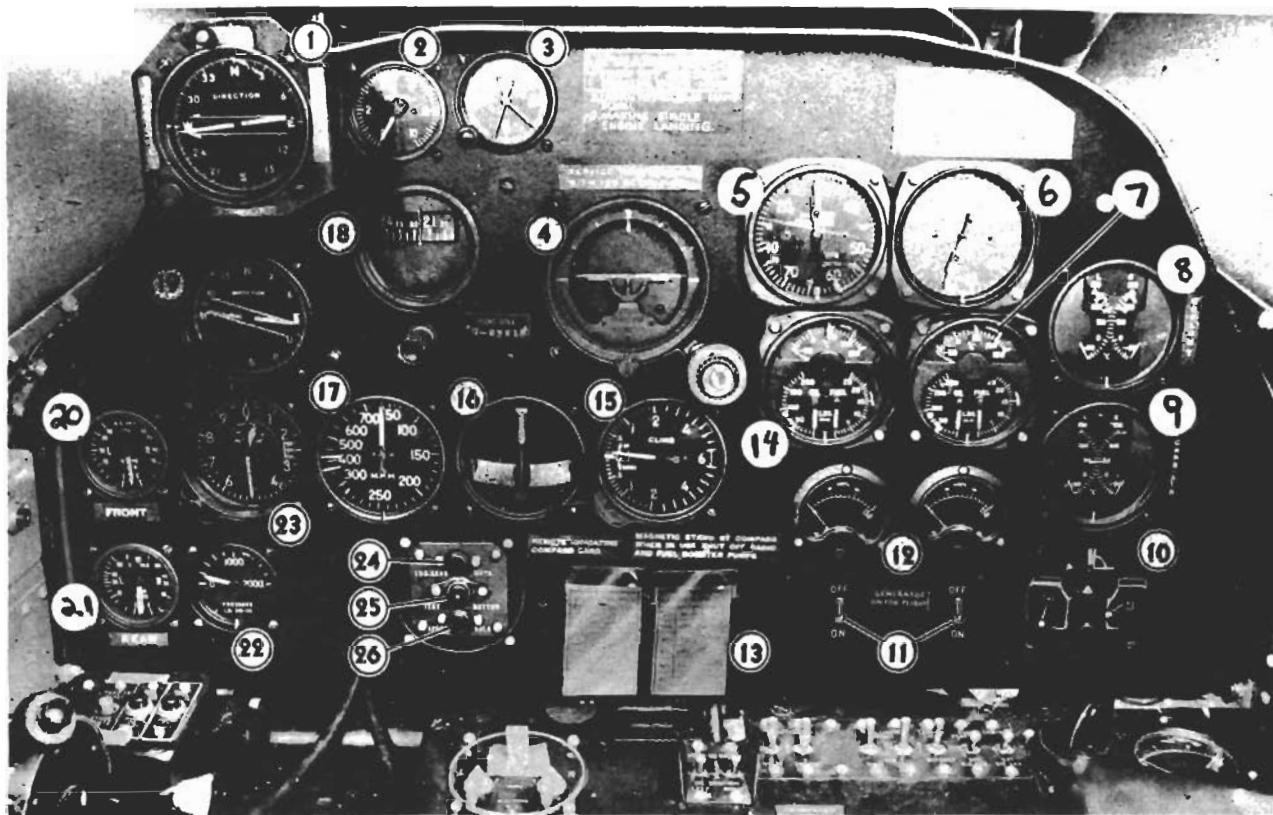
Note

All P-38L-5 airplanes and some P-38L-1 and P-38J airplanes are equipped with a modified fuel system. The new system may be readily identified by the two booster pump master switches (figure 13A-4) and the two speed control switches (figure 13A-3). All airplanes equipped with the old system will be referred to as unmodified airplanes and airplanes equipped with the new system will be referred to as modified airplanes.



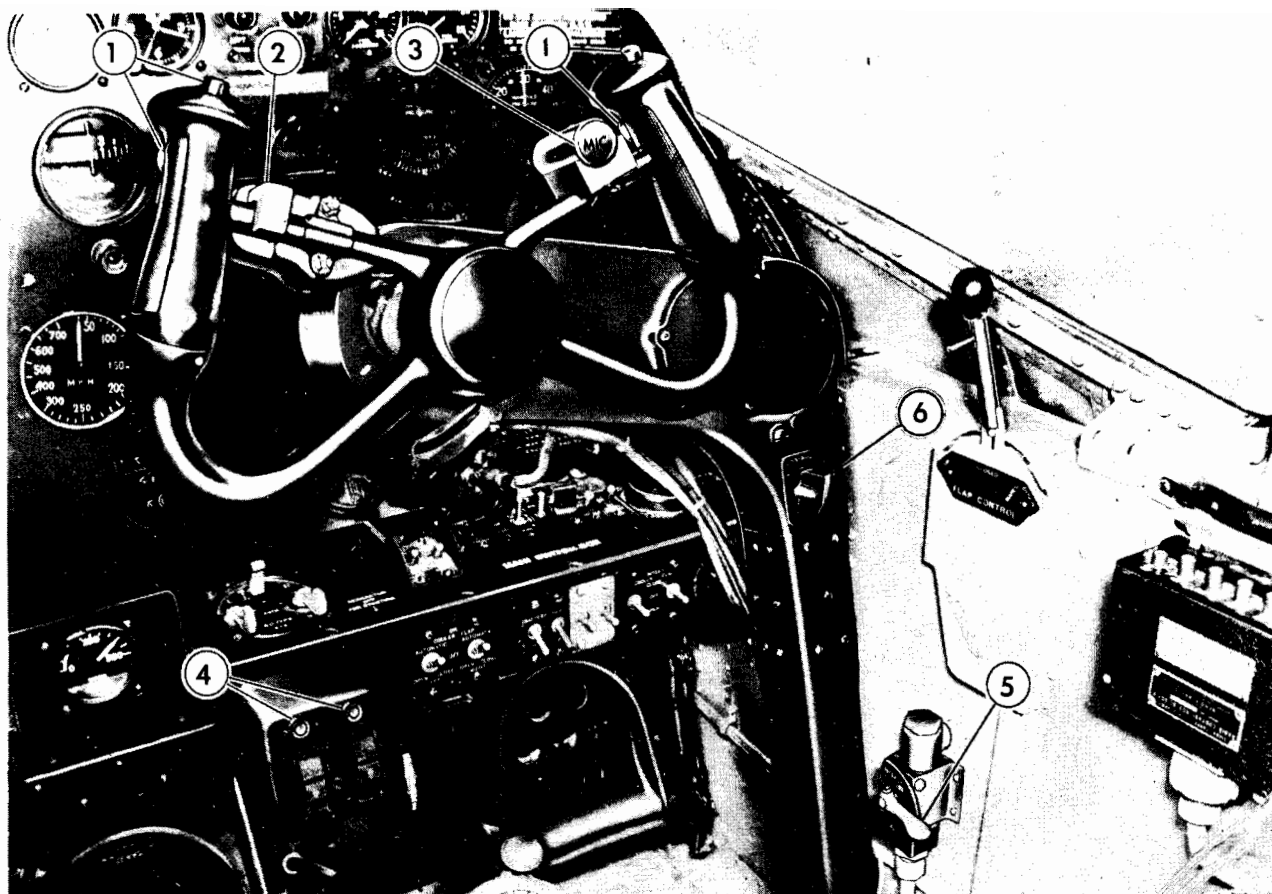
1. Aileron tab control.
2. Flap control lever.
3. Cockpit heat control.
4. Surface controls lock guide angle.
5. Radio OFF push button.
6. Indicator light dimming lever.
7. Frequency selector push buttons.
8. Selector lock lever.
9. Selector switch.
10. Surface controls lock (stowed).
11. Recognition light keying switch.
12. Cockpit light.
13. Recognition light switches.
14. Detrola receiver tuning knob.
15. Detrola receiver volume control.
16. Pilot's relief tube.
17. Rudder trim tab control.
18. Rudder pedal adjustment lever.

Figure 7 — Cockpit — Right-hand Side



1. Standby magnetic compass.
2. Suction gage.
3. Clock.
4. Gyro Horizon.
5. Manifold pressure gages (left and right).
6. Tachometers (left and right).
7. Engine gage right engine (oil temperature and pressure and fuel pressure).
8. Coolant temperature gage.
9. Carburetor air temperature gage.
10. BC-608 contactor.
11. Generator switches.
12. Ammeters.
13. Compass correction cards.
14. Engine gage left engine (oil temperature and pressure and fuel pressure).
15. Rate of climb indicator.
16. Bank and turn indicator.
17. Airspeed indicator.
18. Directional gyro.
19. Remote indicating compass.
20. Front (reserve) fuel tanks quantity gage.
21. Rear (main) fuel tanks quantity gage.
22. Hydraulic pressure gage.
23. Altimeter.
24. Landing gear warning light.
25. Landing gear warning light test button.
26. Spare bulb.

Figure 8 — Typical Instrument Panel (P-38J-25 Panel Shown)



1. Control wheel switches (cannon and machine gun trigger, bomb release, etc.).
2. Dive recovery flap control switch.
3. Microphone switch.

4. Propeller feathering switch warning lights.
5. Aileron boost shut-off valve (late airplanes).
6. Gun sight light rheostat (late airplanes).

Figure 9 — Control Wheel

d. OIL SYSTEM. (See figure 19.)

(1) **OIL DILUTION.**—The oil dilution system is controlled by two switches on the main switch box (figure 5-2).

(2) **TEMPERATURE CONTROL.**—Oil temperature control is automatic when the switches (figure 5-17) are set to automatic. In case of failure of the automatic system, manual control over the oil cooler flaps may be obtained by means of the same switches.

e. THROTTLE CONTROL.—The throttle is mechanically connected to the supercharger regulator so that control of the supercharger as a separate operation has been eliminated. Early airplanes are equipped with turbo-supercharger overspeed warning lights which flicker when rated turbo speed is reached and glow steadily when war emergency speed is reached. Late airplanes are equipped with turbo-supercharger regulators incorporating two overspeed controls. The normal overspeed

control regulates to 24,000 turbo rpm. The War Emergency overspeed control regulates to 26,400 turbo rpm and comes into operation whenever the throttles are advanced past the take-off stops. The overspeed warning lights are deleted from these airplanes. These late airplanes are also equipped with manifold pressure regulators. With these two regulators, each manifold pressure has a corresponding cockpit throttle position which is fixed for all altitudes up to the critical altitude. (Critical altitude is the altitude where the overspeed governor takes control). Above critical altitude, the turbo rpm is held constant at either 24,000 or 26,400, depending on the throttle position, and the manifold pressure will automatically drop approximately $1\frac{1}{2}$ " Hg. for every 1000 feet above critical altitude.

Thus above critical altitude, if it is desired to reduce the manifold pressure, no reduction will be obtained until the throttle is pulled back to the position which would normally give the manifold pressure then being

allowed by the overspeed control. At manifold pressures below that being allowed, the power plant is operating below its critical altitude and throttle operation is normal. If it is desired to increase the manifold pressure, no change will occur until the throttle is past the take-off stop. At this point the War Emergency overspeed control comes into operation and the supercharger speed increases to 26,400 rpm with a corresponding rapid increase in manifold pressure. This increase is approximately 5" Hg. and no intermediate settings can be made.

f. MIXTURE CONTROLS.—The mixture controls (figure 4-6) have four positions, FULL RICH, (Wired off in some planes) RICH, AUTO LEAN, and CUT-OFF.

g. PROPELLER CONTROLS.

(1) PROPELLER CONTROL LEVERS.—

These levers (figure 4-4) select the desired engine rpm for automatic constant speed propeller operation.

(2) PROPELLER SELECTOR SWITCHES (figure 4-5). These switches have four positions:

(a) AUTO CONSTANT SPEED. — The propeller governors are in operation and engine speed will be maintained as set on the propeller levers.

(b) FIXED PITCH. — Propeller pitch is fixed, engine speed depends upon power and airplane speed.

(c) INC RPM.—Engine speed increases with lower pitch,

(d) DEC RPM.—Engine speed decreases with higher pitch.

(3) PROPELLER CIRCUIT BREAKERS.—These circuit breakers open, and the propeller pitch changing mechanism becomes inoperative, when the current required to operate the propellers becomes too high. When the circuit breakers open, the buttons (figure 4-9) pop up, disclosing a red and white band on the buttons. The circuit breakers may be reset by pushing the buttons after allowing approximately 15 seconds for the switches to cool. Only the black portion of the buttons is visible when the circuit breakers are properly set.

(4) FEATHERING SWITCHES. (Figure 4-13.)—These switches turn the propellers to their minimum drag position. Late airplanes are equipped with indicator lights which glow over the proper feathering switch when either a hard left or hard right rudder

control is applied to overcome the yaw caused by failure of one engine at low air speed.

(5) WARNING LIGHTS. (Figure 4-7.) — Propeller warning lights are installed on P-38H airplanes only. They indicate when the propeller circuits are not properly set for take-off and landing, that is, when the circuit breakers are open, or the selector switches are not set to AUTO CONSTANT SPEED. These lights, however, do not warn of an improperly set propeller control lever

(6) VERNIER KNOB. (Figure 4-18.)—The vernier knob provides for fine adjustment of the right-hand propeller pitch control when synchronizing the engines.

(7) FRICTION CONTROL. (Figure 4-19.)—The friction control may be adjusted to prevent the throttles and propeller controls from creeping out of their set position.

b. CARBURETOR AIR TEMPERATURE CONTROLS.

(1) P-38H airplanes are not provided with a control for the carburetor air temperature, as the intercoolers are located in the wings and there are no intercooler flaps. On P-38J, P-38L, and F-5B airplanes a core type intercooler is located under each engine, and the intercooler flap, controlling carburetor air temperature, is operated by the switches (figure 5-12) on the face of the main switch box.

i. CARBURETOR AIR FILTER CONTROL.

(1) This control is located behind the pilot's seat on early airplanes. On late airplanes, this control is located on the engine control stand (figure 4-8).

j. COOLANT TEMPERATURE CONTROLS.

(1) Coolant temperature is automatically regulated between 101°C (214°F) (flaps closed) and 121°C (250°F) (flaps open) with the coolant flap override switches set to OFF.

(2) The override switches (figure 5-13) may be operated to fully OPEN or fully CLOSE the flaps in the event that the regulators fail to maintain the above temperatures. It is not possible to set the coolant flaps to any position except full OPEN or full CLOSE when using the override switches. If hydraulic pressure fails completely, the flaps will assume a faired (mid) position.

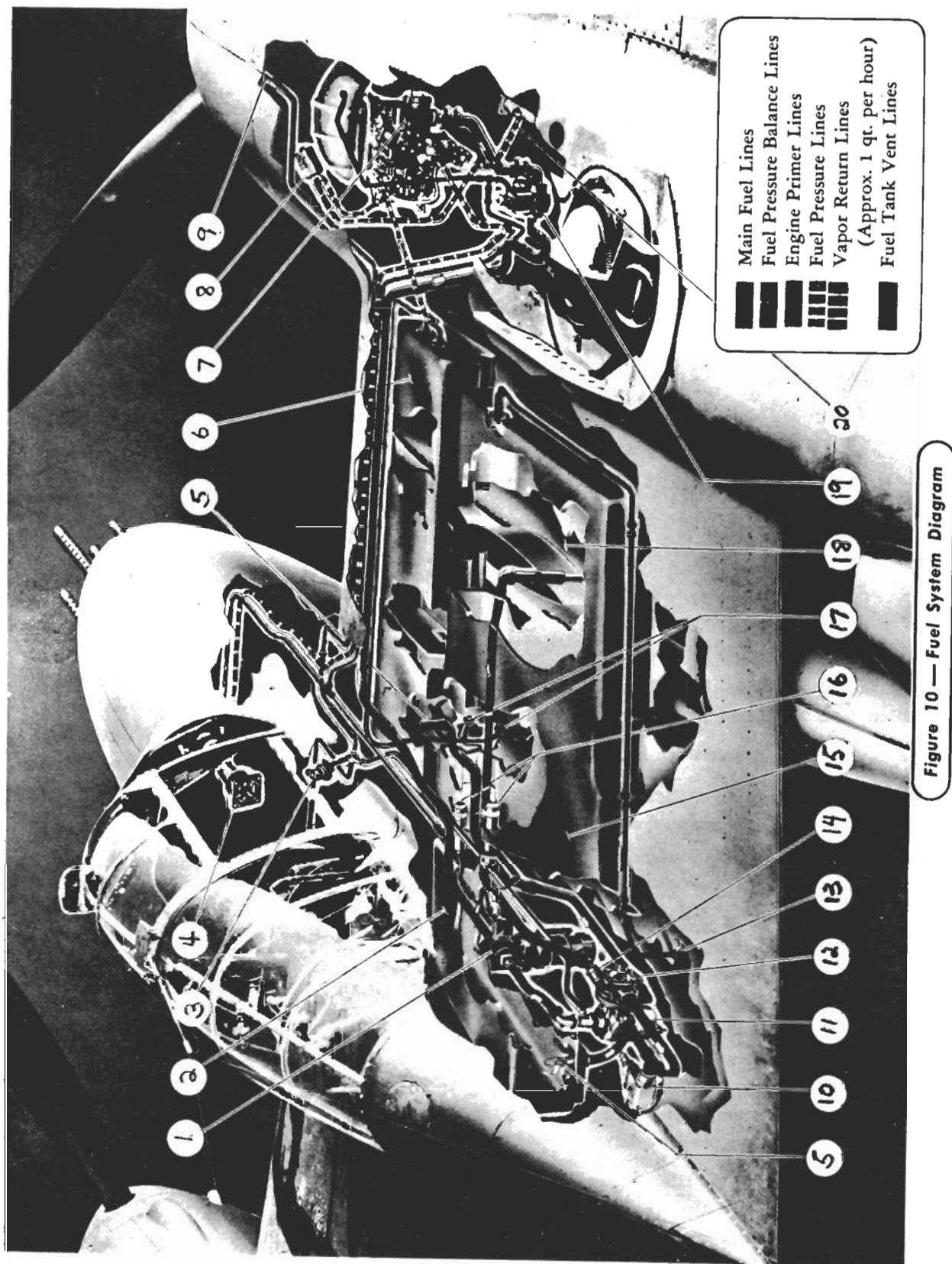


Figure 10 — Fuel System Diagram

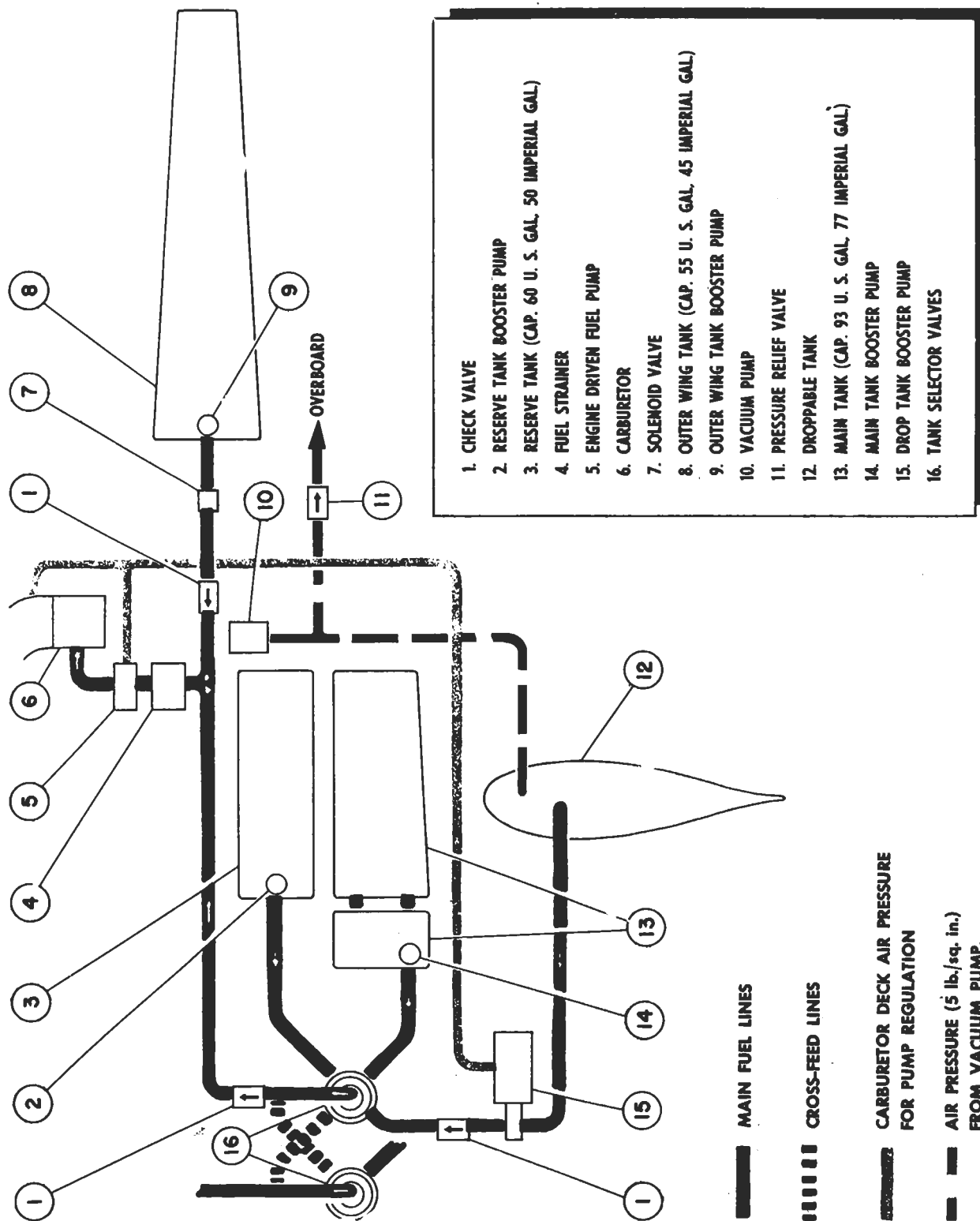
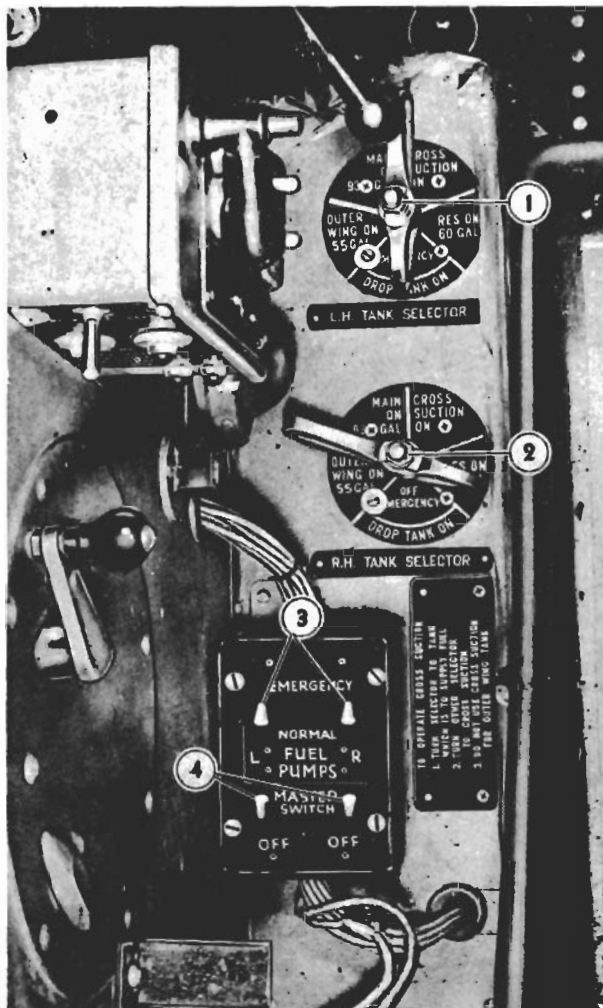


Figure 10A — Simplified Fuel System Diagram (Modified Airplanes)



1. Left-hand tank selector valve.
2. Right-hand tank selector valve.
3. Booster pump speed control switches.
4. Booster pump master switches.

Figure 13A — Fuel System Controls (Modified Airplanes)

1. Electric fuel pump.
2. Surge tank (considered part of main tank).
3. Engine priming pump.
4. Fuel pressure gage.
5. Outlet from tank.
6. Reserve tank (capacity 60 U.S. gal., 50 Imperial gal.).
7. Carburetor.
8. Oil dilution valve.
9. Line to engine primer distributor.
10. Crossfeed valve.
11. Fuel filter.
12. Tank selector valve.
13. Main tank vent.
14. Check valve.
15. Main tank (capacity 93 U.S. gal., 77 Imperial gal.).
16. Passages between main tank and surge tank.
17. Reserve tank vent.
18. Droppable tank (capacity 165 U.S. gal., 137 Imperial gal.).
Or (300 U.S. gal., 250 Imperial gal.).
19. Engine driven fuel pump.
20. Oil dilution line (injects fuel into oil system).
21. Outer wing tank (capacity 55 U.S. gal., 45 Imperial gal.).
(Not shown in figure 10.)

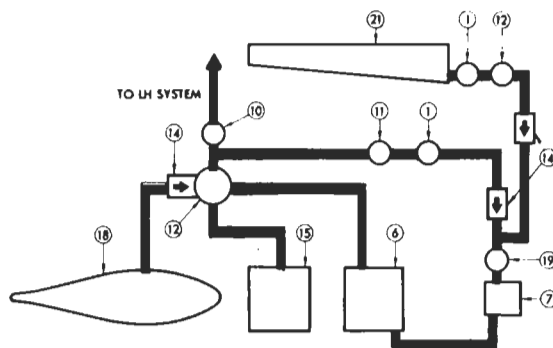
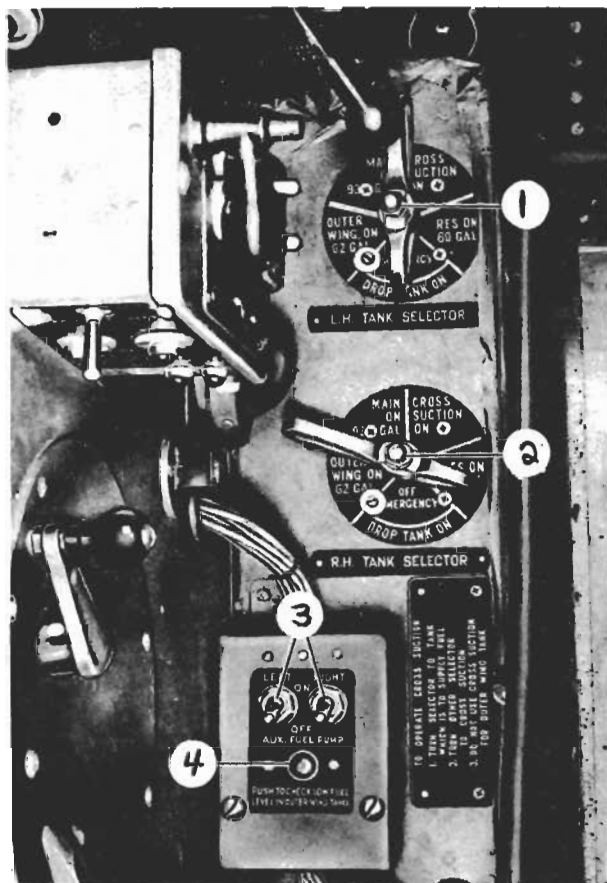
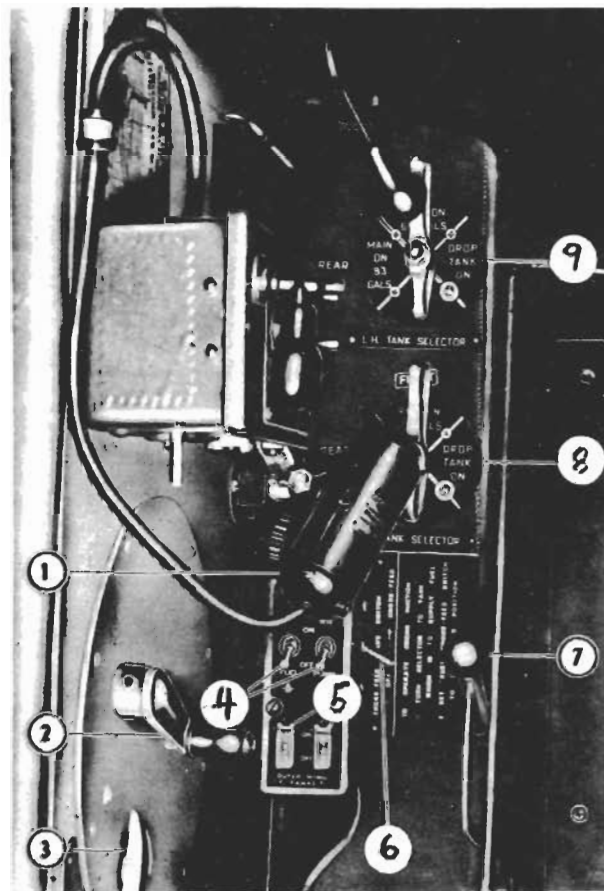


Figure 11 — Simplified Fuel System Diagram

Key to Figures 10 and 11



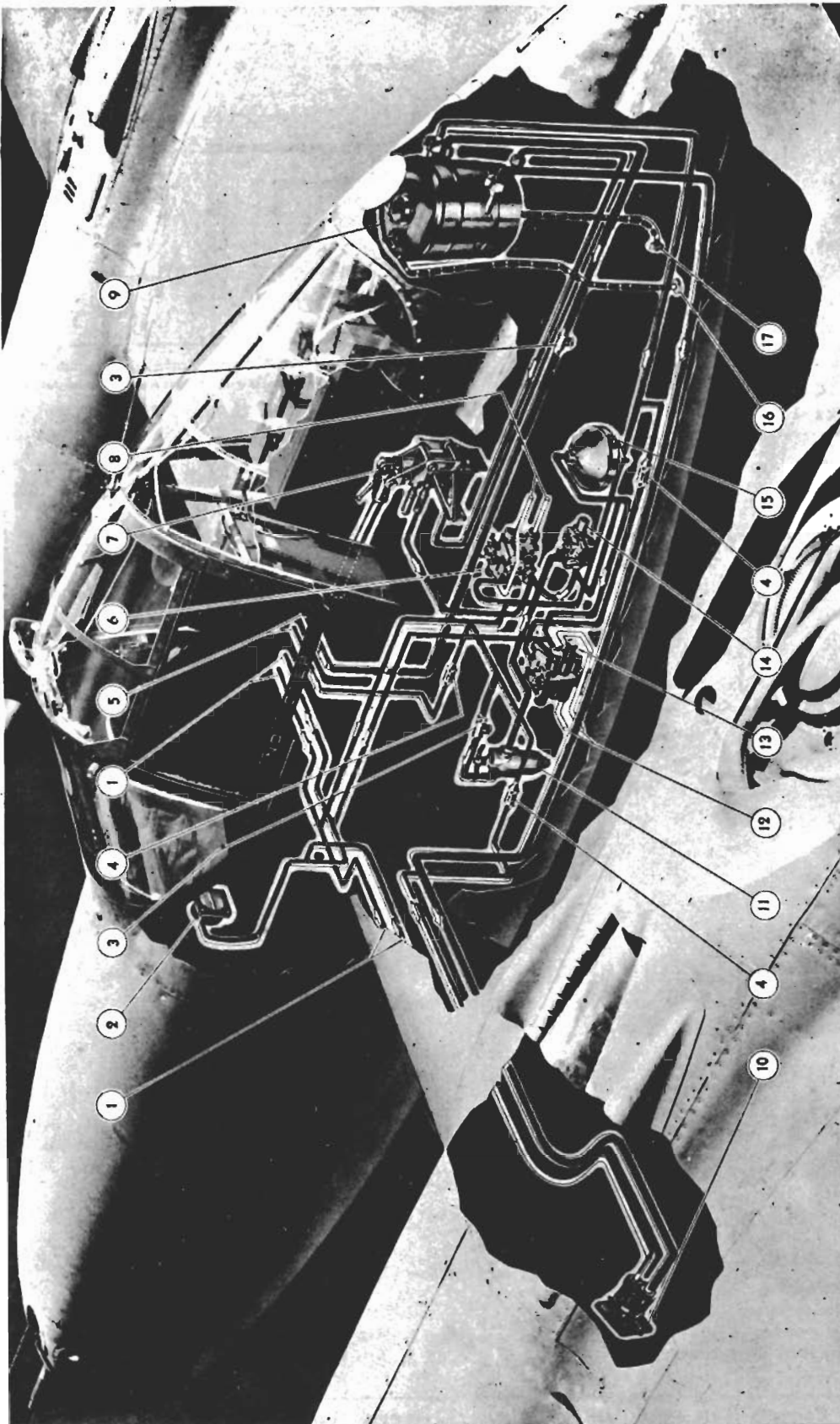
1. Left-hand tank selector valve.
2. Right-hand tank selector valve.
3. Auxiliary fuel pump switches.
4. Outer wing tank low level check button.








1. Spotlight (in alternate position).
2. Left window crank.
3. Window crank ratchet handle.
4. Electric fuel pump switches.
5. Outer wing tank switches.
6. Crossfeed switch.
7. Shoulder harness release.
8. Right-hand tank selector valve.
9. Left-hand tank selector valve.

Figure 12 — Fuel System Controls (Late Airplanes)

Figure 13 — Fuel System Controls (Early Airplanes)



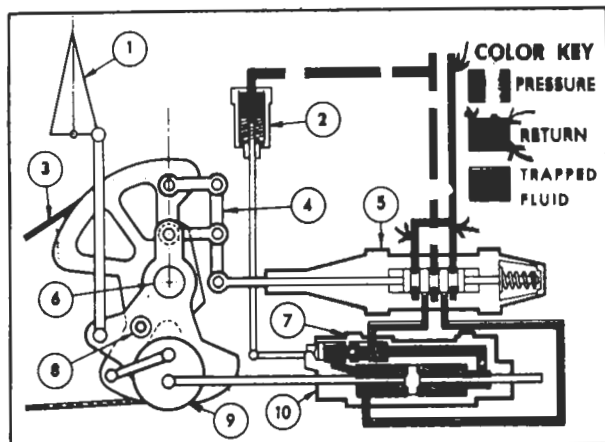
-  Pump Pressure
-  Return to Reservoir
-  Pump Suction
-  System Pressure
-  Tank Drain or Vent

1. To coolant flaps (see figure 20). (2)
2. Hydraulic pressure gage.
3. Ground test connections. (2)
4. Check valves. (3)
5. To right-hand engine pump.
6. Flap control valve.

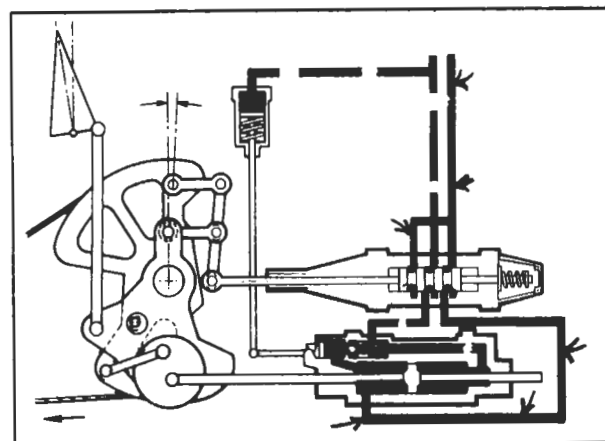
7. Emergency hand pump.
8. To flap system (see figure 16).
9. Main hydraulic reservoir.
10. Engine driven pump.
11. Hydraulic fluid filter.
12. To landing gear (see figure 16).

13. Landing gear control valve.
14. System pressure regulator.
15. Hydraulic pressure accumulator.
16. Vent to atmosphere.
17. Main reservoir drain.

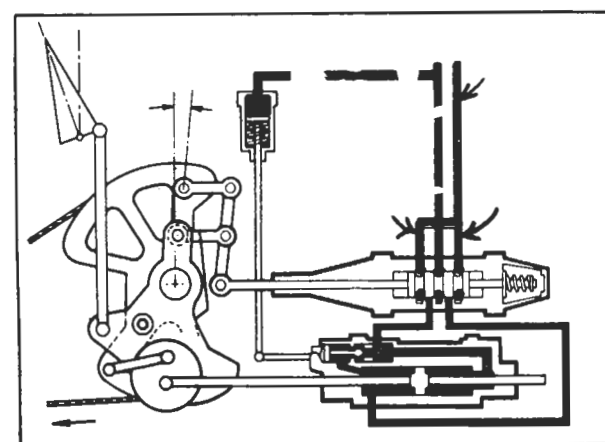
Figure 14 — Basic Hydraulic System Diagram



Aileron Neutral



Aileron Moving



Aileron Movement Completed

1. Aileron.
2. Low pressure by-pass actuator.
3. Control cable.
4. Parallelogram linkage.
5. Control valve.
6. Center pivot point.
7. By-pass valve.
8. Pin in over-size hole.
9. "Feel" lever.
10. Aileron boost cylinder.

Description of Aileron Boost Unit Operation:

In the neutral position, no force is applied to the control cables (figure 15-3) the pin is centered in the over-sized hole (figure 15-8) and the control valve (figure 15-5) is closed.

As a force is applied to the control cable, the large control wheel is rotated slightly, the pin is displaced from the center of the oversize hole, and the resultant distortion of the parallelogram linkage (figure 15-4) moves the control valve to one side. This displacement of the control valve allows hydraulic fluid to flow to the boost cylinder (figure 15-10) and assist the original force in moving the aileron.

As the motion of the control wheel is stopped, hydraulic fluid continues to flow to the boost cylinder until the pin is again centered in the hole and the control valve is returned to the OFF position. In this position, the application of force by the pilot continues to act through the feel lever (figure 15-9) and prevents the air loads on the aileron from displacing the pin from the center of the hole.

In the event of failure of the hydraulic pressure, or shutting off the aileron boost shut-off valve (figure 9-5) in the cockpit, the low pressure by-pass actuator cylinder (figure 15-2) moves under spring tension and opens the by-pass valve (figure 15-7). The opening of the by-pass valve allows free flow of fluid from one side of the boost cylinder to the other and prevents the occurrence of a hydraulic lock condition.

With pressure in the boost system thus reduced to zero, full manual operation of the ailerons is obtained when the pin hits the side of the oversize hole (figure 15-8).

Figure 15 — Aileron Boost Unit Diagram

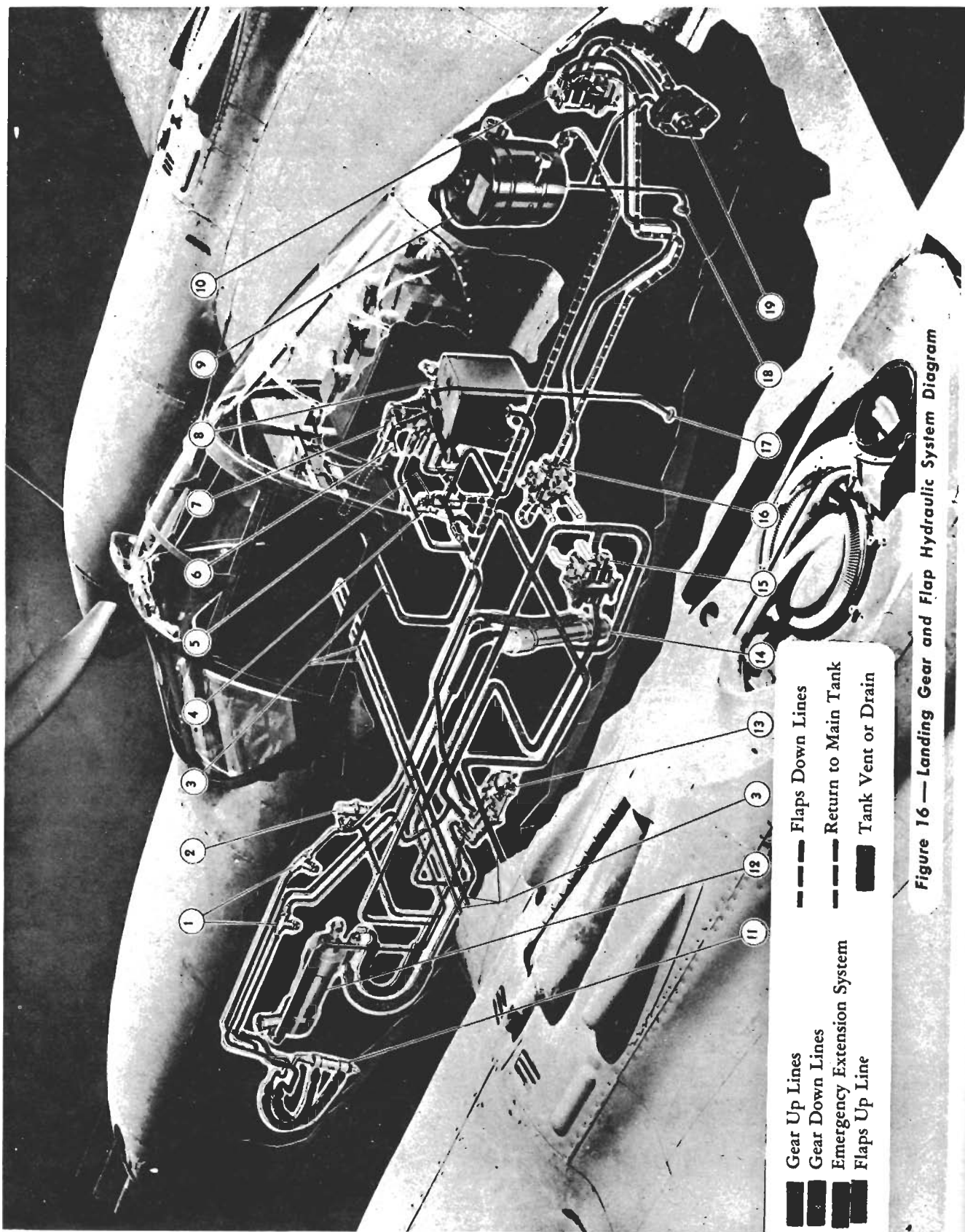
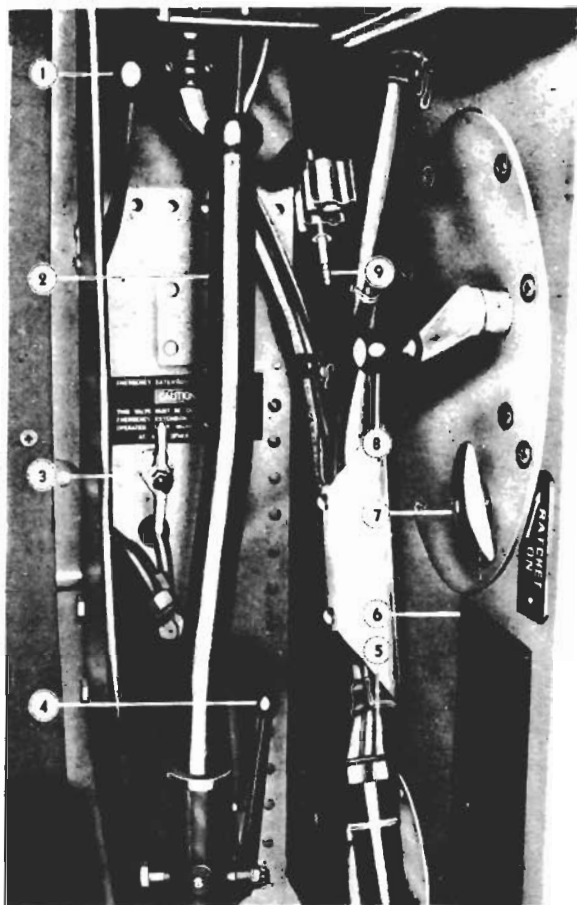


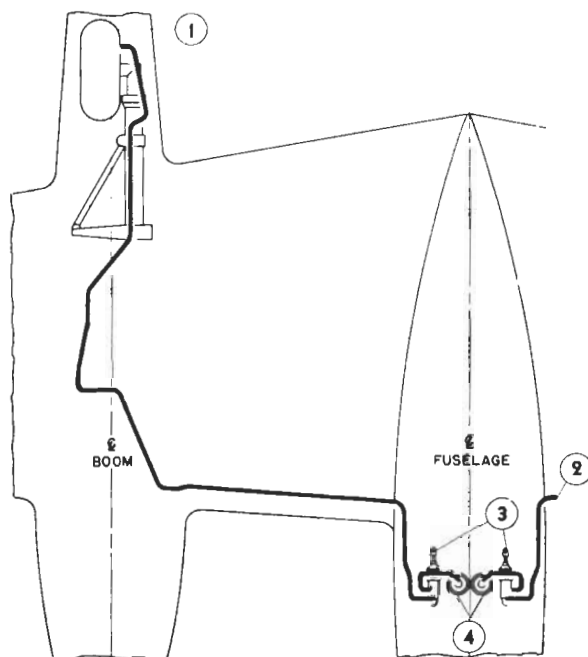
Figure 16 — Landing Gear and Flap Hydraulic System Diagram



1. Seat adjustment release handle.
2. Hand pump handle.
3. Emergency extension by-pass valve. (Early airplanes only.)
4. Source selector valve handle.
5. Radio jack (stowed position).
6. Emergency extension procedure placard.
7. Window ratchet.
8. Right-hand window crank.
9. Radio jack.

Figure 17 — RIGHT SIDE COCKPIT

- Key to Fig. 16**
1. Combination check and relief valve. (Allows free flow in one direction, high pressure flow in the other direction.)
 2. Sequence valve (open only when wheel is UP).
 3. Lines to main landing gear system. (2)
 4. Emergency extension by-pass valve.
 5. Pressure relief valve. (DOWN position).
 6. Source selector valve (DOWN position).
 7. Emergency hand pump.
 8. Emergency extension reservoir.
 9. Main hydraulic reservoir.
 10. AUTOMATIC flap stop valve (stops flaps at UP, DOWN, or MANEUVER depending on setting of control lever).
 11. Landing gear door locking cylinder.
 12. Landing gear extension cylinder (down-lock built in).
 13. Landing gear uplock.
 14. Landing gear door cylinder.
 15. Landing gear control valve.
 16. Flap control valve.
 17. Vent to atmosphere.
 18. Main reservoir drain.
 19. Flap motor and gear box.



1. Main landing gear wheel and brake.
2. To left-hand gear.
3. Brake cylinders.
4. Hydraulic fluid reservoirs.

Figure 18 — Brake System Diagram

1. Automatic oil temperature regulator.
2. Line to oil pressure gage.
3. Oil tank (capacity 13 U.S. gal., 11 Imperial gal.).
4. Supercharger regulator. (EARLY AIRPLANES ONLY)
5. Vents to atmosphere.
6. Oil from engine.
7. Oil cooling radiators.
8. Oil to engine.

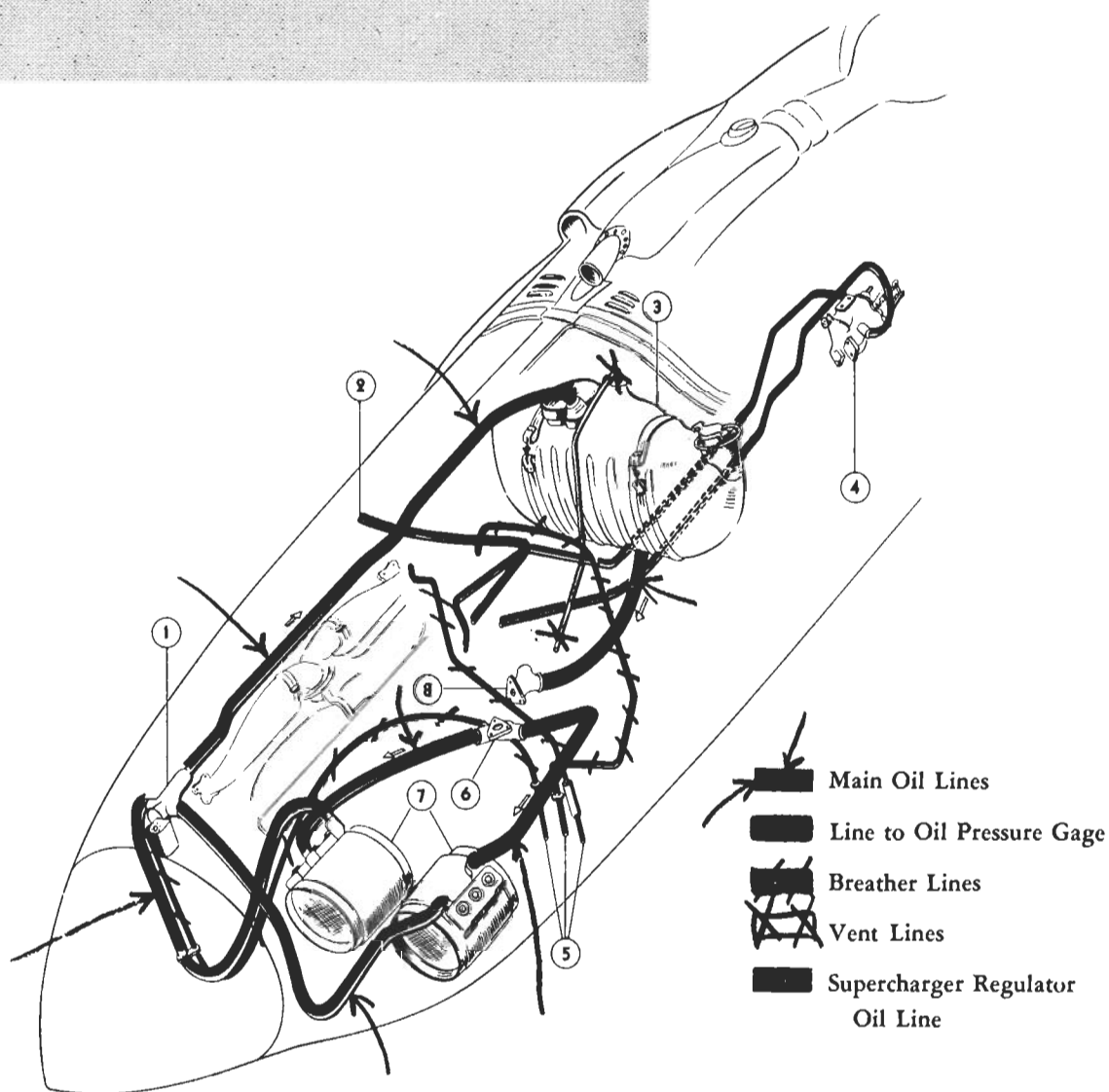


Figure 19 — Oil System Diagram

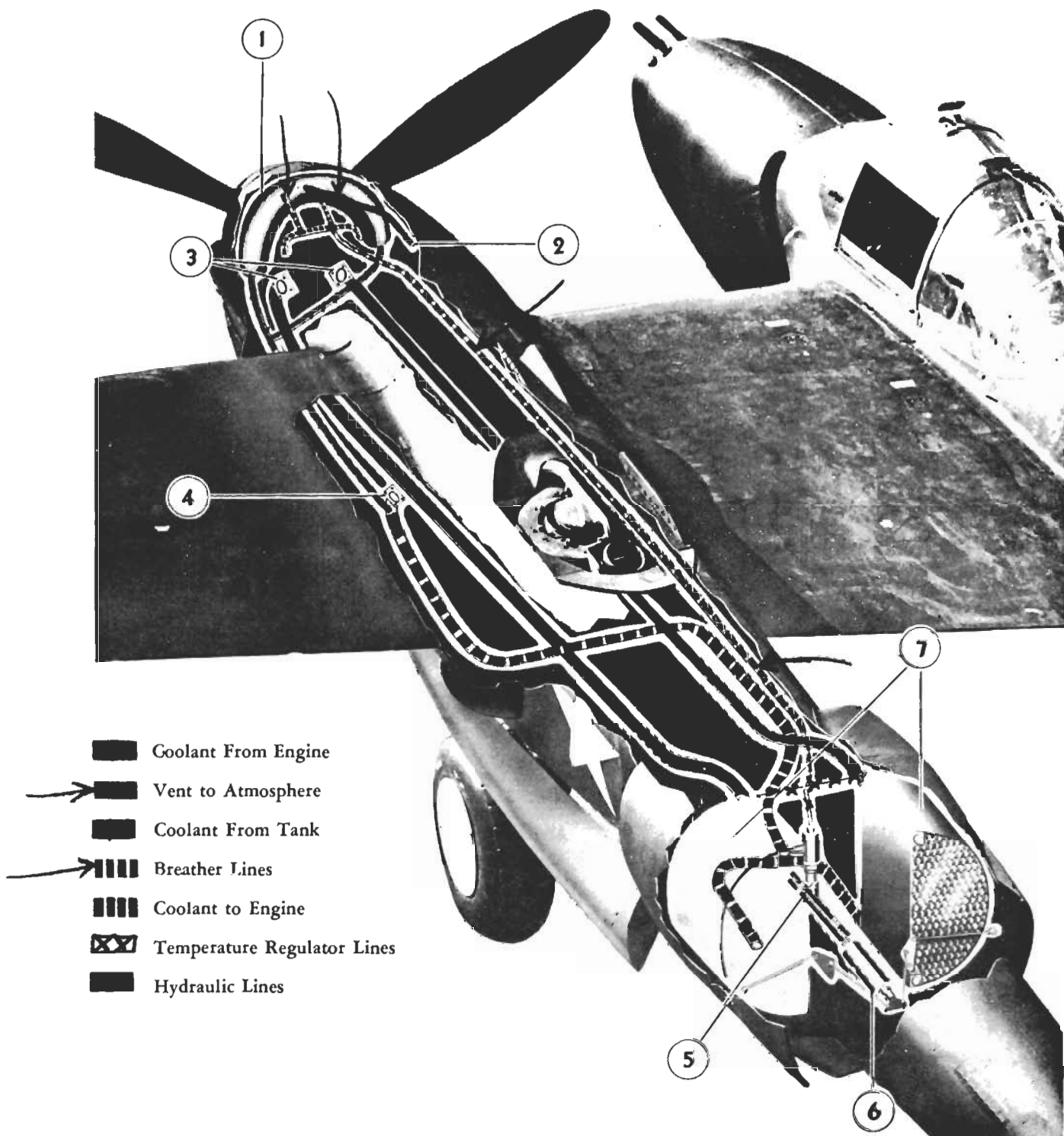
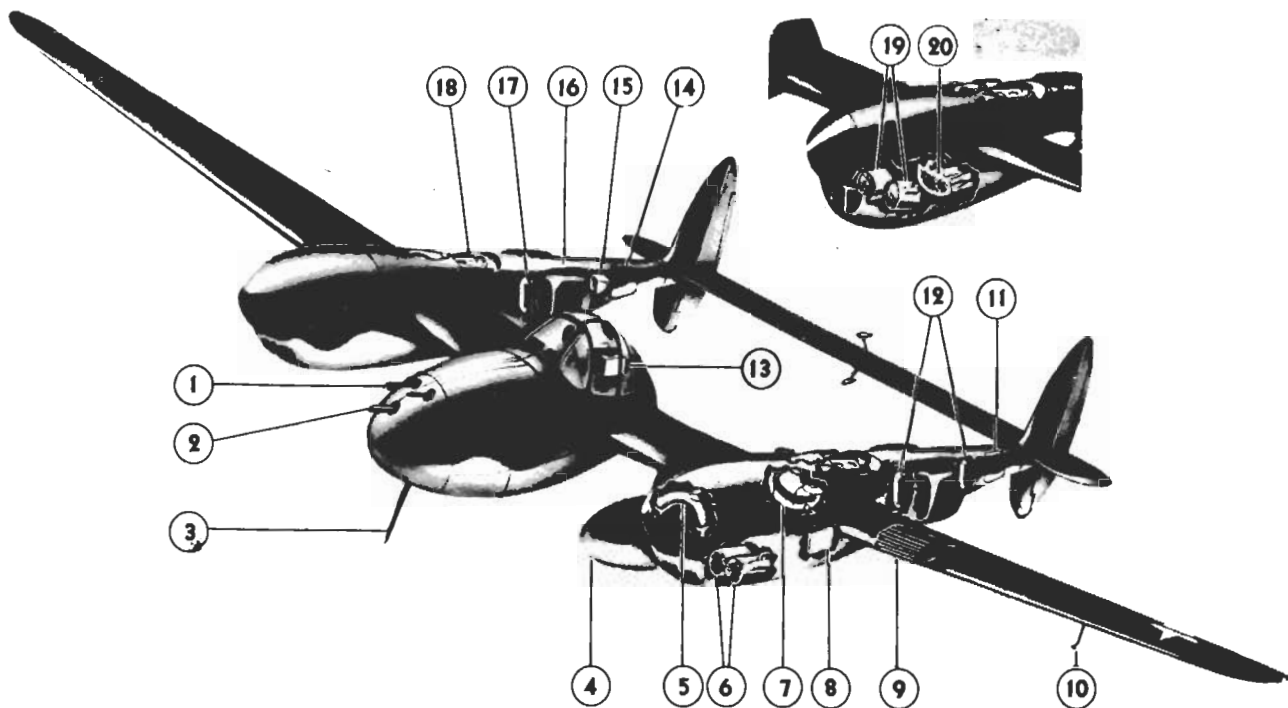


Figure 20 — Coolant System Diagram



- | | |
|--|--|
| 1. .50 caliber machine guns. | 11. Battery compartment door. |
| 2. 20 mm cannon. | 12. Oxygen bottles. |
| 3. Antenna mast. | 13. Radio equipment. |
| 4. Droppable fuel tank. | 14. Baggage compartment door. |
| 5. Coolant tank. | 15. Recognition radio. |
| 6. Oil cooler radiators (P-38H). | 16. Coolant radiators. |
| 7. Oil tank. | 17. Oxygen bottle. |
| 8. Carburetor air filter. | 18. Turbo supercharger. |
| 9. Carburetor air intercooler (P-38H). | 19. Oil cooler radiators (P-38J and F-5B). |
| 10. Airspeed pitot. | 20. Carburetor air intercooler (P-38J and F-5B). |

Figure 21 — Airplane Contents Diagram



Pilot Operating Instructions

1. FLIGHT RESTRICTIONS.

a. MANEUVERS PROHIBITED.

- (1) Snap rolls and intentional spins.
- (2) Continuous inverted flight.

(3) Airspeed or accelerations in excess of those given on the DIVE LIMITS placard (figure 25) which is posted in the cockpit of each airplane. Do not exceed 3.5 G's negative acceleration.

CAUTION

Extreme care must be taken during acrobatic maneuvers which require a downward recovery. Acrobatics should not be attempted at altitudes below 10,000 feet.

b. AIRSPEED LIMITATIONS.

Condition	Maximum Allowable Airspeed (Indicated)
Diving.....	See Placard (figure 25) Section II paragraph 18
Landing gear extended.....	175 mph.
Flaps 100% extended.....	150 mph.
Flaps 50% extended.....	250 mph.
Landing light extended.....	140 mph.
300 Gal. droppable tanks installed.....	250 mph.

2. BEFORE ENTERING PILOT'S COMPARTMENT.

a. Check the loading of the airplane. Determine the approximate take-off weight and the center of gravity position.

WARNING

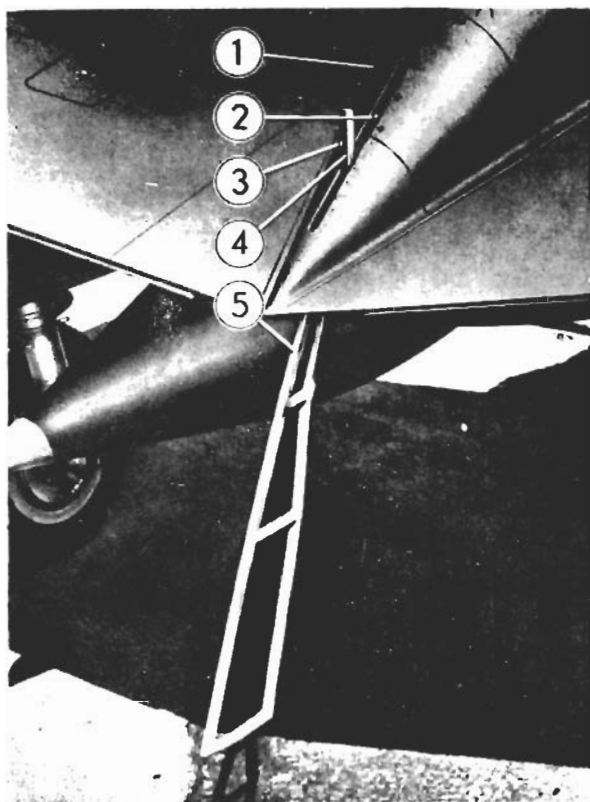
Dangerous instability exists when the center of gravity is aft of 32% mac. (32% gear up corresponds to 28.5% gear down.) Under these tail heavy conditions, full down elevator will be required to prevent stalling the airplane if the airspeed is allowed to drop below 90 mph indicated with flaps down, power on, and landing gear up.

Note

Tail heavy conditions may be relieved by lowering the landing gear.

b. Check that the cannon and machine guns have been charged and that the radio transmitters have been tuned to the proper frequencies.

c. Access to the airplane is by means of the retractable ladder on the rear of the fuselage (figure 22-5). Push the up-lock release (figure 22-2) and raise the handle (figure 22-4) to a vertical position. Force the handle down until the ladder locks in the position shown. To



1. Flush type handhold.
2. Uplock release.
3. Downlock release.
4. Ladder lever.
5. Ladder.

Figure 22 — Access Ladder

retract the ladder, push the downlock release (figure 22-3) and pull the handle straight up until the ladder stows in place, then swing the handle forward until flush with the fuselage contour, and press firmly into place. A flush, hinged handhold (figure 22-1) is built into the left side of the fuselage.

(1) To open the top hatch, turn the two releases (figure 23-3) on the top of the windshield and rotate the hatch backward.

3. ON ENTERING PILOT'S COMPARTMENT.

a. CHECK FOR ALL FLIGHTS.

(1) Battery switch (figure 5-15) OFF if battery cart is used. ON if cart is not used.

(2) Cross-feed switch (figure 13-6) OFF. (Cross-feed switch is replaced by a cross-feed position of the tank selector valves on later airplanes.)

(3) Turn the tank selector valves to OUTER WING ON (or outer wing tank switches ON).

(a) Check operation of outer wing tank booster pumps by checking the fuel pressure.

(b) Check the condition of the low level warning light bulbs by pushing the test button on the side of the warning light box, or (on modified airplanes) press the bulbs into their sockets.

(4) Oxygen pressure (figure 4-34) 400 to 450 lb/sq in.

(5) Bomb selector switches (figure 4-20) ON and arming switch (figure 4-24) SAFE. (These positions insure that tanks or bombs may be dropped quickly in the event of engine failure at take-off.

(6) Throttles (figure 4-2) 1/10 OPEN. ($\frac{3}{4}$ inch.)

(7) Propeller control (figure 4-4) INC RPM. (Full forward.)

(8) Propeller selector switches (figure 4-5) AUTO CONSTANT SPEED.

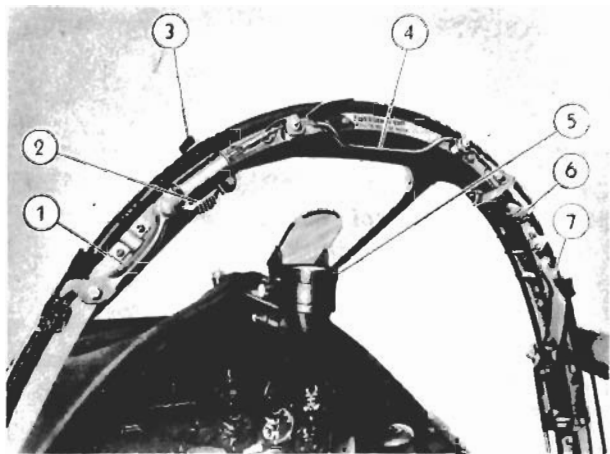
(9) Propeller circuit breakers (figure 4-9) ON.

(10) Propeller feathering switches (figure 4-13) NORMAL.

(11) Mixture (figure 4-6) IDLE CUTOFF.

(12) Oil cooler flap switches (figure 5-17) AUTOMATIC.

(13) Generator switch (figure 5-16 or 8-11) ON.



1. Hatch locking arm (locked position).
2. Hatch release button.
3. External hatch release levers
4. Hatch release handle.
5. Gunsight (late airplanes).
6. Hatch release button.
7. Hatch locking arm (released position).

Figure 23 — Hatch Controls

(14) Coolant flap override switches (figure 5-13) OFF.

(15) Intercooler flaps (figure 5-12) OPEN . (If installed.)

(16) Gun-sight light (figure 5-10) operating and seat adjusted so that the sight reflection is easily visible.

(17) Inverter switch (figure 5-9) (or compass switch on main switch box) ON.

(18) Contactor heater ON if contactor is to be used during the flight.

(19) Armament switch (on control column) OFF.

(20) Fuel quantity (figure 8-20 and 21) adequate. Check the fuel level in outer wing tanks by pressing the low level test button (figure 12-4), or, (on modified airplanes) by operating the low level test switch on the side of the warning light box.

(21) Turbo-supercharger warning lights functioning (if installed).

(22) Carburetor air filters (figure 4-8) AS REQUIRED.

Note

The use of carburetor air filters reduces the critical altitude and range of the airplane and should be avoided in clear air.

(23) Clock and altimeter set.

b. SPECIAL CHECK FOR NIGHT FLYING.—Test by operating.

(1) Landing lights (figure 5-7). (Not more than 5 seconds for test.)

(2) Recognition lights (figure 7-13). (Not more than 10 seconds for test.)

(3) Cockpit lights (figure 5-11).

(4) Fluorescent lights (figure 5-5).

(5) Position lights (figure 5-6).

(6) Spot light (figure 4-1).

4. FUEL SYSTEM MANAGEMENT.

a. GENERAL.—On unmodified airplanes fuel is supplied to each engine by an engine driven fuel pump and one master booster pump which draws fuel from either the main, reserve, or droppable tank depending on the setting of the selector valve. (An additional booster pump for the outer wing tank is installed on airplanes equipped with these tanks.)

On modified airplanes fuel is supplied to each engine by an engine driven fuel pump and an individual

booster pump for each tank. The droppable tanks on modified airplanes are also pressurized to 5 lb/sq in.

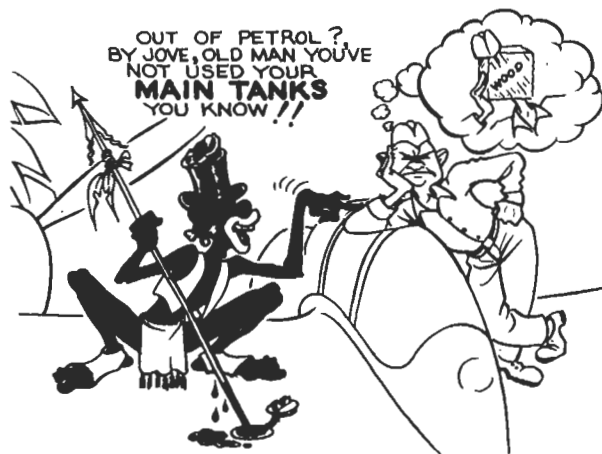
On all airplanes equipped with outer wing tanks, two low level warning lights are installed on the forward left-hand side of the cockpit. These lights automatically come on when fuel for approximately 5 or 6 minutes of engine operation remains in the tank.

b. NORMAL USE.

(1) Warm up, take off and fly for the first 15 minutes on RESERVE tanks. This is to provide space in the reserve tanks for the vapor return from the carburetors. Switch both engines to the *left* droppable tank until almost dry, then shift to *right* droppable tank until almost dry. Determine hourly fuel consumption from the charts in Appendix II. Fuel gages are not installed in the droppable tanks. Do not drop external tanks unless necessary for increased range or for combat. Use up the fuel in the outer wing tanks (if installed); then use main tanks, and switch back to RESERVE for the remainder of the flight.

WARNING

Always check the fuel level in the tank before trying to operate the engine from that tank. To check the fuel level in the outer wing tanks, press the low level test button (figure 12-4), or, (on modified airplanes), operate the low level test switch on the side of the warning light box. It is not possible to check the fuel level in the droppable tanks.



(2) On unmodified airplanes the booster pumps (figures 12-3, or 13-4) should be operated during take-off and landing to prevent engine failure which may result from engine-driven fuel pump failure. The booster

pumps should also be operated during flight whenever necessary to maintain the required 16 to 18 lb/sq in. fuel pressure.

On modified airplanes the booster pump master switches (figure 13A-4) should be on at all times. The speed control switch should be in the EMERGENCY position during take-off and landing to prevent engine failure which may result from engine-driven fuel pump failure. The booster pumps should also be operated on EMERGENCY whenever necessary during flight to maintain the desired fuel pressure of 16 to 18 lb/sq in. The speed control switches have no effect on the droppable tank booster pumps. On modified airplanes the booster pump switches merely supply power to the booster pumps. The proper pump is turned on by contacts on the tank selector valve.

Note

Never exceed 250 mph indicated with 300 gallon droppable tanks installed.

(3) To release droppable fuel tanks:

(a) Raise flaps and landing gear.

(b) Turn tank selector valves (figures 13-8 and 13-9) to MAIN, RESERVE, or OUTER WING.

(c) Move arming switch (figure 4-24) to ARM or SAFE.

(d) Turn selector switches (figure 4-20) ON for tank(s) to be dropped.

(e) Press the release button (figure 4-27) when flying at an angle not greater than 30° from the horizontal.

Note

An emergency droppable tank or bomb release control is installed in late airplanes. It is located either behind and to the left of the seat or directly in front of the pilot's seat.

(f) Full fuel tanks may be dropped without danger at airspeeds up to 400 mph. Empty 150 gallon tanks should be dropped only while flying at an airspeed of 160 mph or less. On late airplanes, droppable tanks are equipped with displacement struts which increase the safe dropping speed to 350 mph.

WARNING

EMPTY 300 GALLON TANKS ARE TO BE DROPPED ONLY IN AN EMERGENCY as the tanks may hit the airplane when released. To drop the tanks, it is necessary to slow the airplane down to 120 mph with landing gear and flaps up to avoid serious damage.

b. LONG RANGE FERRY FLIGHT.

(1) Whenever flying with droppable tanks, it is advisable to operate both engines from the LEFT droppable tank until empty and then operate both engines from the RIGHT droppable tank. This procedure empties the left tank in the minimum time and, if necessary, it can be released sooner than by operating each engine from its own individual tank.

c. CROSS FEED OPERATION.

(1) On early airplanes the left and right fuel systems are connected by an electrically operated "cross-feed" valve which makes it possible to operate either engine from any tank, except the outer wing tank. Late airplanes have a crossfeed position on the tank selector valves. When prolonged single engine flight makes it necessary to use fuel from the dead engine side, or when operating both engines from one droppable tank, operate the fuel system as follows:

(a) Airplanes with four-way fuel tank selector valves.

1. Turn tank selector valve to the tank to supply fuel.

2. Turn crossfeed switch (figure 13-6) to CROSSFEED.

3. Turn other tank selector valve OFF.

(b) Airplanes with five-way fuel tank selector valves.

1. Turn tank selector valve to the tank to supply fuel.

2. Turn other tank selector valve to CROSS-SUCTION.

Note

It is not possible to cross-feed fuel from the outer wing tanks.

CAUTION

Do not attempt to use the booster pumps on modified airplanes during cross-feed operation if there is a leak in the fuel lines to the dead engine. The booster pumps will pressurize the fuel lines, forcing fuel out through the leak.

5. STARTING ENGINES.

Note

Engine fire extinguishers are NOT installed in this airplane. Strict adherence to the following instructions as to mixture control positions will reduce the possibility of fire. If fire does occur, shut off mixture control, tank selector valve, electric fuel pump, and ignition to the affected engine.

a. With ignition OFF and mixture controls at IDLE CUT OFF, turn the engines over by hand two or three revolutions if they have been idle for more than three hours.

b. On unmodified airplanes check the operation of the fuel booster pumps (figure 12-3 or 13-4). The fuel pressure with the engines not operating should be 15 to 16 lb/sq in. On modified airplanes check the operation of the fuel booster pumps (figure 14A-3 and 4) on each tank. The fuel pressures with the engines not operating should be as follows:

(1) With the speed control switches on NORMAL—8 to 10 lb/sq in. on any tank except the droppable tanks.

(2) With the speed control switches on EMERGENCY—16 to 18 lb/sq in. on any tank except the droppable tanks.

(3) Droppable tank fuel pressure—15 to 16 lb/sq in.

c. Turn the booster pumps on unmodified airplanes OFF after above test. On modified airplanes, leave booster pumps ON and return the speed control switches to NORMAL.

d. Return the tank selector valves to RES. ON and prime the left engine. On unmodified airplanes, push the primer handle down, turn 90° to unlock, and prime two to four strokes. On modified airplanes, hold the priming-oil dilution switch (figure 5-2) in the aft position two to four seconds.

Note

Priming time will be shortened if the booster pumps are on EMERGENCY.

e. Turn ignition master switch (figure 5-1) ON.

f. Turn left ignition switch (figure 5-18) to BOTH.

g. Hold the starter switch (figure 5-3) to LH (left-hand) until the inertia starter has reached maximum rpm.

h. Push the engage switch (figure 5-4) to LH, still holding starter switch to LH, and prime as required.

i. As soon as the engine definitely fires, place the mixture control (figure 4-6) to AUTO RICH.

Note

It may be necessary to operate the electric fuel pumps for a short while if the engine driven pumps do not build up pressure immediately.

CAUTION

Due to the fire hazard, electric fuel pumps must not be operated unless the mixture control is in the CUTOFF position or unless the engine is running.

j. Return the mixture control to IDLE CUTOFF if the engine does not continue to run.

k. Stop the engine if oil pressure does not register within 30 seconds.

l. Start the right-hand engine by repeating the preceding paragraphs 5f through j.

m. Lock the primer pump DOWN.

n. Turn battery switch (figure 5-15) ON before disconnecting the battery cart.

Note

If battery power is not sufficient for starting, use the inertia starter handcrank or an external energizer. The handcrank is stowed in left main landing gear wheel-well.

6. ENGINE WARM-UP AND ACCESSORY CHECK.

a. Keep the RPM under 1400 until the oil temperature reaches 40°C (105°F) or shows a definite increase of 10°C (18°F) and the oil pressure is steady below 75 lb/sq in.

b. While the engines are warming up, test the communication equipment with the control tower, or with another airplane.

7. EMERGENCY TAKE-OFF.

a. When necessary, take-off may be made without the normal engine and accessories ground tests *provided* that the oil pressure is steady below 85 lb/sq in. and that the oil temperature has shown a definite increase of at least 10°C (18°F) since starting. If necessary, use the oil dilution system to reduce the oil pressure.

b. Over-dilution is likely to result from diluting the oil in a cold engine. If dilution is necessary during warm-up, oil pressure should be carefully watched during the remainder of warm-up and during take-off to insure that over-dilution has not occurred.

8. ENGINE AND ACCESSORIES OPERATION GROUND TEST.

a. Extend and retract the flaps (figure 7-2) to check

the operation of the hydraulic system. When both engines are operating at 1400 RPM, fifteen to twenty seconds is the normal extension time; twenty-five seconds is the normal extension time if only one engine is operating at 1400 RPM.

b. Check for normal fuel pressure (figures 8-7 and 8-14) 16 to 18 lb/sq in. with electric fuel pumps OFF.

■ Check for idling pressure of 9 lb/sq in.

c. Increase RPM to 2300.

(1) Check propeller control levers (figure 4-4) DEC RPM then INC RPM (full forward).

(2) Check propeller selector switches (figure 4-5) DEC RPM then INC RPM, then return to AUTO CONSTANT SPEED. Be sure that propeller warning lights (on P-38H only) glow when selector switches are out of AUTO CONSTANT SPEED.

(3) Check magnetos. Maximum normal drop, 100 RPM after shifting from both to either left or right magneto. *Engine must run smoothly on either magneto.*

(4) With the generator switch (two on later airplanes) ON, check the voltmeter (figure 5-8) for approximately 28 volts and the ammeter (two on later airplanes) (figure 8-12) for charge.

(5) Move intercooler flap controls to OPEN (if installed) and check operation of turbo-superchargers by opening throttles individually to take-off power (see specific engine Flight Chart in Section III).

CAUTION

Do not operate engines at take-off power for more than two or three seconds while on the ground.

d. Type A-4 Automatic Pilot (F-5B only).

(1) Check vacuum (figure 8-2) 4" to 5" Hg.

(2) Check automatic-pilot oil pressure (figure 33-10) 125 lb/sq in.

(3) Check the artificial horizon unit uncaged (figure 24-5).

(4) Match the directional gyro cards on the directional gyro (figure 24-6).

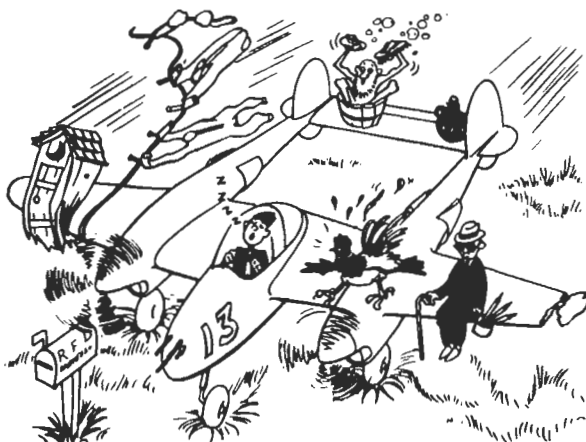
(5) Turn the automatic-pilot control valve (figure 24-10) ON.

(6) Turn the course setting knob (figure 24-4), the elevator trim knob (figure 24-9), and the aileron trim knob (figure 24-7) to check operation of the units.

(7) TURN AUTOMATIC PILOT CONTROL VALVE OFF BEFORE TAKE-OFF.

9. TAXIING INSTRUCTIONS.

The airplane taxis easily and forward visibility is good. Use differential throttle control for turning to save the brakes. There is no danger of nose over or ground loop should it become necessary to turn sharply or to apply full brakes.



WATCH WHERE YOU'RE GOING WHEN TAXIING!!

10. TAKE-OFF.

a. Check the following:

(1) Top hatch—LOCKED IN PLACE. Side windows—cranked (figure 13-2) CLOSED. Side window ratchets (figure 13-3)—ON.

Note

Open side windows will cause buffeting of the tail section.

(2) Propeller levers (figure 4-4) INC RPM (full forward).

(3) Propeller selector switches (figure 4-5) AUTO CONSTANT SPEED.

(4) Mixture (figure 4-6)—AUTO RICH.

(5) Tank selector valves (figures 13-8 and 13-9) RESERVE ON.

(6) Dive flaps (figure 9-2)—UP. (P-38L and late P-38J airplanes.)

(7) Wing flaps UP. Wing flap lever (figure 7-2) CLOSED. Up to 1/2 flaps may be used for short take-off run.)

(8) Flight controls free and proper movement.

Note

Look at surfaces for this check to see that they move in the right direction.

(9) Aileron boost shut-off valve (figure 9-5) ON. Aileron boost should be on at all times. Turn off only in case of emergency.

WARNING

Never turn the aileron boost shut-off valve from OFF to ON during flight.

Note

A slight "nibbling" at the control wheel may be noticed on take-off. This is because retraction of the landing gear may reduce the hydraulic system pressure below 200 lb/sq in. If the system pressure falls below this value, the aileron boost will become inoperative. As the pump increases the pressure, the "nibbling" may be noticed when the boost again becomes effective.

(10) Generator switch(es) (figure 5-16 or 8-11) ON.

(11) Intercooler flaps (figure 5-12) OPEN if installed).

(12) Droppable tanks or bombs prepared for immediate dropping by turning bomb selector switches (figure 4-20) ON and arming switch (figure 4-24) to SAFE.

(13) Fuel booster pumps (figures 12-3, 13-4, or 13A-3) ON. Booster pump speed control switches (figure 13A-4) EMERGENCY. (Speed control switches are on modified airplanes only.)

(14) Rudder, elevator tab 3° back and aileron tabs ZERO.

(15) Taxi a few feet *straight* down the runway so that the nose wheel will be in line when take-off power is applied. Maximum performance take-offs require holding the airplane with brakes at the end of the runway until engine power reaches the desired setting. Because of the tricycle landing gear, there is no tendency for the airplane to take-off by itself, and no feeling of lightness as take-off speed is reached. Start to ease the control column back at about 70 mph, then at 90 or 100 lift the airplane into the air.

(16) Hold brakes, open throttles to 46" Hg. 3000 RPM.

(17) Release brakes and keep manifold pressure below 54" Hg.

CAUTION

Be prepared to reduce power immediately to prevent uncontrollable yaw and roll in case of failure of one engine during take-off.

(18) Retract landing gear as soon as practical after leaving the ground.

Note

Retract the landing gear immediately after the airplane is off the ground so that the flight may be safely continued in the event of one-engine failure after take-off.

(19) Reduce manifold pressure to 43" Hg. at 2600 rpm after clearing all obstacles.

11. ENGINE FAILURE DURING TAKE-OFF.

a. If one engine fails before the airplane leaves the ground, close both throttles immediately and apply full brakes. If it is going to be impossible to stop on the runway, retract the landing gear by turning the landing gear control release knob and raising the landing gear lever.

b. If one engine fails after the airplane leaves the ground, but before the safe single-engine airspeed (120 mph) has been reached, close both throttles and LAND STRAIGHT AHEAD. Retract the landing gear if it is not possible to land on the runway.

c. If one engine fails after reaching the safe airspeed of 120 mph and after the landing gear has started up:

(1) Reduce power enough to regain control, then apply power gradually. Hold enough rudder to prevent the airplane from skidding and hold the wings level.

(2) Release droppable tanks or bombs over an unpopulated area.

(3) Trim rudder tabs.

(4) Set the mixture on the dead engine to IDLE CUT-OFF.

Note

Do not apply so much power that the airplane cannot be held straight with the rudder. A manifold pressure of 45" Hg. at 3000 RPM should be enough to accelerate the airplane to a good single engine climbing airspeed of 165 mph.

(5) Feather the dead engine's propeller.

(6) Turn OFF electric fuel pump of dead engine.

(7) Circle the field and land, do not make turns into the dead engine unless trim and speed have been establish.

12. CLIMB.

a. Mixture (figure 4-6) AUTO RICH.

b. Intercooler flaps (figure 5-12) OPEN (if installed).

c. Refer to the Take-off, Climb and Landing Chart in Appendix II for the best climbing speeds at different weights, powers and altitudes. The average best-climbing-speed at sea level is 160 mph.

d. On P-38H airplanes, carburetor air temperature is critical in a high power climb between 15,000 and 25,000 feet. Above 25,000 feet turbo-supercharger overspeed is critical. Excessive temperatures will cause detonation and very rough engine operation resulting in loss of power and probable engine damage.

(1) On P-38J, P-38L, and F-5B airplanes *with intercooler flaps OPEN*, the manifold pressure is limited by the rating of the engine up to 25,000 feet. Above 25,000 feet, turbo-supercharger overspeed is again critical.

e. The following maximum manifold pressures are to be used for "War Emergency" only. *Never exceed 60 inches Hg.* In a climb above 25,000 feet these limits indicate the approximate values at which the turbo-supercharger warning lights should glow. In level flight it should be possible to obtain the manifold pressure shown below at slightly higher altitudes.

Altitude Feet	P-38J, P-38L and F-5B		
	P-38H B-13 turbo	B-33 turbo	B-33 turbo
up to 7,000	60	60	60
20,000	55	55	60
25,000	45	52	60
30,000	35	49	53
35,000	30	44	45
40,000	20	36	37

Early P-38H airplanes are equipped with type B-13 turbo-superchargers. Later P-38H airplanes and P-38J, P-38L, and F-5B airplanes are equipped with type B-33 turbo-superchargers.

f. Refer to the Specific Engine Flight Chart in Section III for power time limitations.

g. Refer to the Take-off, Climb and Landing Chart in Appendix II for rate of climb. Note correction to be made during hot weather.

13. GENERAL FLYING CHARACTERISTICS.

a. Due to the counter-rotating propellers, there is no noticeable torque effect in any two engine flying with this airplane. Rudder and aileron trim tab settings do not require adjustment as a result of changes in airspeed and power.

b. TO INCREASE POWER IN FLIGHT.

(1) Move mixture controls (figure 4-6) to AUTO RICH if maximum cruising power is to be exceeded (see Specific Engine Flight Chart in Section III).

(2) Move propeller controls (figure 4-4) to the new RPM.

(3) Move throttles (figure 4-2) to the new manifold pressure.

c. TO DECREASE POWER IN FLIGHT.

(1) Move throttles (figure 4-2) to the new manifold pressure.

(2) Move propeller controls (figure 4-4) to the new RPM.

(3) Re-adjust the throttles.

(4) Move mixture controls (figure 4-6) to AUTO LEAN if permissible.

d. The turbo-superchargers are controlled by the same levers which operate the throttles. On P-38H, F-5B, and early P-38J airplanes supercharger overspeed is indicated by warning lights. Rated turbo-supercharger speed is 24,000 RPM and overspeed is 26,400 RPM allowable for 5 minutes. The warning lights start to flicker at 25,600 turbo-supercharger RPM and burn continuously at 26,400 RPM. Operation within the flickering range is permissible only during "War Emergencies" and the throttles must be retarded when the flicker changes to continuous burning. P-38L and late P-38J airplanes are equipped with supercharger regulators incorporating an overspeed control and on these airplanes overspeed warning lights are not installed.

e. On P-38J, P-38L, and F-5B airplanes the intercooler flaps should be open for take-off and climbs and nearly closed at all other times. Carburetor air temperature should not be allowed to exceed 45°C (113°F). P-38H airplanes are not equipped with intercooler flaps.

f. Flight operations should be planned from the Flight Operations Instructions Charts in Appendix II. When using these charts, make sure that the chart being used is applicable to the airplane. Charts are clearly marked to indicate the airplane model, the weight, and the external load items carried. If the weight or external load is to be changed during the flight, be sure to use the proper charts.

g. The airplane is stable at all normal speeds. The airplane becomes slightly nose-heavy when the flaps and landing gear are extended. Release of droppable fuel tanks causes no noticeable change. Two-engine cruising below 170 mph indicated airspeed is not recommended because the airplane requires more attention and range is not increased.

14. AUTOMATIC PILOT OPERATING INSTRUCTIONS. (F-5B Only)

a. Trim the airplane for "hands off" flight.

b. Set the directional gyro (lower) card to agree with the compass and rotate the course setting knob until the upper and lower cards coincide.

c. Check the directional gyro. (Uncaged)

d. Turn the automatic pilot elevator and aileron trim knobs to zero.

e. Turn the automatic pilot control valve ON.

f. Adjust the course setting knob and the trim knobs to hold the airplane in straight and level flight.

g. Adjust the speed valves (figure 24-2) to obtain the proper rate of control.

h. Disengage automatic pilot every 15 minutes in flight to reset the directional gyro with the compass and retrim the airplane for "hands off" flight.

i. Make small directional changes (flat turns) by turning the course setting knobs slowly to the new heading.

j. Make faster turns by setting the airplane in a bank with aileron trim knob, caging the directional gyro, and rotating the course setting knob until the ball bank indicator is centered.

CAUTION

The operating limits are 55° in a climb or dive and 90° in a bank. Both gyros should be caged during any maneuvers which might exceed these limits.

15. STALLS.

a. With power OFF, the airplane stalls at the following indicated airspeeds at the gross weight noted:

	15,000 lb.	17,000 lb.	19,000 lb.
Flaps and landing gear UP	94 mph	100 mph	105 mph
Flaps and landing gear DOWN	69 mph	74 mph	78 mph

b. As stalling speed is approached, the center section stalls first with noticeable shaking of the airplane, however, the ailerons remain effective.

c. In either "power-on" or "power-off" stalls with flaps and landing gear up, the airplane "mushes" straight forward in a well-controlled stall. With flaps and landing gear down, there appears to be a slight tendency for one wing to drop. There is, however, no tendency to spin. Under these conditions, the nose drops slightly and, as the speed increases, the wing will come up.

16. SPINS.

a. Deliberate spinning is prohibited because the spin tends to flatten out after two or three turns. When this occurs, the control column is forced back and engine power must be used to help get the control column forward. Before flattening out, normal recovery may be made without power. Recovery is made by applying full opposite rudder and easing the control column forward.

17. ACROBATICS.

CAUTION

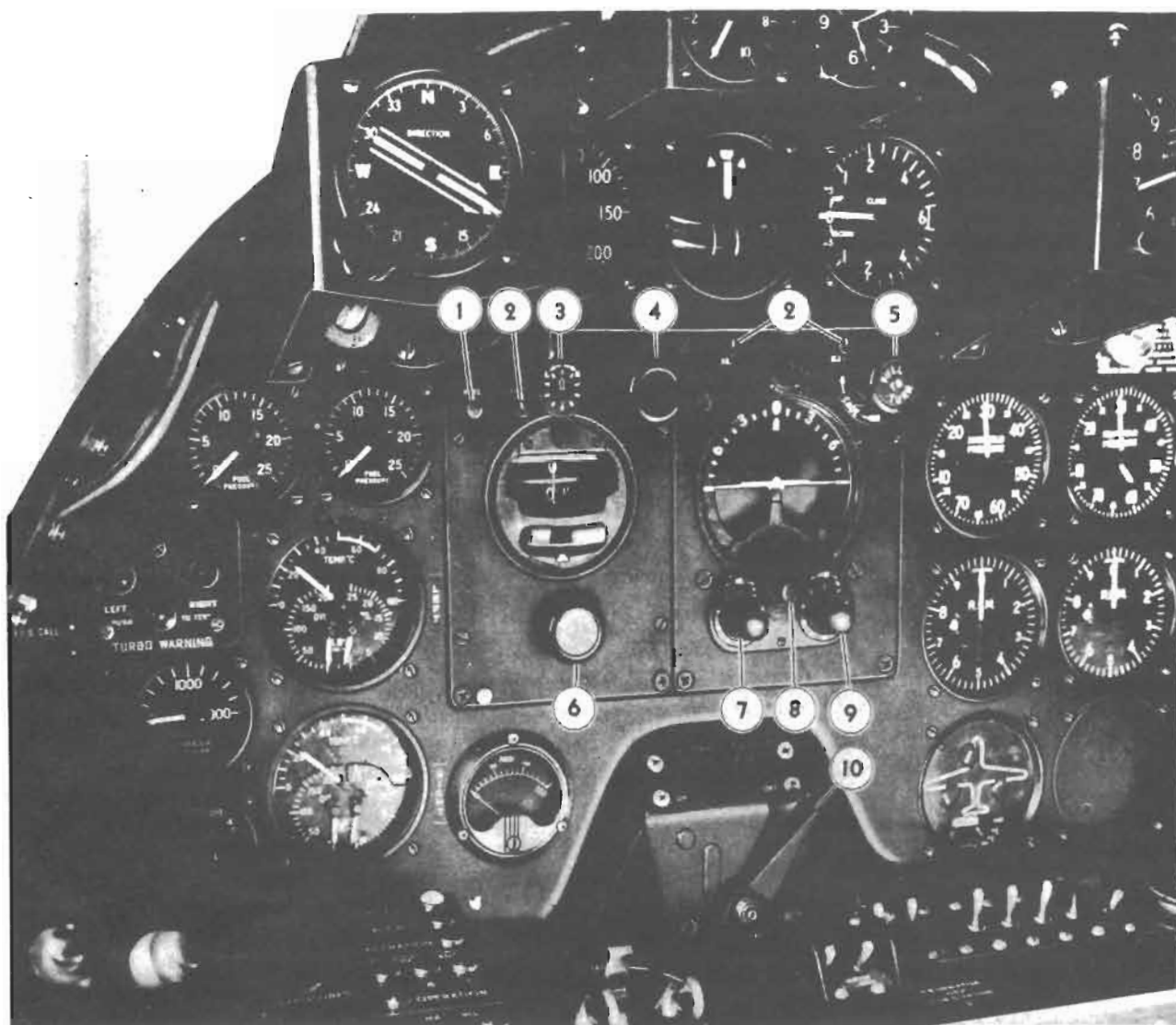
Cage all gyro instruments before engaging in acrobatics.

a. Although such maneuvers as loops, Immelmans, and rolls are permitted with this airplane, the pilot is cautioned to exercise extreme care in acrobatic maneuvers which require a downward recovery as the loss of altitude in downward recovery is very rapid. In general, acrobatics should *not* be attempted at altitudes below 10,000 feet until the pilot becomes familiar with the speed at which the airplane can gain and lose altitude.

18. DIVING.

CAUTION

Manifold pressure must be kept at or above 20 inches Hg. during extended shallow dives in order to prevent possible malfunctioning or misfiring of engines when throttles are opened after the pullout from the dive. In steep dives with dive recovery flaps extended the throttles may be closed completely without danger.



- | | |
|---|------------------------------------|
| 1. Rate control (ground adjustment only). | 6. Directional gyro caging knob. |
| 2. Speed valves—(3). | 7. Aileron trim knob. |
| 3. Course setting knob indicator. | 8. Miniature airplane control. |
| 4. Course setting knob. | 9. Elevator trim knob. |
| 5. Artificial horizon caging knob. | 10. Automatic pilot control valve. |

Figure 24—Automatic Pilot Controls

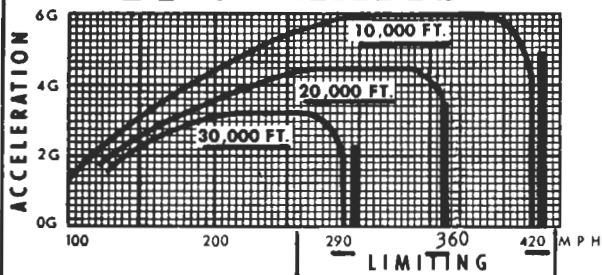
a. The diving speed is restricted as indicated on the placard (figure 25)—a copy of which is posted in the cockpit of each airplane. As the airplane approaches the critical speed, it becomes rapidly nose heavy and starts to buffet as if it were about to stall. If this condition is allowed to develop, the nose heavy condition will become more pronounced, and it will be very difficult to pull out.

b. The speed at which the above condition occurs depends upon the altitude and the acceleration (or G's)

which is being applied in a pullout. Figure 25 shows the placard consisting of three curves of indicated airspeed plotted against acceleration and indicates the safe range at the altitudes shown on each curve.

c. For example: If a straight dive is made in excess of 360 mph. (indicated) at 20,000 feet, the airplane will become nose heavy and start to buffet. Or if a pullout of over 4.5 G's is made at 300 mph at 20,000 feet, the same condition will be evident.

DIVE LIMITS



SPEED IN M.P.H. INDICATED

OUTSIDE OF ABOVE LIMITS BUFFETING AND DIVE TENDENCY MAY BE EXPECTED. IF EXPERIENCED, REDUCE ACCELERATION OR SPEED.

Figure 25 — Dive Placard

d. When the above conditions are noticed, the following action should be taken immediately.

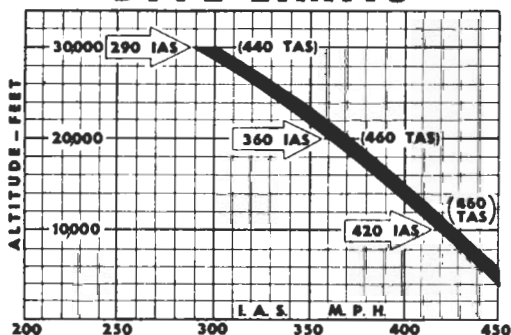
(1) In accelerated maneuvers (dive pullouts or steep turns) buffeting may be stopped by reducing the acceleration.

(2) In steady dives at high speed, buffeting may be stopped by reducing the airplane speed and pulling out using minimum acceleration. Use the elevator tab (figure 4-35) if necessary to assist recovery.

WARNING

Elevator tab must be used with care in order to prevent an extreme tail heavy condition after buffeting stops.

DIVE LIMITS



BUFFETING AND DIVE TENDENCY EXPERIENCED WHEN EXCEEDING ABOVE LIMITS MAY BE REDUCED BY EXTENDING DIVE RECOVERY FLAPS. DO NOT EXCEED PLACARD LIMITS MORE THAN 20 M. P. H. WITH DIVE RECOVERY FLAPS EXTENDED.

Figure 25A — Dive Placard

e. A new placard (figure 25A) will be installed in late airplanes and may be made retroactive to airplanes already in service. This new placard indicates the safe speed range at any altitude for one G flight. As the airplane approaches this critical one G condition, its ability to pull out is gradually reduced and at the critical speed, buffeting and nose heaviness will occur.

f. DIVE RECOVERY FLAPS.—P-38L and Later P38J airplanes are provided with dive recovery flaps to improve the dive recovery characteristics of the airplane. As described above, the airplane without these flaps becomes very nose heavy and starts to buffet above placard dive speeds. This condition is caused by a high speed stall and a consequent decrease in lift in the wing producing the nose heavy condition. The dive recovery flaps which are installed under the wings between the booms and the ailerons, restore the lift to this portion of the wing and thus cause the uncontrollable nose heaviness to occur at a higher speed. The flaps also add some drag to the airplane, which in conjunction with the higher allowable dive speed, permits safe dives at a much steeper diving angle. The dive recovery flaps should be extended before starting the dive or immediately after the dive has started before a buffeting speed has been reached. If the airplane is buffeting before the dive recovery flaps are extended, the buffeting will momentarily increase and then diminish. With these flaps extended, the nose heaviness is definitely reduced but the diving speed should never be allowed to exceed the placard by more than 15 or 20 mph. With the dive recovery flaps extended before entering the dive, angles of dive up to 45° may be safely accomplished. Without dive recovery flaps extended, the maximum angle for extended dives is 15°. Diving characteristics are better with power off than with power on.

WARNING

Although the dive recovery flaps greatly improve the diving characteristics of the airplane, dangerous buffeting and nose heaviness will still be encountered at diving angles above 45° if the diving speed is allowed to exceed the placard limits by more than 15 to 20 mph.

19. NIGHT FLYING.

a. Very little light need be used for normal cruising flight as all instrument dials are coated with phosphorescent paint. Fluorescent lights (figure 5-5) should be dimmed down until the instruments are barely visible. This will enable the eyes to become accustomed to the

darkness. The instrument glare shield should be installed for all night flying.

b. When more light is needed, cockpit lights (figure 5-11) may be turned on, or the spotlight (figure 4-1) may be focused on any point where local light is desired and adjusted to the required brilliancy.

c. Position lights are turned on by a switch (figure 5-6) on the main switch box.

d. Landing lights (figure 5-7) may be used for take-off and landing. They should not, however, be extended at any time when the airspeed is greater than 140 mph.

e. Recognition lights may be used as required.

20. APPROACH AND LANDING.

a. GENERAL.—The landing technique is similar to that of airplanes with conventional tricycle landing gear and the landing attitude is about the same; i.e., main wheels first, tail slightly down.

Note

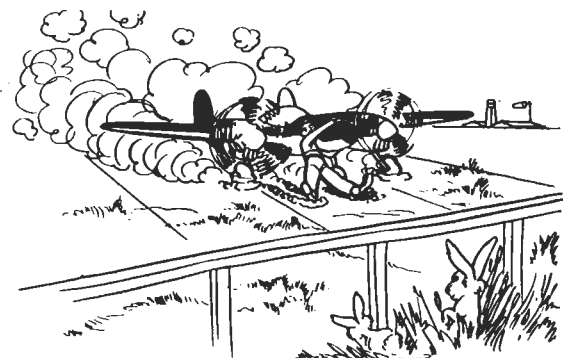
Extreme tail-low landings, possible only with flaps UP, may cause the fins to strike the runway.

(1) With the landing gear DOWN and flaps at MANEUVER, start the approach at 120 mph indicated air speed. When the approach is assured, put the flaps all the way down, come over the fence at 110 mph, flare off to about 80 mph and wait for contact.

(2) If, for some reason, the flaps cannot be lowered, land a little faster and allow for more flare-off and a flatter gliding angle.

b. NORMAL LANDING.

(1) Tank selector valves (figures 13-8 and 13-9) to MAIN or RESERVE, whichever contains the most fuel.



IF HE'DA LANDED AT 85, INSTEAD OF 120 . . .

(2) Mixture controls (figure 4-6) AUTO RICH.

(3) Propeller levers (figure 4-4) to about 2600 rpm position.

(4) Electric fuel pumps (figure 13-4) ON.

(5) Landing gear (figure 4-32) DOWN (not over 175 mph).

(6) Pump the brake pedals a few times to insure that brakes are working.

(7) Wing flaps (figure 7-2) DOWN (not over 150 mph).

Note

Lift the flap lever trigger through the quadrant notch to place lever to DOWN.

(8) Intercooler flaps (figure 5-12) OPEN unless operating in extreme low temperatures. (If installed.)

(9) Flaps UP before taxiing.

c. CROSS WIND LANDING.

(1) Same as the normal landing. The tricycle landing gear reduces danger of groundlooping from landing in a reasonably strong cross wind. If the drift seems excessive, the up-wind wing may be lowered until just before contact.

d. SINGLE ENGINE APPROACH AND LANDING.

CAUTION

Concentrate sharply on your approach—because once you have fully extended the flaps, and the landing gear or descended below five hundred feet, you cannot again circle the field and you must make a landing. If, however, the flaps are not fully extended and your elevation is still five hundred feet or more, and you want to go around again, proceed as follows before beginning to circle:

Apply as much power as can be held, at the same time retracting the landing gear, Accelerate to at least 160 mph, and Raise the flaps (unless they are already fully raised).

Do not make turns into the dead engine unless trim and speed have been established. It is recommended that the inexperienced pilot practice single engine landing by completely closing one throttle and setting the corresponding propeller lever to the "DECREASE RPM" (full rear) position. With this procedure the throttle engine will present nearly the same drag as a feathered propeller and if necessary both throttles may be opened to go around. Don't forget to push the propeller levers forward if it is necessary to go around.

- (1) Secure radio clearance for emergency landing.
- (2) Turn aileron control booster OFF to conserve hydraulic power for landing gear and flap operation.
- (3) Start approach allowing 1000 feet above field for each two miles away.
- (4) Extend landing gear at 160 mph.

Note

Allow more time for landing gear and flap extension when only one engine is operating.

- (5) Extend flaps to MANEUVER position at 140 mph.
- (6) Reduce power carefully as needed.
- (7) Neutralize rudder tab.
- (8) Continue approach at not less than 120 mph.
- (9) Do not extend full flaps until certain the airplane will make the field.

10) FURTHER INFORMATION.

(a) At rated power, 44" Hg, 2600 rpm, the airplane will barely hold altitude with landing gear extended and flaps UP.

(b) With landing gear extended the airplane will not hold altitude at any flap extension.

(c) Things to avoid:

1. Extension of landing gear or flaps except when necessary for landing.

2. Accelerating throttle rapidly from reduced power to full power.

3. Low flat approaches with landing gear and flaps fully extended attempting to drag the airplane into the field with power. The technique should be developed to be always reducing power on the approach and avoid being forced to apply excessive power at low airspeeds.

e. TAKE-OFF IF LANDING IS NOT COMPLETED.

(1) Open throttles to take-off stop and after propeller rpm has stabilized, push propeller control forward to take-off position.

(2) Retract landing gear.

CAUTION

Pull the airplane up in a climb sufficient to stay below 150 mph indicated airspeed until the flaps are retracted.

(3) Retract flaps and proceed in normal take-off technique.

21. STOPPING ENGINES.

a. If a cold weather start is anticipated, the oil system should be diluted before stopping the engines. Idle the engines until oil temperature is below 70°C (158°F),

adjust the throttles to approximately 1000 rpm, hold the oil dilution switches (figure 5-2) ON for the required time as shown in chart below, pull the mixture controls (figure 4-6) to IDLE CUT-OFF, and release the oil dilution switches after the engines stop turning.

Oil Dilution Time in Minutes

Outside Air Temperatures	Dilution Minutes
4° to -12°C (40° to 10°F)	3
-12° to -29°C (10° to -20°F)	5
-29° to -46°C (-20° to -50°F)	8

b. If the oil temperature is above 40°C (104°F) upon completion of the diluting process, the engine should be shut down and allowed to cool until the oil temperature is below 40°C. The engine should then be restarted and the same diluting process repeated. This will avoid the possibility of evaporation of the fuel due to the high temperature nullifying the effect of the first dilution.

c. If oil dilution is not necessary, set the mixture controls to IDLE CUTOFF at 1200 rpm, and slowly move the throttles fully open.

d. After the engines stop turning, turn the ignition switches (figure 4-11) OFF and leave the mixture in CUTOFF.

22. BEFORE LEAVING PILOT'S COMPARTMENT.

a. Turn all switches and valves OFF.

b. Lock the controls.

(1) To set surface controls lock (figure 7-10), put rudders in neutral, push the right end of the locking tube forward of the guide angle (figure 7-4) and place the left end of the locking tube in the clip provided (figure 4-3). Strap control wheel to center of surface control lock. (Later airplanes are not equipped with rudder locks.)

c. Set the parking brake:

(1) Apply the toe brakes, pull the parking brake handle (figure 4-14) out and release the toe brakes.

CAUTION

Do not set the parking brake while the brake discs are hot.

23. TIEING DOWN.

a. This airplane is tied down by means of ropes which are passed through tie down lugs on each landing gear shock strut, or around the strut if lugs are not provided, and tied to stakes firmly anchored to the ground. Stakes and ropes are provided in the mooring kit which is stowed in the baggage compartment.

b. If extremely high wind conditions are encountered, additional ropes may be tied around the tail end of each boom and secured to some solid anchor point.

Flight Operating Data



1. AIRSPEED AND ALTIMETER CORRECTION TABLE.

Calibrated* I.A.S.	Airplane I.A.S. Gear and Flaps UP	Airplane I.A.S. Gear and Flaps DOWN	*ALTIMETER INSTALLATION ERRORS Gear and Flaps UP (feet)				
			Sea Level	10000	20000	30000	40000
100	83	89	+110	+130	+200	+270	+430
120	106	116	+100	+125	+195	+265	+420
140	130	140	+90	+115	+175	+250	+380
160	153	...	+70	+100	+140	+190	+300
180	176	...	+50	+60	+90	+130	+190
200	200	...	0	0	0	0	0
220	226	...	-80	-110	-160	-230	...
240	248	...	-130	-170	-250	-330	...
260	269	...	-160	-230	-370	-430	...
280	291	...	-210	-290	-400	-550	...
300	312	...	-260	-360	-500
320	334	...	-325	-450	-620
340	356	...	-400	-540	-750
360	377	...	-500	-650

*Includes "Installation Errors" only. Does not include
"Instrument Errors" as obtained by the Field Test Set.

+ Add to altimeter reading
— Subtract from altimeter reading

AIRPLANE MODELS

SPECIFIC ENGINE

ENGINE MODELS

P-38H.....

V-1710-89 (RH).....

.....

V-1710-91 (LH).....

CONDITION	FUEL PRESSURE (LB./SQ. IN.)	OIL PRESSURE (LB./SQ. IN.)	OIL TEMP.		COOLANT TEMP.		MAX. PERMISSIBLE DIVING RPM:		
			°C	°F	°C	°F	CONDITION	ALLOWABLE OIL CONSUMPTION	
							NORMAL RATED (MAX. CONT.)	U.S.QT./HR.	IMP.PT./HR.
DESIRED	18-18	60-70	75-95	167-209	101-121	219-250		13	IMP.PT./HR.
MAXIMUM	18	85	95	203	125	257		9	IMP.PT./HR.
MINIMUM	12	55	40	104	85	185		7	IMP.PT./HR.
IDLING	9	15						1120	1120 OR 1100
								(W)	

SUPERCHARGER TYPE: EXHAUST DRIVEN TURBINE

FUEL GRADE: 100/130; SPEC. AN-F-28

OPERATING CONDITION	RPM	MANIFOLD PRESSURE (BOOST)	HORSE- POWER	CRITICAL ALTITUDE		BLOWER	USE LOW BLOWER BELOW:	MIXTURE CONTROL POSITION	FUEL FLOW (GAL./HR./ENG.)		MAXIMUM CYL. TEMP.		MAXIMUM DURATION (MINUTES)		
				WITH RAM	NO RAM				U.S.	IMP.	°C	°F			
TAKE-OFF	3000	54	1425		22,000**			AUTO-RICH	167				15		
WAR EMERGENCY	3000	60	1600	10,000	7,000*			AUTO-RICH	180				5		
MILITARY	3000	54	1425	24,900	22,000**			AUTO-RICH	167				15		
NORMAL RATED (MAX. CONT.)	2600	44	1100	34,000	31,600**			AUTO-RICH	113				NO LIMIT		
MAXIMUM CRUISE	2300	35	795	38,000	36,700**			AUTO-LEAN	63				NO LIMIT		
MINIMUM SPECIFIC CONSUMPTION	2000'	32	645	30,000				AUTO-LEAN	51				NO LIMIT		
	1700	30	515	20,000					40						
	1600	30	485	15,000					37						
	1500	28	425	10,000					34						
	1600	27	400	5,000					31						
REMARKS:	NEVER EXCEED THE FOLLOWING MANIFOLD PRESSURE AT THE ALTITUDE SHOWN REGARDLESS OF ENGINE R.P.M.				ALTITUDE		7,000 TO 10,000		25,000		30,000		35,000		40,000
							CLIMB	LEVEL FLIGHT	CLIMB	LEVEL FLIGHT	CLIMB	LEVEL FLIGHT	CLIMB	LEVEL FLIGHT	
							54	60	52	54	49	51	44	46	
							54	54	52	54	48	51	43	43	

AIRPLANE MODELS

SPECIFIC ENGINE
FLIGHT CHART

ENGINE MODELS

P-38J.....

V-1710-89 (R.H.).....

F-5B.....

V-1710-91 (L.H.).....

CONDITION	FUEL PRESSURE (LB./SQ. IN.)	OIL PRESSURE (LB./SQ. IN.)	OIL TEMP.		COOLANT TEMP.		MAX. PERMISSIBLE DIVING RPM:		
			°C	°F	°C	°F	CONDITION	ALLOWABLE OIL CONSUMPTION	
							NORMAL RATED (MAX. CONT.)	U.S.QT./HR.	IMP.PT./HR
DESIRED	16-18	60-70	75-95	167-203	101-121	214-250		13	IMP.PT./HR
MAXIMUM	18	85	95	203	125	257		9	IMP.PT./HR
MINIMUM	12	55	40	104	85	185		7	IMP.PT./HR
IDLING	9	15						1120	1120 OR 1100

SUPERCHARGER TYPE: EXHAUST DRIVEN TURBINE FUEL GRADE: 100/130; SPEC. AN-F-28

OPERATING CONDITION	RPM	MANIFOLD PRESSURE (BOOST)	HORSE- POWER	CRITICAL ALTITUDE		BLO WERS	USE LOW BLOWER BELOW:	MIXTURE CONTROL POSITION	FUEL FLOW (GAL./HR/ENG.)		MAXIMUM CYL. TEMP.		MAXIMUM DURATION (MINUTES)
				WITH RAM	NO RAM				U.S.	IMP.	°C	°F	
TAKE-OFF	3000	54	1425		26,600**			AUTO-RICH	167				15 -
WAR EMERGENCY	3000	60	1600	28,700	25,800*			AUTO-RICH	180				5
MILITARY	3000	54	1425	29,000	26,600**			AUTO-RICH	167				15
NORMAL RATED (MAX. CONT.)	2600	44	1100	33,800	31,200**			AUTO-RICH	113				NO LIMIT
MAXIMUM CRUISE	2300	35	795	37,800	36,500**			AUTO-LEAN	63				NO LIMIT
MINIMUM SPECIFIC CONSUMPTION	2000 1700 1600 1600 1600	30 31 31 29 28	620 545 525 455 425	30,000 20,000 15,000 10,000 5,000				AUTO-LEAN	48 42 40 35 33				NO LIMIT

REMARKS:	NEVER EXCEED THE FOLLOWING MANIFOLD PRESSURES AT THE ALTITUDE SHOWN REGARDLESS OF ENGINE R.P.M.	ALTITUDE											
		UP TO 26,000'			30,000 FT.				35,000 FT.			40,000 FT.	
		LEVEL		CLIMB	LEVEL		CLIMB	LEVEL		CLIMB	LEVEL		
		FLIGHT	FLIGHT		FLIGHT	FLIGHT		FLIGHT	FLIGHT				
		60	60		53	57	45	48		37	39		
		54	54		49	52	41	43		33	34		

*WAR EMERGENCY POWER
**MILITARY POWER OR LESS

AIRPLANE MODELS

SPECIFIC ENGINE

ENGINE MODELS

V-1710-111 (RH).....

V-1710-113 (LH).....

P-38L

CONDITION	FUEL PRESSURE (LB./SQ. IN.)	OIL PRESSURE (LB./SQ. IN.)	OIL TEMP.		COOLANT TEMP.		MAX. PERMISSIBLE DIVING RPM:
			°C	°F	°C	°F	
DESIRED	16-18	60-70	75-95	165-205	101-121	214-250	ALLOWABLE OIL CONSUMPTION
							NORMAL RATED (MAX. CONT.)
MAXIMUM	18	85	95	203	125	257	13. U.S.QT/HR. IMP.PT/HR
							MAX. CRUISE
							8. U.S.QT/HR. IMP.PT/HR
MINIMUM	12	55	40	104	85	185	7. U.S.QT/HR. IMP.PT/HR
							MIN. SPECIFIC
IDLING	9	15					OIL GRADE: (S) 1120. (NW) 1120. OR. 1100. . .

SUPERCHARGER TYPE: EXHAUST DRIVEN TURBINE

FUEL GRADE: 100/130; SPEC. AN-F-28

OPERATING CONDITION	RPM	MANIFOLD PRESSURE (BOOST)	HORSE- POWER	CRITICAL ALTITUDE		BLOWER	USE LOW BLOWER BELOW:	MIXTURE CONTROL POSITION	FUEL FLOW (GAL./HR./ENG.)		MAXIMUM CYL. TEMP.		MAXIMUM DURATION (MINUTES)
				WITH RAM	NO RAM				U.S.	IMP.	°C	°F	
TAKE-OFF	3000	54	1425		26,600**			AUTO-RICH	167				15
WAR EMERGENCY	3000	60	1600	28,700	25,800*			AUTO-RICH	180				5
MILITARY	3000	54	1425	29,000	26,600**			AUTO-RICH	167				15
NORMAL RATED (MAX. CONT.)	2600	44	1100	33,800	31,200**			AUTO-RICH	113				NO LIMIT
MAXIMUM CRUISE	2300	35	795	37,800	36,500**			AUTO-LEAN	63				NO LIMIT
MINIMUM SPECIFIC CONSUMPTION	2000	30	620	30,000				AUTO-LEAN	48				NO LIMIT
	1700	31	545	20,000					42				
	1600	31	525	15,000					40				
	1600	29	455	10,000					35				
1600	28	425	5,000						33				

REMARKS: NEVER EXCEED THE FOLLOWING MANIFOLD PRESSURES AT THE ALTITUDE SHOWN REGARDLESS OF ENGINE R.P.M.

***WAR EMERGENCY POWER
***MILITARY POWER OR LESS

OPERATING LIMITS USING GRADE 91 FUEL

NOTE—USE GRADE 91 FUEL ONLY IF MODIFICATIONS ACCORDING TO TO-02-5A-66 HAVE BEEN COMPLIED WITH.

AIRPLANE MODELS:				ENGINE MODELS:			
P-38H	P-38J	P-38L	F-5B	V-1710-89	V-1710-111	V-1710-91	V-1710-113
FUEL: GRADE 91/96 SPEC. AN-F-26							
OPERATING CONDITIONS	MANIFOLD PRESSURE (Max. In. Hg)	R. P. M. (Max.)	HORSE POWER	MIXTURE CONTROL POSITION	BLOWER	MAXIMUM DURATION (Min.)	
TAKE-OFF	47	3000	1325	AUTO RICH	SEA LEVEL	5	
	40	3000	1098	AUTO RICH	25,000 FT. ALTITUDE	5	
NORMAL RATED	38	2600	1000	AUTO RICH		NO LIMIT	
MAXIMUM CRUISING	32	2300	755	AUTO RICH		NO LIMIT	
DESIRED CRUISING	28	2200	605	AUTO LEAN		NO LIMIT	
REMARKS: 1. Reduce above manifold pressures 2" Hg for each 5°C (9°F) above 43°C (110°F) carburetor air temperature. 2. Reduce above manifold pressures 2" Hg for each 1,000 ft. above 27,000 ft. until 20" Hg is reached.							

Emergency Operating Instructions

1. ENGINE FAILURE DURING FLIGHT.

a. FAILURE OF ONE ENGINE.

(1) **PERFORMANCE.**—The airplane flies well on one engine. Using normal rated power, it will climb to about 26,500 feet, and can be flown at more than 255 mph (true speed) in level flight at 20,000 feet.

(2) FEATHERING EMERGENCY.

(a) Reduce the power from the live engine if necessary to maintain directional control. This should not be necessary if the indicated airspeed is 125 mph or more.

(b) Apply all the power to the good engine that can be held, preventing yaw at all times.

(c) Hold 125 mph or more (at least 160 mph preferred).

(d) Release droppable fuel tanks, bombs, or chemical tanks, immediately.

(e) Trim rudder tab slowly to take pressure off rudder pedal.

(f) Carefully move mixture control of bad engine to IDLE CUT-OFF.

(g) Carefully select propeller feathering switch (figure 4-13) of bad engine and feather propeller.

WARNING

If the propeller does not feather, then attempt to feather it by holding the selector switch (figure 4-5) in the DEC RPM position. If the propeller still will not feather then it is desirable to fly at a low airspeed (130 to 140 mph) to keep the propeller windmilling at the lowest possible rpm.

(b) Turn off electric fuel pump switch and fuel tank selector valve control of failed engine.

(i) Close coolant and oil cooler scoops of failed engine.

(j) If the left engine has failed, and consequently the generator has stopped, take action indicated under ELECTRICAL FAILURE, Section IV, paragraph 10. (This is not applicable to F-5B, P-38L, and late P-38J airplanes which have a generator on each engine.)

(k) Climb to safe altitude and reduce power to maintain 160 mph indicated airspeed.

(l) Return to field and fly around until familiar with airplane's behavior on single engine.

(3) SINGLE ENGINE APPROACH AND LANDING.

CAUTION

Concentrate sharply on your approach—because once you have fully extended the flaps, and the landing gear or descended below 500 feet, you cannot again circle the field and you must make a landing. If, however, the flaps are not fully extended and your elevation is still 500 feet or more, and you want to go around again, proceed as follows before beginning to circle:

(1) Apply as much power as can be held, at the same time retracting the landing gear,

(2) Accelerate to at least 160 MPH, and

(3) Raise the flaps.

It is recommended that the inexperienced pilot practice single engine landing by completely closing one throttle and setting the corresponding propeller lever to the "DECREASE RPM" (full rear) position. With this procedure the throttled engine will present nearly the same drag as a feathered propeller and if necessary both throttles may be opened to go around. Don't forget to push the propeller levers forward if it is necessary to go around.

Note

URNS CAN BE MADE SAFELY IN EITHER DIRECTION AS LONG AS AIRSPEED IS HELD CONSTANT ABOVE CRITICAL SINGLE ENGINE SPEED, AND AIRPLANE IS PROPERLY TRIMMED.

(a) Secure radio clearance for emergency landing.

(b) Turn aileron control booster OFF to conserve hydraulic power for landing gear and flap operation.

(c) Start approach allowing 1000 feet above field for each two miles away.

(d) Extend landing gear at 160 mph.

Note

Allow more time for landing gear and flap extension when only one engine is operating.

(e) Extend flaps to MANEUVER position at 140 mph.

(f) Reduce power carefully as needed.

(g) Neutralize rudder tab.

(h) Continue approach at not less than 120 mph.

(i) Do not extend full flaps until certain the airplane will make the field.

(4) FURTHER INFORMATION.

(a) At rated power, 44" Hg. 2600 rpm, the airplane will barely hold altitude with landing gear extended and flaps up.

(b) With landing gear extended the airplane will not hold altitude at any flap extension.

(c) Things to avoid:

1. Extension of landing gear or flaps except when necessary for landing.

2. Accelerating throttle rapidly from reduced power to full power.

3. Low flat approaches with landing gear and flaps fully extended attempting to drag the airplane into the field with power. The technique should be developed to be always reducing power on the approach and avoid being forced to apply excessive power at low airspeeds.

(5) FEATHERING—PRACTICE.

(a) Close throttle.

(b) Mixture—IDLE CUT-OFF.

(c) Move propeller feathering switch to feathering position.

CAUTION

On all airplanes, except F-5B, P-38L and late P-38J airplanes which have a generator on each engine, shut down the right-hand engine so that the generator which is on the left engine will remain in operation.

(6) UNFEATHERING IN FLIGHT.

(a) Propeller control (figure 4-4) DEC RPM (full rearward).

(b) Throttle 1/10 to 1/4 open.

(c) Return propeller feathering switch to NORMAL position.

(d) Lift the guard on the propeller selector switch of the feathered propeller.

Note

It is recommended that the above operations be performed immediately after feathering is completed so that the engine may be started quickly in case the live engine should fail during practice.

(e) Hold the propeller selector switch (figure 4-5) in INC RPM until the engine is turning 600 to 800 rpm then place it in AUTO CONSTANT SPEED. Place mixture control in AUTO RICH. The engine should start.

(f) Warm up the engine before operating at full power.

b. FAILURE OF BOTH ENGINES.

(1) Drop external tanks or bombs.

(2) Turn fuel selector valves OFF.

(3) Set mixture controls to IDLE CUTOFF.

(4) Turn ignition OFF.

(5) Turn battery switch OFF.

(6) Release the cockpit canopy and roll down both side windows.

(7) Extend flaps by use of hand pump if there is sufficient time.

(8) Leave the landing gear up.

(9) Make a normal approach at 8 or 10 mph over the stalling speed and set the airplane on the ground slightly before the stall is reached.

2. FIRE.

a. There are no fire extinguishers installed in this airplane. If an engine fire occurs, shut off tank selector valve to that engine, turn boost pump OFF, and move mixture control to IDLE CUTOFF.

3. EMERGENCY EXIT.

a. RECOMMENDED METHOD. — Slow down as much as possible, (below 200 mph) and trim the airplane in an approximately level attitude. If time permits head airplane towards an unpopulated area. Pull the emergency hatch release control (figure 23-4) to release the top hatch, crank or push either side window down, crawl out and slide off the wing head first.

b. ALTERNATE METHOD. — If it is still possible to control the airplane, turn the airplane upside down, unhook the safety belt, and fall out.

4. DROPPABLE FUEL TANK OR BOMB EMERGENCY RELEASE.

a. A droppable tank or bomb emergency release is installed in later airplanes. It is located either behind and to the left of the pilot's seat or directly in front of the pilot's seat.



BE CAREFUL WHERE YOU DROP YOUR TANKS !!

5. EMERGENCY WING FLAP OPERATION.

a. If the pressure from the engine driven hydraulic pump fails to extend the flaps, they may be extended as follows:

(1) Turn coolant override switches (figure 5-13) OFF.

(2) Turn the aileron boost shutoff valve (figure 9-5) OFF.

(3) Move flap control (figure 7-2) to MANEUVER or DOWN.

(4) Turn source selector valve handle (figure 17-4) to UP. (This is the normal position.)

(5) Operate the hand pump (figure 17-2) until the desired flap extension is obtained.

b. If the above system fails to extend the flaps, leave the control at DOWN while extending the landing gear. Oil from the return side of the landing gear cylinders may fill the system enough to cause the flaps to operate.

Note

The auxiliary system should be used for flap extension because in some cases there may not be sufficient fluid to extend both landing gear and flaps, and the landing gear may be extended with the emergency system fluid.

CAUTION

If one engine has failed in addition to the hydraulic system failure, remember that the airplane cannot maintain level flight on one engine with landing gear and flaps extended. Under these conditions it may be desirable to land with flaps UP.

6. EMERGENCY LANDING GEAR OPERATION.

a. Check to see that the coolant override switches and the aileron booster shut-off valve are in the OFF position. Place all controls in the normal position for landing gear extension and operate the hand pump (figure 17-2).

Note

Nose gear door must open and nose gear must unlock before any pressure will flow to the main gear door cylinders.

b. If, after considerable pumping, no reading is given on the position indicator operate the emergency system as follows:

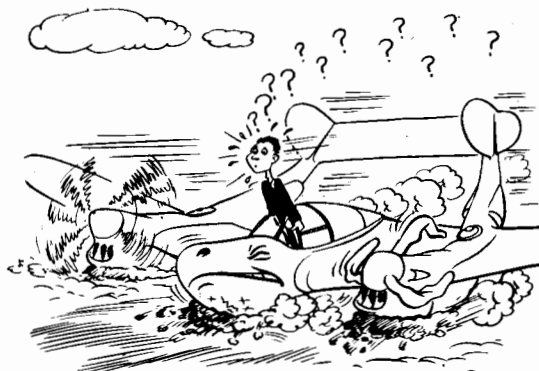
(1) Source-selector valve handle; (figure 17-4) break safety wire and push DOWN.

(2) Bypass valve (figure 17-3); break safety wire and CLOSE tightly. (On later airplanes, the bypass valve is controlled by the source selector valve control. If bypass valve is not provided, disregard operation (2).)

(3) Check landing gear control DOWN and operate the hand pump while yawing the airplane from side to side.

Note

This system opens the landing gear doors by forcing them with the wheels.



YOU HAVE AN EMERGENCY EXTENSION SYSTEM...
WHY NOT LEARN TO USE IT ?

7. EMERGENCY LANDING WITH WHEELS RETRACTED.

a. When a belly landing is necessary, it should be made without external tanks. Use 1/2 to full flaps.

Note

When imminent belly landing is due to apparent hydraulic failure, unless forced down by lack of fuel or approaching darkness, don't give up until all methods of landing gear extension have been tried thoroughly and exhaustively.

b. Make a normal approach at 8 or 10 mph over the stalling speed and set the airplane on the ground slightly before the stall is reached. Set the mixture control to CUTOFF and turn the battery switch OFF before contact.

8. EMERGENCY LANDING IN WATER (DITCHING).

a. Unless the water is very smooth, it will probably be more desirable to bail out of the airplane than to try to land it on the water.

b. If a water landing is necessary, preparations for abandoning the airplane should be made while still in the air. Release bombs or droppable tanks and the top hatch, and push down both side windows. Leave the shoulder harness on to prevent the shock of landing from throwing the head forward into the bullet-proof glass and the gunsight.

(1) Since a much flatter approach can be made with power on, the landing should be made before the fuel has been completely used up.

(2) Wind and surface conditions should be noted so that the approach may be made along the swell and as near into the wind as possible.

c. Make contact with landing gear and flaps UP, (water landings with gear down are invariably fatal) and at an airspeed slightly above the stalling point.

d. After landing, release the safety belt and swim clear of the airplane before it sinks. The airplane can not be expected to stay afloat once it comes to rest. The seat back cushion provided with the airplane is also a life preserver.

9. ICING CONDITIONS.

a. Pitot heat (figure 5-14) ON.

b. The formation of carburetor ice is unlikely due to

the injection type carburetors and the heating effect of the turbo-superchargers. It is possible, however, that ice could form while flying at low powers in a humid atmosphere.

c. If icing conditions are present during a landing approach, move the throttles occasionally to prevent ice from freezing them in a closed position. Extend landing gear, lower flaps to maneuvering position, and make the approach under partial power.

d. If carburetor ice forms in cruising flight, it may be removed by increasing the power boldly and putting the airplane into a steep climb. Intercooler shutters (on P-38J, P-38L and F-5B airplanes only) should be closed as far as possible without exceeding the maximum 45°C (113°F) carburetor air temperature.

e. Ice formation on the windshield may be removed by turning the cockpit heat ON and directing the flexible heater tube to the desired point on the glass.

10. ELECTRICAL FAILURE.

a. Electrical failure may be indicated by a zero reading on the voltmeter (figure 5-8) and the ammeter (figure 8-12) and by rapidly diminishing battery power (the first indication of a low battery will be given by failure of the propeller governors to hold the proper rpm). When the above conditions are noticed:

(1) Set the propeller selector switches to FIXED PITCH.

(2) Set the oil cooler flap switches to OFF.

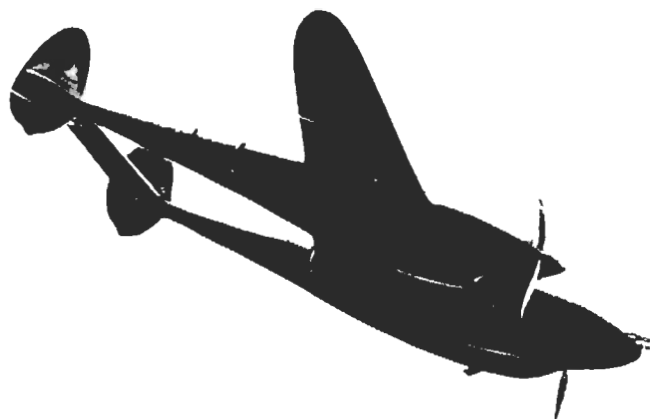
(3) Restrict the use of all lights and radio.

(4) If practicable, turn the battery switch OFF until electrical power is needed.

(5) If it is necessary to land with the propeller selector switches on MANUAL, the following setting should be made to insure that sufficient power will be available and that the engines will not overspeed in the event of mislanding. Make this adjustment, if possible, while there is still sufficient battery power to operate the propellers.

(*a*) Altitude—Not over 5,000 feet above the airport.

(*b*) Adjust the throttles and propeller selector switches to obtain 2600 rpm and 25" Hg. manifold pressure at an approximate airspeed of 180 mph in level flight.



Operational Equipment

1. HEATING AND VENTILATING.

a. COCKPIT HEAT.—Cockpit heat is supplied by an intensifier tube in the right engine exhaust (both engines on late airplanes) and controlled by a knob (figure 7-3) on the windshield support. Heat outlets are arranged to supply warm air to the windshield and removable hatch. The foot heat outlet may be closed off by operating the heat control (figure 26-5) on the floor under the right foot. On later airplanes, a heated flying suit plug and rheostat are located on the left side of the cockpit.

b. VENTILATING AIR.—Ventilating air enters from the left wing fuselage fillet. The rate of flow may be varied by rotating the ventilator (figure 4-30) as desired.

c. ARMAMENT OR CAMERA COMPARTMENT HEAT.—Armament or camera compartment heat is supplied by an intensifier tube in the left engine exhaust on early airplanes. The heat control (figure 4-23) on the left windshield support is used to turn the heat ON and OFF. On late airplanes the left engine heat has been diverted to the cockpit. Electric gun heaters are used on these airplanes and are turned on by a switch on the main switch box.

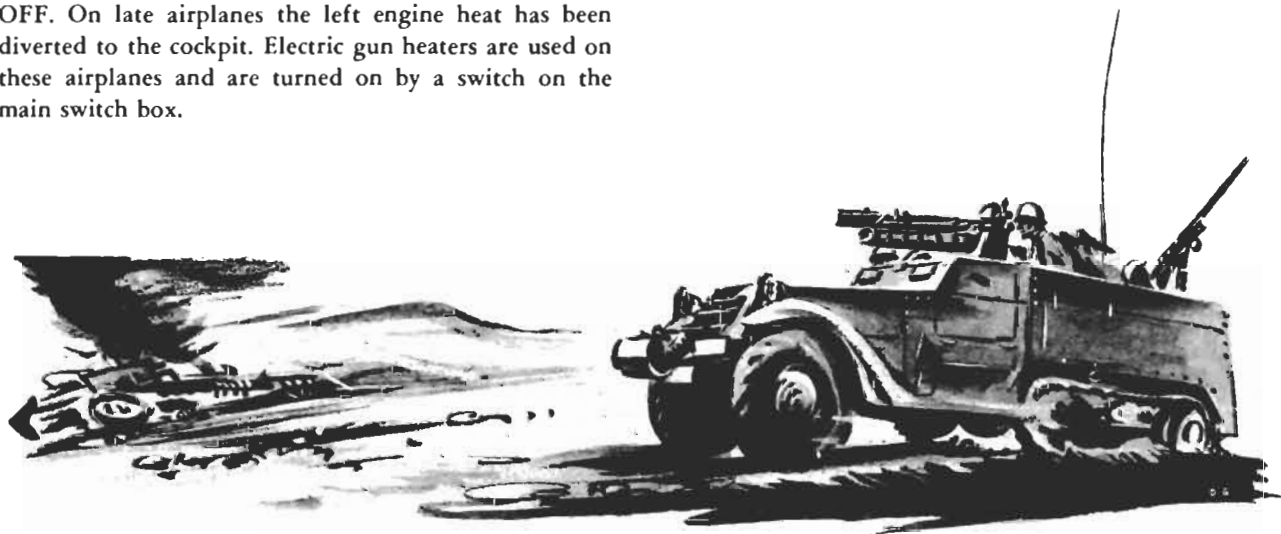
2. OXYGEN SYSTEM.

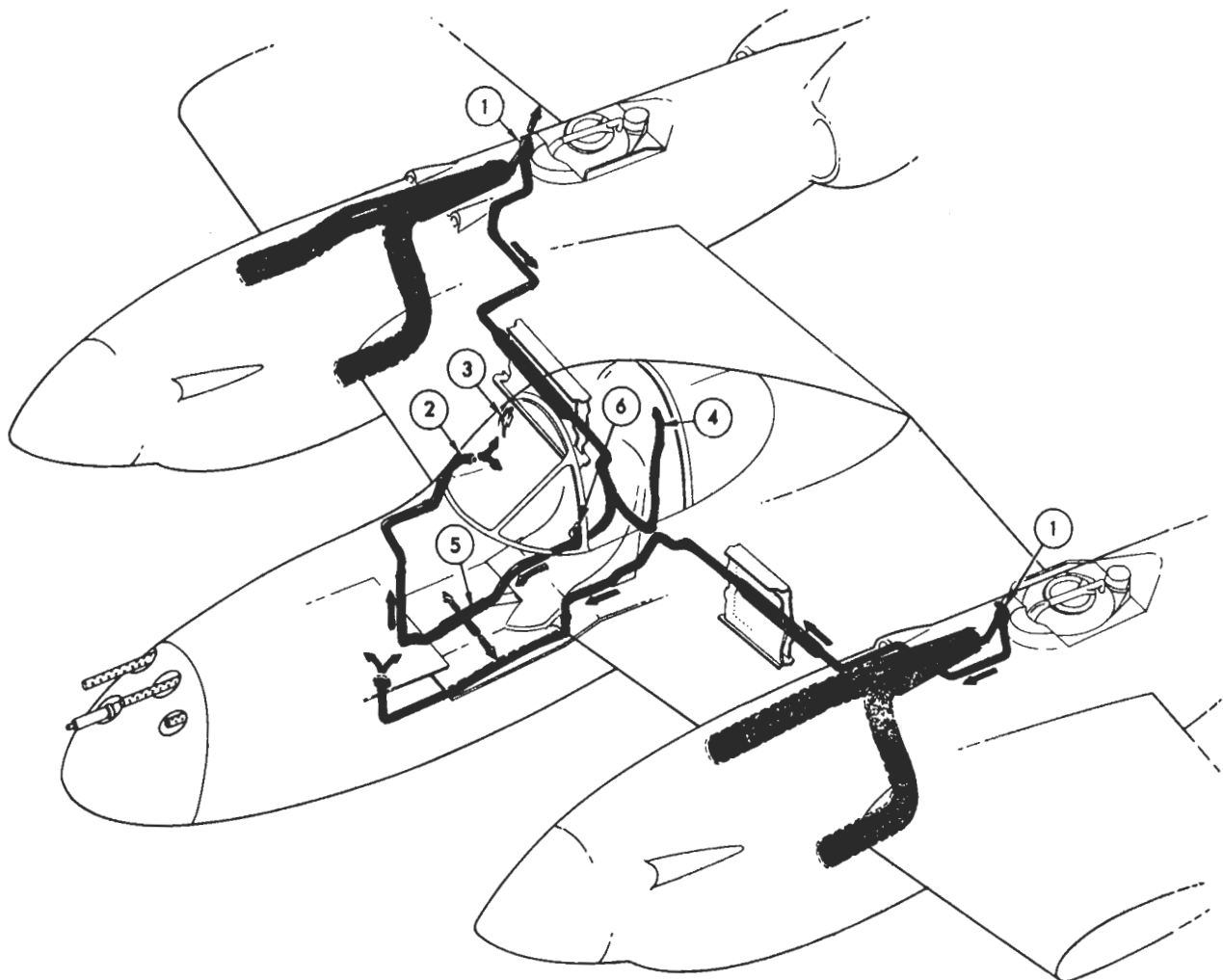
a. Full pressure in the three oxygen bottles is 400 to 450 lb/sq in.




Altitude 1,000 Feet	Hours of Supply	
	Auto Mix—ON *	Auto Mix—OFF
0	8.8	1.6
5	8.3 **	1.9
10	11.0 14.1	2.2
15	11.0 8.1	2.9
20	11.2 7.0	3.7
25	6.3 6.1	4.6
30	6.4 6.4	6.1
35	8.4 8.4	8.4
40	12.0 12.0	12.0

* Pioneer regulator

** Air Reduction Sales Regulator





-  Exhaust Manifold
-  Cockpit Heat Lines
-  Armament Heat Lines
(Diverted to cockpit on late airplanes)

1. Heat by-pass (shut-off) valves, (2)
2. Windshield defroster tube.
3. Cockpit heat control.
4. Flexible hatch defroster tube.
5. Foot heat shut-off valve.
6. Gun (or camera) heat control.
(Cockpit heat control on late airplanes.)

Figure 26 — Heating System Diagram

b. Fit the demand-type mask snugly to the face and test the fit by pinching the tube and sucking lightly. The mask should collapse. Clip the mask tube to the clothing allowing for all necessary head movement without pulling the mask away from the face.

WARNING

It must be remembered that the suction of breathing is used to open the oxygen valve, and that any leakage in the mask will cause this valve to become inoperative and will stop the flow of oxygen.

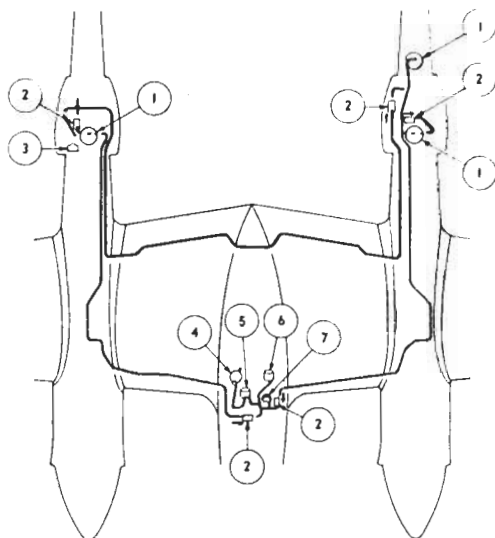
(c) Make certain the knurled collar at the outlet end of the regulator is tight. Examine top diaphragm to see that it is not ruptured or distorted.

(d) Turn emergency knob (Figure 27-4) "ON" to check the flow. Check the pressure gage to see that there is no perceptible pressure drop. Turn emergency knob "OFF" and ascertain that it does not leak. Leave it in this position.

(e) Turn the auto-mix (Figure 27-3) to "OFF". Note on flow indicator (Figure 27-1) that on inhalation the top diaphragm goes down and that nearly 100 percent oxygen is received. Turn the auto-mix to "ON" and note that there is little or no indication of oxygen flow on the indicator. Leave in this position.

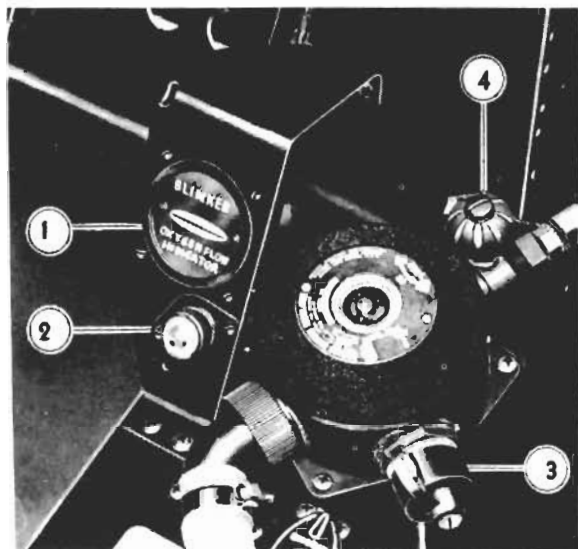
(f) Check pressure of the system. It must not be less than 365 lbs./sq.in. Before take-off, make certain that the pressure gage (Figure 4-34) shows sufficient oxygen

■ Oxygen Lines
■ Refill Lines



1. Oxygen bottles.
2. Check valves.
3. Refill connection.
4. Demand type oxygen regulator.
5. Oxygen flowmeter.
6. Oxygen pressure gage.
7. Pressure warning light.

Figure 28 — Oxygen System Diagram



1. Oxygen flow indicator
2. Oxygen pressure warning light.
3. Auto-mix lever.
4. Emergency knob.

Figure 27 — Oxygen Controls

supply for the mission. The oxygen pressure warning unit (Figure 27-2) indicates when the pressure drops below approximately 100 lbs./sq.in.

(g) With the auto-mix and the emergency controls both turned OFF, pure oxygen will be supplied as required by normal breathing. This setting, however, is wasteful of oxygen and should be used only in an emergency, when denitrogenizing for high altitude flight has been accomplished, or during night flights for improved vision.

3. COMMUNICATIONS EQUIPMENT.

a. These airplanes may be equipped with an SCR-522 command set as shown in figure 7 or an SCR-274N command set illustrated in figure 29. An SCR-695-A radio and a Detrola model 438 beacon receiver have been provided. Provisions are made for the installation of a BC-608 contactor.

b. OPERATION OF THE SCR-522 RADIO.

(1) Plug the headphones and microphones into the jack (figure 17-5).

(2) Check the generator switch (figure 5-16 or 8-11) and the battery switch (figure 5-15) ON.

(3) Set the selector switch (figure 7-9) to REM. The T and R positions of this switch are not to be used.

(4) Push the A, B, C, or D button (figure 7-7), depending on which pre-tuned frequency is to be used, and wait about one minute for warm-up. Lights beside each button indicate which button has been pushed. The lever (figure 7-6) beside the OFF button may be used to dim the indicator lights.

(5) Press the microphone button and speak slowly and clearly.

(6) Release the microphone button to receive.

Note

The T and R positions are provided for use on installations employing a separate radio operator. Transmission is impossible with the switch in the R position; reception is impossible with the switch in the T position. The lock lever (figure 7-8) spring loads the selector switch to the R position and prevents operation in the REM position.

(7) To turn off the equipment, push the OFF button (figure 7-5).

Note

This radio is not equipped with a volume control. Most installations have a remote volume control installed on the junction box on the right hand side of the cockpit over the hydraulic hand pump.

c. OPERATION OF THE SCR-274N RADIO.—

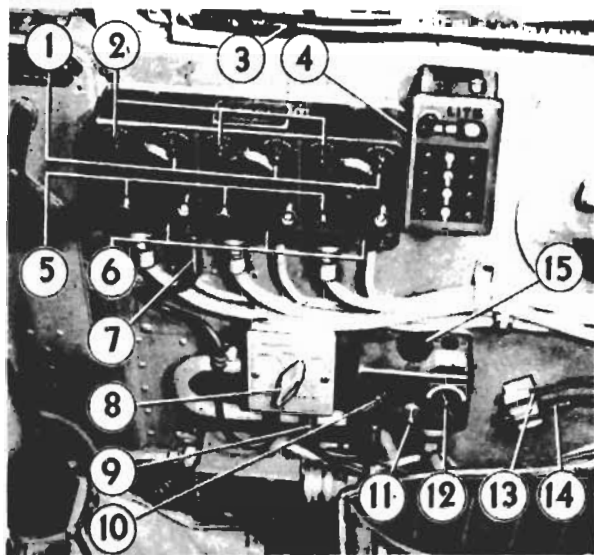
There are three separate receivers, and any two of four available transmitters may be installed.

<i>Receiver Frequencies</i>	<i>Transmitter Frequencies</i>
190 to 550 kilocycles	3 to 4 megacycles
3 to 6 megacycles	4.0 to 5.3 megacycles
6 to 9.1 megacycles	5.3 to 7.0 megacycles
	7.0 to 9.1 megacycles

The receivers may be tuned in flight; the transmitters are pre-tuned to the proper frequency on the ground.

(1) RECEIVING.

(a) Plug the headphones into the jack JK-26 (figure 29-14) which is, in turn, plugged into the bottom of the filter box (figure 29-9).



1. Main power switch (receivers).
2. Headset selector switches.
3. Surface controls lock.
4. Recognition light switch box.
5. Volume controls.
6. Tuning cranks.
7. Tel. plug (in B jack).
8. Radio range filter switch.
9. Headset jack.
10. Emission selector switch.
11. Main power switch (transmitter).
12. Transmitter selector switch.
13. Microphone jack (stowed).
14. Headset extension cord jack (stowed).
15. Transmitter key.

Figure 29 — SCR 274N Radio Controls

(b) Determine the frequency to be used and turn the main power switch (figure 29-1) for that receiver to MCW or CW, depending on the type of reception desired.

(c) For normal reception, set the range selector switch (figure 29-8) to BOTH. To receive radio range without voice interference, set the selector switch to RANGE. To eliminate radio range interference from a station broadcasting range and voice signals on the same frequency, set the selector switch to VOICE.

(d) Note which jack (A or B) the TEL plug (figure 29-7) is in and turn the headset selector switch (figure 29-2) of the receiver being used to corresponding position (A or B).

(e) Adjust the volume (figure 29-5) to obtain a light frying noise and tune in the desired station using the crank (figure 29-6).

Note

Two or three receivers may be turned on simultaneously by following the instructions contained in paragraphs (d) and (e) above for each receiver.

(2) TRANSMITTING.

(a) Select the desired frequency with the transmitter selector switch (figure 29-12).

(b) Plug the microphone into the jack JK-48 (figure 29-13) which is plugged into the bottom of the transmitter control box.

(c) Turn the main power switch (figure 29-11) ON and allow 15 seconds for the transmitter to warm up.

(d) Turn the emission selector (figure 29-10) to CW, MCW, or TONE as desired.

(e) If MCW was selected, press the microphone button and talk slowly and clearly.

(f) If CW or TONE is selected, operate the key (figure 29-15).

d. DETROLA MODEL 438 BEACON RECEIVER.— This receiver is in addition to the SCR-522 or the SCR-274N radio when the airplane leaves the factory. If it is still installed:

(1) Plug headphones into the jack (figure 17-5) which is plugged into the face of the radio panel.

(2) Turn the volume control (figure 7-15) up until the background noise is heard.

(3) Tune to the desired frequency with the tuning knob (figure 7-14).

(4) Frequencies covered are 200 to 400 kilocycles.

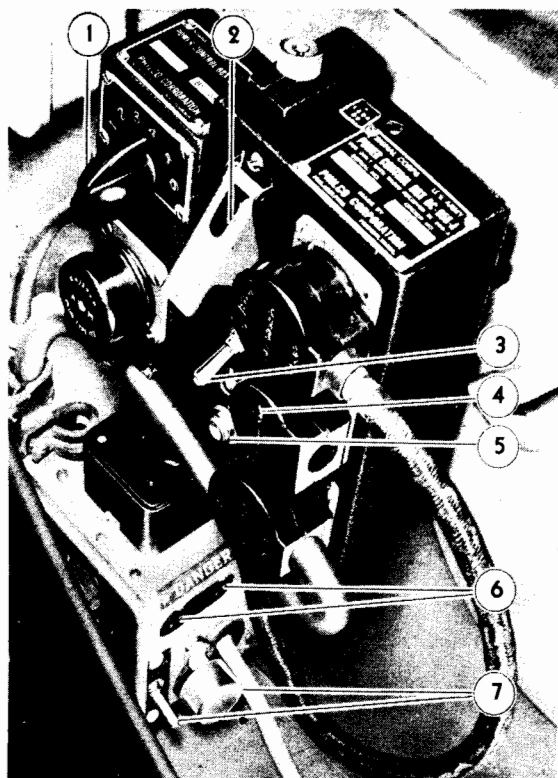
e. SCR-695-A RADIO.

(1) Before take-off insert the destructor plug in the radio located just forward of, and accessible through the baggage compartment door on the right boom. The destructor plug socket is on the right side, set apart from the group of three sockets on the left side of the radio.

DANGER

Do not insert the destructor plug if the red warning lights, located over the radio set in the boom, are lighted. Make sure that the warning light system has been checked and is in proper working order.

(2) After take-off set the "ON-OFF" switch (figure 30-3) to "ON".



1. Six-position switch.
2. Emergency switch.
3. ON-OFF switch.
4. Protection plug (to prevent short circuit with safety belt).
5. Headset jack.
6. Destructor buttons.
7. "G" band switches.

Figure 30 — SCR 695-A Radio Controls

(3) Set the six-position switch (figure 30-1) to the position specified by the Communications Officer-in-Charge. (In the absence of specific information, set to "1".)

(4) Set the "G" band switches (figure 30-7) as directed by the Communications Officer.

(5) Details concerning the use of the EMERGENCY switch (figure 30-2) and destructor buttons (figure 30-6) should be obtained from the Communications Officer.

(6) To stop the equipment, turn all switches OFF.

(7) Disconnect the destructor plug as soon as the airplane lands in friendly territory.

f. **BC-608 CONTACTOR.** — Provisions have been made for the installation of this unit when in British squadrons or when operating in conjunction with British airplanes.

(1) Before take-off, turn the contactor heater switch ON, wind the clock and synchronize it with the control clock by setting the clock switch (figure 31-3) to RUN at the given signal.

(2) Turn the command transmitter ON.

(3) Turn the contactor switch (figure 31-1) to IN.

(a) Automatic transmission takes place during the 15 seconds the clock hand is between the 12:00 and the 3:00 o'clock position. Normal transmission and reception is not possible during this time.

g. **MN-26Y RADIO COMPASS.**—This radio compass may be installed in some airplanes. The compass is intended for use during the ferry flights, and may be removed in combat zones.

(1) GENERAL.

(a) The master control switch marked OFF, COMP., REC. ANT., and REC. LOOP controls all radio compass equipment functions other than tuning and adjustment of signal level.

(b) The COMP setting is used for obtaining communications reception, visual on-course indication of homing and bearings.

(c) The REC. ANT setting is used for communication and aural radio range reception.

(d) The REC. LOOP setting is used for obtaining communications reception during conditions of severe rain and snow static, aural radio range reception, aural null homing from communication stations.

(e) The OFF setting opens all current consuming circuits thus rendering the equipment inoperative.

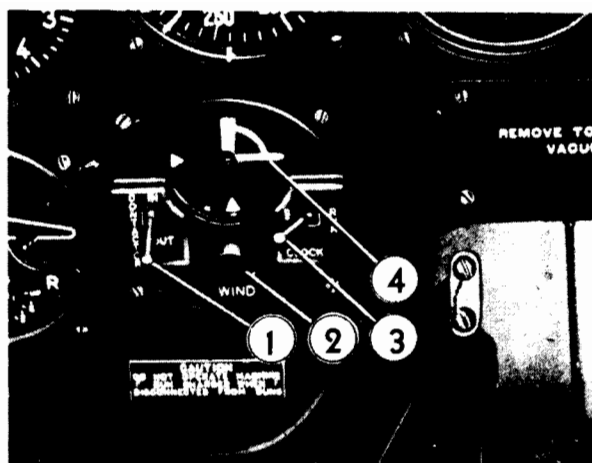
(f) The left-right indicator indicates whether the station is to the right or left of the centerline of the airplane.

(2) OPERATION OF THE MN-26Y RADIO COMPASS.

(a) Turn the main switch ON and set the band selector switch to the frequency range desired.

(b) Tune to the desired station and adjust the volume (audio) control. Set the CW switch to ON for code reception, OFF for voice reception.

(c) Set the master switch according to the operation to be conducted.



1. Contactor in-out switch.
2. Clock winding knob.
3. Clock stop-run switch.
4. Clock indicator.

Figure 31 — BC 608 Contactor Controls

4. ARMAMENT.

a. The P-38H, P-38J and P-38L airplanes are equipped with four, .50 caliber machine guns and one 20 mm cannon and a small motion picture camera. Space is provided for carrying 500 rounds of ammunition for each .50 caliber gun and 150 rounds for the 20 mm cannon. The motion picture camera operates with the guns to record results.

(1) Before take-off check that machine guns and cannon have been charged. (P-38H airplanes only are equipped with a machine gun charger which may be operated in flight as follows:)

(a) Pull the charging selector (figure 4-16) out and turn it to the gun to be charged.

(b) Pull the charging handle (figure 4-10) all the way back and then push it forward.

(c) Strike the charging selector with the heel of the hand.

Note

Never attempt to move the charging selector unless the charging handle is in the full forward position.

(d) Charge the other three machine guns in the same manner.

(2) Turn the armament switch (on control column) to COMBAT.

(3) Push the machine gun button or the cannon button (on control wheel) to fire the guns.

(a) If it is desired to use the camera without the guns, set the armament switch to CAMERA and operate either the gun or cannon button.

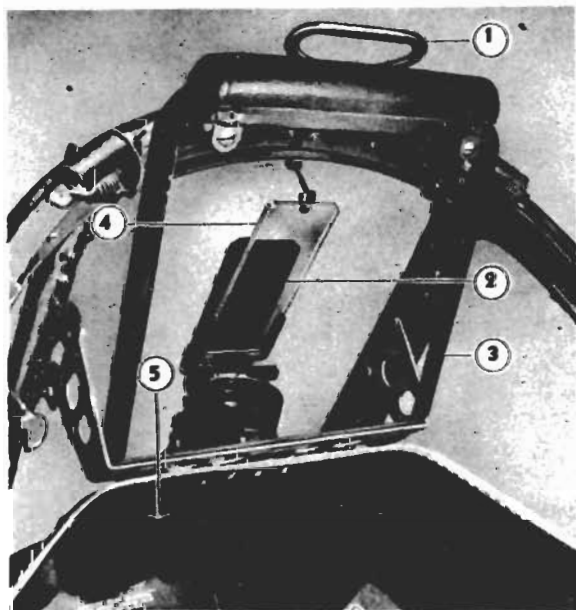
Note

Turn gun heat (figure 4-23, or switch on main switch box) ON whenever the outside air temperature is below freezing.

(4) The optical gun sight is turned ON by the rheostat (figure 5-10) which adjusts the intensity of the reflection on the glass behind the windshield. Adjust the seat vertically until the sight reflection is easily visible.

(5) When flying in bright sunlight, place the dark glass (figure 4-31) in position over the gunsight (figure 32-2). Later airplanes are not equipped with gunsight dark glass.

(6) To drop bombs, set the arming switch (figure 4-24) to ARM (or safe), set the bomb selector switch (figure 4-20) to ON for the bomb to be dropped and, while flying at less than 400 mph IAS and not more than 30° from the horizontal, with flaps UP, press the release button (figure 4-27).



1. Emergency hatch release.
2. Gunsight dark glass.
3. Bullet proof glass.
4. Reflecting glass.
5. Glare shield.

Figure 32 — Gun Sight (Early Airplanes)

5. PHOTOGRAPHIC EQUIPMENT.

a. On F-5B airplanes all the cameras are installed in the fuselage nose which is divided into three camera compartments. The forward compartment accommodates one 6-inch K-17 chart camera. The center compartment will accommodate two 6-inch K-17 chart cameras or one 12-inch or 24-inch K-17 reconnaissance camera. The aft compartment accommodates two 24-inch K-17, or one 24-inch K-18, reconnaissance cameras.

b. CAMERA OPERATION.

(1) Before take-off the pilot should determine which cameras are installed and that the airplane is properly balanced. THIS IS IMPORTANT (see Section II, paragraph 2, a.

(2) TO OPERATE CAMERAS:

(a) Set camera diaphragm control (figure 33-4) as required. (Bright, medium, or cloudy.)

(b) Turn camera switches (figures 33-6, 7 or 8) ON for each group of cameras to be used.

Note

Camera heat (figure 4-23) should be adjusted to maintain approximately 5°C (40°F) on the camera compartment temperature indicator (figure 33-1). Some airplanes may be equipped with ethylene glycol electric fan heaters which will be operated through a switch and rheostat in the cockpit.

(c) FOR INTERVALOMETER OPERATION:

1. Determine time interval from the INTERVALOMETER SETTING table (figure 34).

2. Set the intervalometer hand to the selected time interval. Push and turn knob (figure 33-11) to set hand.

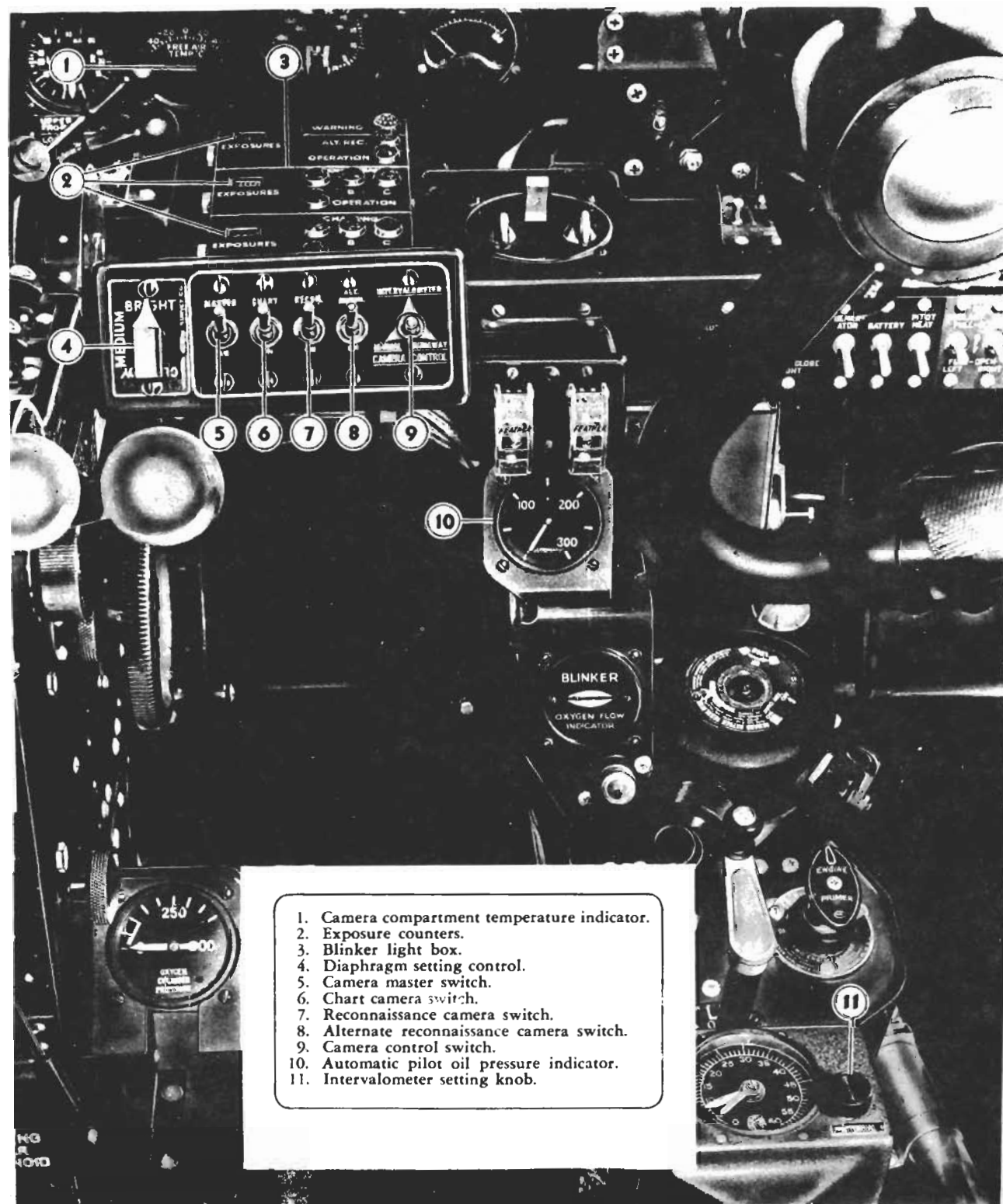
3. Set the camera control switch (figure 33-9) to INTERVALOMETER.

4. Turn camera master switch (figure 33-5) ON. An exposure will be made immediately.

5. Amber lights flash when exposure is made. Green lights indicate when film is winding. The white light gives a three-second warning before exposure.

6. To take an extra picture at any instant, set the control switch to RUNAWAY and immediately return it to INTERVALOMETER.

(d) FOR MANUAL CONTROL:



1. Camera compartment temperature indicator.
2. Exposure counters.
3. Blinker light box.
4. Diaphragm setting control.
5. Camera master switch.
6. Chart camera switch.
7. Reconnaissance camera switch.
8. Alternate reconnaissance camera switch.
9. Camera control switch.
10. Automatic pilot oil pressure indicator.
11. Intervalometer setting knob.

Figure 33 — A-1 Camera Controls

(F-5B ONLY)

1. Set camera control switch to **MANUAL**.
2. Turn camera master switch **ON**.
3. Pull the camera control trigger, (on the control wheel) to make an exposure.

(e) **FOR RUNAWAY OPERATION:**

1. Set camera control switch to **RUNAWAY**.
2. Turn camera master switch **ON**.

Note

Because the time to wind varies slightly for each camera, exposures will not be made simultaneously during runaway operation. Pictures will, therefore, not be suitable for stereoscopic viewing.

3. Runaway operation may also be obtained by setting the controls for manual operation and holding the trigger down.

AIRPLANE'S AIRSPEED INDICATOR READING—MPH

INTERVALOMETER SETTING—SEC.

Airplane's Altimeter Reading (1,000 Ft.) — Altimeter Set at 29.92" Hg.																		Altitude Above Ground — 1,000 Ft.																	
6	8	10	12	14	16	18	20	22	24	26	28	30	32	34	36	38	6	8	10	12	14	16	18	20	22	24	26	28	30	32	34	36	38		
180	170	165	160	155	150	145	140	130	125								2"	3"	4†	5†	5†	6†	6†	9	9	9	9	12	12	12	15	15	15		
210	200	195	185	180	170	165	160	150	145	140	135	125					2"	3"	4†	4†	5†	6†	6†	9	9	9	9	9	12	12	12	12	12		
235	230	220	215	205	195	185	180	170	165	160	150	145	140	135	125		2"	3"	3"	4†	4†	5†	6†	6†	6†	9	9	9	9	9	12	12	12		
260	255	245	240	230	225	215	205	195	190	180	170	165	160	150	145	140	2"	2"	3"	3"	4†	5†	5†	6†	6†	6†	9	9	9	9	9	9	12		
285	275	270	260	250	245	235	230	220	210	200	190	180	175	170	160	155	2"	2"	3"	3"	4†	4†	5†	5†	6†	6†	6†	6†	9	9	9	9	9		
310	300	290	280	275	265	260	250	240	230	220	215	205	195	185	180	170	1"	2"	2"	3"	3"	4†	4†	5†	5†	6†	6†	6†	6†	9	9	9	9		
330	320	310	300	295	285	280	270	260	250	240	230	225	215	205	195	190	1"	2"	2"	3"	3"	4†	4†	5†	5†	5†	6†	6†	6†	6†	9	9	9		
355	345	335	325	315	310	300	290	280	270	260	250	240	230	225	215	205	1"	2"	2"	3"	3"	4†	4†	5†	5†	5†	6†	6†	6†	6†	9	9	9		
380	370	360	350	340	330	320	310	300	290	280	270	260	250	240	230	220	1"	1"	2"	2"	3"	3"	4†	4†	4†	5†	5†	5†	6†	6†	6†	6†	9		

To determine intervalometer setting: (1) Find airplane's altimeter reading in left hand side of table; (2) Follow down to first indicated airspeed higher than airplane's airspeed; (3) Follow across horizontal to intervalometer setting below distance above the ground.

Time intervals marked (") are too short for runaway operation with K-17 camera.

Time intervals marked (†) are too short for runaway operation with K-18 camera.

Figure 34 — Intervalometer Setting (60% Overlap)



APPENDIX I

GLOSSARY OF NOMENCLATURE (U. S. A.-BRITISH)

<i>AMERICAN</i>	<i>BRITISH</i>	<i>AMERICAN</i>	<i>BRITISH</i>
Accumulator (hydraulic)	Pressure reservoir	Lean mixture	Weak mixture
Air filter	Air cleaner	Left	Port
Airfoil	Aerofoil	Level off	Flatten out
Antenna	Aerial	Life raft	Dinghy
Battery (electrical)	Electrical accumulator	Manifold pressure	Boost pressure
Ceiling	Cloud height	Mooring line	Mooring guy
Check valve	Non-return valve	Mooring rings	Picketing rings
Command set (radio)	Pilot controller set	Oil pan	Crankcase sump
Course (direction with respect to true north)	Track angle	Panel (center wing)	Centre section plane
Critical speed	Stalling speed	Panel (outboard)	Outer plane
Cylinder (hydraulic)	Jack (hydraulic)	Radio mast	Rod aerial
Drift	Drift-angle	Radio range beacon	Radio track beacon
Empennage	Tail unit	Right	Starboard
Flare (signal)	Signal star, signal projectile	Screen (oil)	Filter
Gage (fuel)	Fuel-contents gauge	Stabilizer (horizontal)	Tail plane
Gasoline (gas, fuel)	Petrol or fuel	Stabilizer (vertical)	Fin
Generator	Dynamo	Tachometer	Engine speed indicator, revolution counter
Ground (electrical)	Earth	Take-off distance	Take-off run
Gyro horizon	Artificial horizon	Tube (radio)	Valve
Heading	Course	Valve (fuel or oil)	Cock
Indicated air speed (ias)	Air-speed-indicator reading	Weight (empty)	Tare
Landing gear	Alighting gear, under- carriage	Weight (gross)	All-up weight
		Windshield	Windscreen
		Wing	Main plane

APPENDIX II

Flight Operating Charts, Tables, Curves, and Diagrams

1. FLIGHT PLANNING.

The following outline may be used as a guide to assist personnel in the use of the FLIGHT OPERATION INSTRUCTION CHART for flight planning purposes.

a. If the flight plan calls for a continuous flight where the desired cruising power and air speed are reasonably constant after take-off and climb to 5,000 feet, the fuel required and flight time may be computed as a "single-section flight."

(1) Within the limits of the airplane, the fuel required and flying time for a given mission depend largely upon the speed desired. With all other factors remaining equal in an airplane, speed is obtained at a sacrifice of range, and range is obtained at a sacrifice of speed. The speed is usually determined after considering the urgency of the flight plotted against the range required. The time of take-off is adjusted so as to have the flight arrive at its destination at the predetermined time.

Fuel should be used in the following sequence:

1. Reserve tanks for first 15 minutes.
2. External tanks.
3. Outer wing tanks (if installed).
4. Main tanks.
5. Reserve tanks.

(2) Select the FLIGHT OPERATION INSTRUCTION CHART corresponding to the weight and external load items of the airplane. Locate the largest figure entered under gph (gallons per hour) in column 1 on the lower half of the chart. Multiply this figure by the number and/or fraction of hours desired for reserve fuel. Add the resulting figure to the number of gallons set forth in footnote No. 2, and subtract the total from the amount of fuel in the airplane prior to starting the engines. The figure obtained as a result of this computation will represent the amount of gasoline available

and applicable for flight planning purposes on the "Range in Air Miles" section of the FLIGHT OPERATION INSTRUCTION CHART.

(3) Select a figure in the fuel column equal to, or the next entry less than, the available amount of fuel in the airplane as determined in paragraph 2, *a*, (2) above. Move horizontally to the right or left and select a figure equal to, or the next entry greater than the air miles (with no wind) to be flown. Operating values contained in the column number in which this figure appears, represent the highest cruising speed possible at the range desired; however, the airplane may be operated in accordance with values contained under OPERATING DATA in any column of a higher number with the flight plan being completed at a sacrifice of speed but at an increase in fuel economy.

(4) Using the same column number selected by application of instructions contained in the preceding paragraph, read the gallons per hour given at the altitude to be flown and divide this figure into the number of gallons available for cruising, as determined in paragraph (2) above. This will give the calculated flight duration in hours, which can then be converted into hours and minutes and deducted from the desired arrival time at destination in order to obtain the take-off time (without consideration for wind). To allow for wind, determine the calculated ground speed by dividing the flight duration in hours into the range selected in paragraph (3) and calculate a new corrected ground speed with the aid of a navigator's triangle of velocities.

(5) The airplane and engine operating values listed below "Operating Data" in any column except I are calculated to give constant miles per gallon at any altitude listed. Therefore, the airplane may be operated at any altitude and at the corresponding set of values given so long as they are in the same column listing the range desired.

CAUTION

Ranges listed in column I under "Max. Cont. Power" are correct only at sea level and 12,000 feet.

(6) The flight plan may be readily changed at any time enroute, and the chart will show the balance of range at various cruising powers by following the "Instructions for Using Chart" printed on each page.

(7) In using the FLIGHT OPERATION INSTRUCTION CHARTS set the propeller control

to give the desired rpm and open the throttle to give the desired indicated air speed. Use the manifold pressure only as an approximate value for reference.

b. If the original flight plan calls for a mission requiring changes in power, speed, or gross load, in accordance with "GR. WT," increments shown in the series of "FLIGHT OPERATION INSTRUCTION CHARTS" provided, the total flight should be broken down into a series of individual short flights, each computed as outlined in paragraph a, in its entirety, and then added together to make up the total flight and its requirements.



"WHAT WAS THAT?"

AIRPLANE MODELS

P-38H

TAKE-OFF, CLIMB & LANDING CHART

ENGINE MODELS

V-1710-89(RH)

V-1710-91(LH)

TAKE-OFF DISTANCE (IN FEET)

GROSS WEIGHT (IN LBS.)	HEAD WIND MPH	HARD SURFACE RUNWAY						SOFT SURFACE RUNWAY					
		AT SEA LEVEL			AT 3,000 FT.			AT 3,000 FT.			AT 3,000 FT.		
		TO CLEAR 50' OBL.			TO CLEAR 50' OBL.			TO CLEAR 50' OBL.			TO CLEAR 50' OBL.		
		GROUND RUN	TIME FROM S.L.	PT/MIN	GROUND RUN	TIME FROM S.L.	PT/MIN	GROUND RUN	TIME FROM S.L.	PT/MIN	GROUND RUN	TIME FROM S.L.	PT/MIN
16,100	0	840	1320	950	1510	1100	1770	880	1360	1020	1580	1190	1870
	17	15	600	980	690	1150	790	630	1040	720	1190	850	1410
	34	30	390	710	450	820	520	420	740	470	850	550	1010
	51	45	220	470	240	530	310	240	480	280	570	320	670
	68	60	1140	1790	1300	2080	1510	2511	1180	1840	1380	2180	1620
18,100	0	1140	1790	1300	2080	1510	2511	1180	1840	1380	2180	1620	2630
	17	15	830	1380	960	1620	1110	1960	860	1410	1010	1680	1190
	34	30	560	1010	660	1200	750	1450	590	1040	700	1250	810
	51	45	340	700	400	830	460	1010	350	710	420	850	500
	68	60	1370	2220	1580	2660	1840	3340	1470	2330	1710	2800	2000
19,500	0	1370	2220	1580	2660	1840	3340	1470	2330	1710	2800	2000	3510
	17	15	1020	1750	1180	2100	1360	2650	1090	1820	1270	2200	1480
	34	30	710	1310	830	1590	950	2020	760	1370	890	1660	1040
	51	45	450	940	530	1140	620	1490	480	970	550	1170	670
	68	60	1470	2420	1840	3040	2140	3640	1740	2840	2140	3440	2340

NOTE: INCREASE DISTANCE 9 % FOR EACH 10°C ABOVE 0°C (10 % FOR EACH 20°F ABOVE 32°F)

CLIMB DATA

GROSS WEIGHT (IN LBS.)	TYPE OF CLIMB	3000 RPM & 54" IN. HG						2300 RPM & 54" IN. HG					
		S.L. TO 5,000 FT. ALT.			15,000 FT. ALT.			25,000 FT. ALT.			35,000 FT. ALT.		
		BEST I.A.S.			BEST I.A.S.			BEST I.A.S.			BEST I.A.S.		
		MPH	KNOTS	PT/MIN	MPH	KNOTS	PT/MIN	MPH	KNOTS	PT/MIN	MPH	KNOTS	PT/MIN
16,100	COMBAT	178	155	3500	2	175	152	3400	3	61	53	172	149
	FERRY	151	131	1700	3	151	131	1600	6	60	52	151	131
	CLIMB	178	155	2800	2	175	152	2700	4	65	56	172	149
18,100	COMBAT	151	131	1200	4	151	131	1200	8	65	56	151	131
	FERRY	151	131	1200	4	151	131	1200	8	65	56	151	131
	CLIMB	178	155	2500	2	175	152	2300	4	68	59	172	149
19,500	COMBAT	151	131	1000	5	151	131	1000	10	70	61	151	131
	FERRY	151	131	1000	5	151	131	1000	10	70	61	151	131
	CLIMB	178	155	2000	4	175	152	1800	8	73	64	172	149

NOTE: INCREASE ELAPSED CLIMBING TIME 5 % FOR EACH 10°C ABOVE 0°C (10 % FOR EACH 20°F ABOVE 32°F) FUEL INCLUDES WARM-UP AND TAKE-OFF ALLOWANCE

LANDING DISTANCE (IN FEET)

GROSS WEIGHT (IN LBS.)	BEST I.A.S. APPROACH	HARD DRY SURFACE						FIRM DRY SOG					
		AT SEA LEVEL			AT 3,000 FT.			AT SEA LEVEL			AT 3,000 FT.		
		TO CLEAR 50' OBL.			TO CLEAR 50' OBL.			TO CLEAR 50' OBL.			TO CLEAR 50' OBL.		
		MPH	KNOTS	GROUND RUN	TIME FROM S.L.	PT/MIN	GROUND RUN	MPH	KNOTS	GROUND RUN	TIME FROM S.L.	PT/MIN	GROUND RUN
13,500	115	101	1760	1170	1920	1300	2100	1460	1920	1330	2100	1480	2310
	120	104	1950	1330	2130	1480	2350	1660	2140	1520	2350	1700	2580
	125	107	2140	1520	2350	1660	2580	1840	2350	1700	2580	1900	2970
15,500	115	101	1760	1170	1920	1300	2100	1460	1920	1330	2100	1480	2310
	120	104	1950	1330	2130	1480	2350	1660	2140	1520	2350	1700	2580
	125	107	2140	1520	2350	1660	2580	1840	2350	1700	2580	1900	2970

NOTE: FOR GROUND TEMPERATURES ABOVE 35°C (95°F) INCREASE APPROX 1 A.S. 10 % AND ALLOW 20 % INCREASE IN GROUND ROLL.

REMARKS * FOR COMBAT CLIMB USE 3000 R.P.M. AND 54 IN. HG. TO 20,000 FEET. REDUCE MAN. PRESS. AS FOLLOWS ABOVE 20,000 FT. STILL USING 3000 R.P.M.

ALTITUDE			
25,000	30,000	35,000	40,000
45	35	30	20
52	49	41	33

1 A.S. Indicated Air Speed
M.P.H. Miles Per Hour
KNOTS U.S. Knots
U.S. U.S. Gallons
IMPERIAL Imperial Gallons
NOTE: All Distances are Average
RED FIGURES HAVE NOT BEEN FLIGHT CHECKED

AIRPLANE MODELS

P-38J

F-5B

TAKE-OFF, CLIMB & LANDING CHART

ENGINE MODELS

V-1710-89(RH)

V-1710-91(LH)

FORM ASC-510
DEC 19, 1943

TAKE-OFF DISTANCE (IN FEET)

GROSS WEIGHT (IN LBS.)	HARD SURFACE RUNWAY				SOD-TURF RUNWAY				SOFT SURFACE RUNWAY			
	HEAD WIND		AT SEA LEVEL		AT 3,000 FT.		AT 6,000 FT.		AT SEA LEVEL		AT 3,000 FT.	
	KNOTS		TO CLEAR 50' OB.		GROUND RUN		TO CLEAR 50' OB.		GROUND RUN		TO CLEAR 50' OB.	
	MPH	KT	FT.	FT.	FT.	FT.	FT.	FT.	FT.	FT.	FT.	FT.
17,400	0	0	1030	1620	1170	1850	1360	2210	1060	1240	1940	1450
	17	15	750	1230	860	1440	980	1720	770	1260	1490	1080
	34	30	500	890	580	1050	660	1260	520	1080	1300	910
	51	45	300	610	340	700	390	850	310	620	730	430
	0	0	1740	2190	1560	2820	1810	3280	1450	2300	2750	1970
19,400	0	0	1000	1600	1160	1800	1350	2200	1070	1250	1910	1470
	17	15	730	1200	840	1400	960	1700	750	1230	1460	1060
	34	30	500	880	580	1040	660	1250	520	1070	1300	910
	51	45	300	600	340	690	390	840	310	610	720	420
	0	0	1740	2190	1560	2820	1810	3280	1450	2300	2750	1970
21,400	0	0	1000	1600	1160	1800	1350	2200	1070	1250	1910	1470
	17	15	730	1200	840	1400	960	1700	750	1230	1460	1060
	34	30	500	880	580	1040	660	1250	520	1070	1300	910
	51	45	300	600	340	690	390	840	310	610	720	420
	0	0	1740	2190	1560	2820	1810	3280	1450	2300	2750	1970

NOTE: INCREASE DISTANCE 9 % FOR EACH 10°C ABOVE 0°C (1.0 % FOR EACH 20°F ABOVE 32°F)

ENGINE LIMITS FOR TAKE-OFF 3000 RPM & 54 IN. HG

CLIMB DATA

GROSS WEIGHT (IN LBS.)	COMBAT MISSIONS USE				CLIMB DATA				FERRY MISSIONS USE			
	S.L. TO 5000 FT. ALT.		10,000 FT. ALT.		15,000 FT. ALT.		25,000 FT. ALT.		35,000 FT. ALT.		45,000 FT. ALT.	
	BEST I.A.S.		TIME FROM S.L.		TIME FROM S.L.		TIME FROM S.L.		TIME FROM S.L.		TIME FROM S.L.	
	MPH	KT	FT./MIN	FT./MIN	MPH	KT	MPH	KT	MPH	KT	MPH	KT
17,400	180	156	3200	2	178	154	3100	4	59	49	175	152
	160	139	1400	4	160	139	1400	7	57	48	160	139
	140	122	800	7	140	122	800	10	55	46	140	122
	120	104	500	10	120	104	500	13	53	44	120	104
	100	87	300	13	100	87	300	16	51	42	100	87
19,400	180	156	2800	2	178	154	2400	4	63	53	175	152
	160	139	1100	5	160	139	1100	10	62	52	160	139
	140	122	600	8	140	122	600	13	60	50	140	122
	120	104	350	11	120	104	350	16	58	48	120	104
	100	87	200	14	100	87	200	18	56	46	100	87
21,400	185	161	2200	3	183	159	2000	5	57	47	180	156
	165	143	800	7	165	143	700	13	71	59	165	143
	145	125	400	10	145	125	400	13	69	57	145	125
	125	107	200	13	125	107	200	16	67	55	125	107
	105	91	100	16	105	91	100	19	65	53	105	91

NOTE: INCREASED ELAPSED CLIMBING TIME 5 % FOR EACH 10°C ABOVE 0°C FREE AIR TEMPERATURE (5 % FOR EACH 20°F ABOVE 32°F) FUEL INCLUDES WARM-UP AND TAKE-OFF ALLOWANCE

LANDING DISTANCE (IN FEET)

GROSS WEIGHT (IN LBS.)	HARD DRY SURFACE				FIRM DRY SOD				WET OR SLIPPERY			
	AT SEA LEVEL		AT 3,000 FT.		AT SEA LEVEL		AT 3,000 FT.		AT SEA LEVEL		AT 3,000 FT.	
	TO CLEAR 50' OB.		GROUND ROLL		TO CLEAR 50' OB.		GROUND ROLL		TO CLEAR 50' OB.		GROUND ROLL	
	MPH	KT	FT.	FT.	MPH	KT	FT.	FT.	MPH	KT	FT.	FT.
14,000	120	104	1800	1090	1960	1220	2160	1370	1970	1250	2160	1400
	100	87	1000	800	1100	700	1200	900	1100	700	1200	900
	80	69	600	480	700	420	800	500	800	500	800	500
	60	52	350	280	400	250	450	300	400	250	450	300
	40	34	200	160	250	150	300	180	250	150	300	180
16,000	125	109	2010	1260	2200	1400	2430	1570	2210	1270	2430	1600
	105	91	1100	890	1200	750	1300	950	1200	750	1300	950
	85	73	650	520	750	450	850	550	750	450	850	550
	65	56	380	300	450	280	500	320	450	280	500	320
	45	39	220	180	280	170	320	200	280	170	320	200

NOTE: FOR GROUND TEMPERATURES ABOVE 35°C (95°F) INCREASE APPROACH I.A.S. 10 % AND ALLOW 20 % INCREASE IN GROUND ROLL.

REMARKS * FOR COMBAT CLIMB USE 3000 R.P.M. AND 54 IN. HG. TO 25,000 FEET THEN GRADUALLY REDUCE MANIFOLD PRESSURE TO 48.5 IN. HG. AT 30,000 FEET AND TO 41.0 IN. HG. AT 35,000 FT.

I.A.S.: Indicated Air Speed
M.P.H.: Miles Per Hour
S.L.: Sea Level
U.S.: U. S. Gallons
IMP.: Imperial Gallons
NOTE: All Distances are Average
RED FIGURES HAVE NOT BEEN FLIGHT CHECKED

AIRPLANE MODELS

P-38L

SPC. A.M.-8
Rev. 10, 1942
FORM ASC-510

TAKE-OFF, CLIMB & LANDING CHART

ENGINE MODELS

V-1710-III(RH)

V-1710-III(LH)

TAKE-OFF DISTANCE (IN FEET)

GROSS WEIGHT (IN LBS.)	HARD SURFACE RUNWAY				SOFT SURFACE RUNWAY			
	HEAD WIND		AT SEA LEVEL		AT SEA LEVEL		AT SEA LEVEL	
	MPH	KNOTS	GROUND RUN 50' OBL.	TO CLEAR 50' OBL.	GROUND RUN 50' OBL.	TO CLEAR 50' OBL.	GROUND RUN 50' OBL.	TO CLEAR 50' OBL.
17,400	0	0	1030	1620	1170	1850	1360	2210
	17	15	750	1250	860	1440	990	1720
	34	30	500	880	580	1050	660	1280
	51	45	300	610	340	700	390	850
	0	0	1350	2190	1560	2620	1810	3280
19,400	17	15	1000	1720	1160	2060	1350	2600
	34	30	690	1290	820	1560	940	1970
	51	45	440	920	520	1120	610	1450
	0	0	1740	2910	2000	3560	2320	4960
	17	15	1300	2400	1520	2840	1760	4020
21,400	34	30	930	1740	1090	2180	1280	3140
	51	45	610	1250	730	1610	830	2330
	0	0	2000	3560	2320	4960	1860	3040
	17	15	1300	2400	1520	2840	1760	4020
	34	30	930	1740	1090	2180	1280	3140

NOTE: INCREASE DISTANCE 9 % FOR EACH 10°C ABOVE 0°C (10 % FOR EACH 20°F ABOVE 32°F)

CLIMB DATA

GROSS WEIGHT IN LBS.	TYPE O. CLIMB	COMBAT MISSIONS USE				CLIMB DATA				FERRY MISSIONS USE				IN. HG.			
		3000		54		15,000		25,000		2300		35,000		3000		54	
		S.L. TO 5000 FT. ALT.	BEST I.A.S. MPH	TIME PT/MIN	FUEL FROM S.L. U.S.	S.L. TO 10,000 FT. ALT.	BEST I.A.S. MPH	TIME PT/MIN	FUEL FROM S.L. U.S.	S.L. TO 15,000 FT. ALT.	BEST I.A.S. MPH	TIME PT/MIN	FUEL FROM S.L. U.S.	S.L. TO 20,000 FT. ALT.	BEST I.A.S. MPH	TIME PT/MIN	FUEL FROM S.L. U.S.
17,400	COMBAT	180	156	3200	2	178	154	3100	4	59	49	175	152	2900	11	66	55
		160	139	1400	4	160	139	1400	7	57	48	160	139	1400	11	66	55
		180	156	2800	2	178	154	2400	4	63	53	175	152	2300	7	75	62
19,400	COMBAT	160	139	1100	5	160	139	1100	10	62	52	160	139	1000	15	72	60
		185	161	2200	3	183	159	2000	5	67	56	180	156	1900	8	81	67
		165	143	800	7	165	143	700	13	71	59	165	143	700	20	87	72
21,400	COMBAT	185	161	2200	3	183	159	2000	5	67	56	180	156	1900	8	81	67
		165	143	800	7	165	143	700	13	71	59	165	143	700	20	87	72
		185	161	2200	3	183	159	2000	5	67	56	180	156	1900	8	81	67

NOTE: INCREASED ELAPSED CLIMBING TIME 5 % FOR EACH 10°C ABOVE 0°C FREE AIR TEMPERATURE (5 % FOR EACH 20°F ABOVE 32°F) FUEL INCLUDES WARM-UP AND TAKE-OFF ALLOWANCE

LANDING DISTANCE (IN FEET)

GROSS WEIGHT IN LBS.	BEST I.A.S. APPROACH	HARD DRY SURFACE				FIRM DRY SO D				WET OR SLIPPERY			
		AT SEA LEVEL		AT 3,000 FT.		AT SEA LEVEL		AT 3,000 FT.		AT SEA LEVEL		AT 3,000 FT.	
		TO CLEAR 50' OBL.	GROUND ROLL	TO CLEAR 50' OBL.	GROUND ROLL	TO CLEAR 50' OBL.	GROUND ROLL	TO CLEAR 50' OBL.	GROUND ROLL	TO CLEAR 50' OBL.	GROUND ROLL	TO CLEAR 50' OBL.	GROUND ROLL
14,000	120	1800	1090	1960	1220	2160	1370	1970	1250	2160	1400	2370	1560
		2010	1260	2200	1400	2430	1570	2210	1440	2420	1600	2670	1790
		125	109	1260	1400	1400	1570	1400	1250	1440	1600	1600	1250
16,000	125	2010	1260	2200	1400	2430	1570	2210	1440	2420	1600	2670	1790
		125	109	1260	1400	1400	1570	1400	1250	1440	1600	1600	1250
		125	109	1260	1400	1400	1570	1400	1250	1440	1600	1600	1250

NOTE: FOR GROUND TEMPERATURES ABOVE 35°C (95°F) INCREASE APPROACH I.A.S. 10% AND ALLOW 20% INCREASE IN GROUND ROLL.

REMARKS * FOR COMBAT CLIMB USE 3000 R.P.M. AND 54 IN. HG. TO 26,600 FEET THEN GRADUALLY REDUCE MANIFOLD PRESSURE TO 48.5 IN. HG. AT 30,000 FEET AND TO 41.0 IN. HG. AT 35,000 FEET.

1 I.A.S. Indicated Air Speed
2 M.P.H. Miles Per Hour
3 K.T.S. Knots
4 U.S. U.S. Gallons
5 IMP. Imperial Gallons
6 NOTE: All Distances are Average
7 RED FIGURES HAVE NOT BEEN FLIGHT CHECKED

MODEL(S)										FLIGHT OPERATION INSTRUCTION CHART										EXTERNAL LOAD ITEMS									
P-38 H										ENGINE(S): V-1710-89 & V-1710-91										CHART WEIGHT LIMITS: 16,100 TO 13,500 POUNDS									
LIMITS										INSTRUCTIONS FOR USING CHART: Select figure in FUEL column equal to or less than amount of fuel to be used for cruising. Move horizontally to left or right and select RANGE value equal to or greater than the statute or nautical air miles to be flown. Vertically below and opposite desired cruising altitude (ALT.) read optimum R. P. M., I. A. S. and MIXTURE setting required.										NOTES: Column I is for emergency high speed cruising only. Columns II, III, IV and V give progressive increase in range at a sacrifice in speed. Manifold pressure (M. P.), gallons per hour (G. P. H.) and true airspeed (T. A. S.) are approximate values for reference. For efficiency maintain indicated airspeed (I. A. S.) hourly. Adjust RPM slightly if necessary to avoid exceeding manifold pressure more than 3 in. Hg.									
WAR MAX										MIXTURE POSITION										TIME LIMIT									
MILITARY POWER										M. P. (IN. HG.)										BLOWER POSITION									
NORMAL RATED										M. P.										MIXTURE POSITION									
										44										NONE									
										15										334									
										5										360									
										AR										5									
										AR										360									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									
										AR										226									

MODEL(S)				FLIGHT OPERATION INSTRUCTION CHART				EXTERNAL LOAD ITEMS			
P-38H				ENGINE(S): V-1710-89 & V-1710-91				TWO 165 GALLON TANKS			
				CHART WEIGHT LIMITS: 18,100 TO 13,500 POUNDS							
				INSTRUCTIONS FOR USING CHART: Select figure in FUEL column equal to or less than amount of fuel to be used for cruising. Move horizontally to left or right and select RANGE value equal to or greater than the statute or nautical air miles to be flown. Vertically below and opposite desired cruising altitude (ALT.) read optimum R. P. M., I. A. S. and MIXTURE setting required.							
				NOTES: Column I is for emergency high speed cruising only. Columns II, III, IV and V give progressive increase in range at sacrifice in speed. Manifold pressure (M. P.), gallons per hour (G. P. H.) and true airspeed (T. A. S.) are approximate values for reference. For efficiency maintain indicated airspeed (I. A. S.) hourly. Adjust RPM slightly if necessary to avoid exceeding manifold pressure more than 3 in. Hg.							

MODEL(S)										FLIGHT OPERATION INSTRUCTION CHART										EXTERNAL LOAD ITEMS									
ENGINE(S): V-1710-89 & V-1710-91										P-38H										TWO 300 GALLON TANKS									
CHART WEIGHT LIMITS: 19,500 TO 13,700 POUNDS										INSTRUCTIONS FOR USING CHART: Select figure in FUEL column equal to or less than amount of fuel to be used for cruising. Move horizontally to left or right and select RANGE value equal to or greater than the statute or nautical air miles to be flown. Vertically below and opposite desired cruising altitude (ALT.) read optimum R. P. M., I. A. S. and MIXTURE setting required.										NOTES: Column I is for emergency high speed cruising only. Columns II, III, IV and V give progressive increase in range at a sacrifice in speed. Manifold pressure (M. P.), gallons per hour (G. P. H.) and true airspeed (T. A. S.) are approximate values for reference. For efficiency maintain indicated airspeed (I. A. S.) hourly. Adjust RPM slightly if necessary to avoid exceeding manifold pressure more than 3 in. Hg.									
LIMITS	R. P. M.	M. P. (IN. HG.)	BLOWER POSITION	MIXTURE POSITION	TIME LIMIT	TOTAL G. P. H.	I		II		III		IV		V														
							STATUTE	NAUTICAL	STATUTE	NAUTICAL	STATUTE	NAUTICAL	STATUTE	NAUTICAL	STATUTE	NAUTICAL													
WAR MAX	3000	60		AR	5	360																							
MILITARY POWER	3000	54		AR	15	334																							
NORMAL RATED	2600	44		AR	NONE	226																							
STATUTE	RANGE IN AIR MILES		FUEL U. S. GAL.	RANGE IN AIR MILES		FUEL U. S. GAL.	RANGE IN AIR MILES		FUEL U. S. GAL.	RANGE IN AIR MILES		FUEL U. S. GAL.	RANGE IN AIR MILES		FUEL U. S. GAL.	RANGE IN AIR MILES													
	STATUTE	NAUTICAL		STATUTE	NAUTICAL		STATUTE	NAUTICAL		STATUTE	NAUTICAL		STATUTE	NAUTICAL		STATUTE	NAUTICAL												
Use Column 2																													
							900	1180	1030	1510	1310	1840	1780	1600	900	2210	1920												
							840	1130	990	1460	1260			1550	840	2200	1910												
							810								810	2110	1840												
							730	1020	890	1310	1140	1610		1390	730	1910	1660												
							650	910	790	1170	1010	1430		1240	650	1700	1480												
							570	800	700	1030	890	1250		1090	570	1490	1290												
							490	690	600	880	760	1080		940	490	1280	1110												
							410	570	500	740	640	900		780	410	1070	930												
							330	460	400	590	510	730		630	330	860	750												
							250	350	310	450	390	550		480	250	650	570												
							170	240	210	310	270	370		320	170	440	390												
							90	130	110	160	140	200		170	90	230	200												
MAXIMUM CONTINUOUS																													
R. P. M.	I. A. S. M.P.H.	M. P. (IN. HG.)	G. P. H.	T. A. S.	R. P. M.	I. A. S. M.P.H.	M. P. (IN. HG.)	G. P. H.	T. A. S.	R. P. M.	I. A. S. M.P.H.	M. P. (IN. HG.)	G. P. H.	T. A. S.	ALT. Feet	R. P. M.	I. A. S. M.P.H.	M. P. (IN. HG.)	G. P. H.	T. A. S.									
Use Column 2																													

MODEL(S)				FLIGHT OPERATION INSTRUCTION CHART				EXTERNAL LOAD ITEMS			
P-38J and F-5B				ENGINE(S): V-1710-89 & V-1710-91				CHART WEIGHT LIMITS: 17,400 TO 13,500 POUNDS			
								TANK SUPPORTS ONLY			

P-38J and F-5B

FLIGHT OPERATION INSTRUCTION CHART

EXTERNAL LOAD ITEMS

TWO '65 GAL. TANKS

ENGINE(S): V-1710-89 & V-1710-91

CHART WEIGHT LIMITS: 19,400 TO 14,500 POUNDS

LIMITS	R. P. M.	M. P. H. (IN. HG.)	FLOWER POSITION	MIXTURE	TIME LIMIT	TOTAL G. P. H.
WAR MAX.	3000	60	AR	AR	5	360
MILITARY POWER	3000	54	AR	AR	15	334
NORMAL RATED	2600	44	AR	AR	NONE	226

NOTES: Column I is for emergency high speed cruising only. Columns II, III, IV and V give progressive increase in range at a sacrifice in speed. Manifold pressure (M. P.), gallons per hour (G. P. H.) and true airspeed (T. A. S.) are approximate values for reference. For efficient c.y maintain indicated airspeed (I. A. S.) hourly. Adjust RPM slightly if necessary to avoid exceeding manifold pressure more than 3 in. Hg.

I										II				III				IV				V																	
RANGE IN AIR MILES										RANGE IN AIR MILES				RANGE IN AIR MILES				RANGE IN AIR MILES				RANGE IN AIR MILES																	
STATUTE										STATUTE				STATUTE				STATUTE				STATUTE																	
AT S. L. AT 12,000										AT S. L. AT 12,000				AT S. L. AT 12,000				AT S. L. AT 12,000				AT S. L. AT 12,000																	
690	770	710	590	660	620	630	980	830	1060	1220	1130	1080	920	800	700	610	510	420	330	230	150	90																	
640	710	550	520	580	540	570	880	770	980	1130	1030	890	800	700	610	510	420	330	230	150	90																		
580	640	500	500	560	520	550	800	700	890	1030	930	800	700	610	510	420	330	230	150	90																			
520	580	450	450	500	460	490	710	620	800	920	810	700	610	510	420	330	230	150	90																				
450	510	390	390	440	400	430	630	550	700	810	700	610	510	420	330	230	150	90																					
390	440	340	340	380	340	370	550	480	610	700	590	510	420	330	230	150	90																						
330	370	290	290	320	280	310	460	400	510	590	490	420	330	230	150	90																							
270	310	240	240	260	220	250	380	330	420	490	380	330	230	150	90																								
210	240	180	180	210	170	190	290	260	330	380	270	230	160	110	60	10	10	10	10	10	10																		
150	170	130	130	150	110	130	210	180	230	270	180	160	110	60	10	10	10	10	10	10																			
90	100	80	80	90	60	70	130	110	140	160	110	90	60	40	10	10	10	10	10	10																			
MAXIMUM CONTINUOUS										OPERATING DATA										OPERATING DATA										MAXIMUM RANGE									
R. P. M.	I. A. S.	M. P. H.	M. P. H.	G. P. H.	T. A. S.	ALT.	R. P. M.	I. A. S.	M. P. H.	G. P. H.	T. A. S.	ALT.	R. P. M.	I. A. S.	M. P. H.	G. P. H.	T. A. S.	ALT.	R. P. M.	I. A. S.	M. P. H.	G. P. H.	T. A. S.	ALT.	R. P. M.	I. A. S.	M. P. H.	G. P. H.	T. A. S.										
2600	227	A. R.	44	245	354	30000	2550	221	A. R.	41	223	317	2350	200	A. R.	36	161	322	2250	190	A. L.	34	126	309	30000	2000	165	A. L.	30	95	276								
2600	238	A. R.	44	245	330	25000	2500	229	A. R.	40	210	327	2350	208	A. R.	35	151	304	2150	196	A. L.	34	119	291	25000	1850	170	A. L.	31	89	260								
2600	248	A. R.	44	245	326	20000	2450	234	A. R.	39	198	308	2300	216	A. R.	34	144	287	2050	204	A. L.	34	113	275	20000	1700	175	A. L.	31	84	244								
2600	260	A. R.	44	245	314	15000	2450	242	A. R.	38	189	294	2300	226	A. L.	35	131	276	2000	210	A. L.	33	107	260	15000	1600	178	A. L.	31	79	229								
2600	266	A. R.	44	245	307	12000	2400	246	A. R.	38	183	285	2300	231	A. L.	35	131	268	1900	212	A. L.	33	102	249	12000	1600	176	A. L.	30	74	216								
2600	271	A. R.	44	245	298	9000	2400	249	A. R.	37	175	274	2300	235	A. L.	35	131	261	1850	212	A. L.	33	97	239	9000	1600	174	A. L.	29	70	204								
2600	276	A. R.	44	245	290	6000	2400	251	A. R.	36	169	264	2250	237	A. L.	34	125	251	1800	213	A. L.	33	94	229	6000	1600	171	A. L.	28	66	192								
2600	280	A. R.	44	245	282	3000	2350	253	A. R.	36	164	256	2150	238	A. L.	34	120	241	1700	213	A. L.	33	90	219	3000	1600	169	A. L.	27	62	181								
2600	285	A. R.	44	245	275	S. L.	2350	255	A. R.	36	158	247	2100	239	A. L.	34	115	232	1650	213	A. L.	32	86	210	S. L.	1600	163	A. L.	26	58	169								

NOTES

○ ALLOW 60 GAL. FOR WARM-UP, TAKE-OFF & INITIAL CLIMB PLUS ALLOWANCE FOR WIND, RESERVE & COMBAT AS REQD.

EXAMPLE

AT 19,400 LB. GROSS WT. WITH 740 GAL. OF FUEL (AFTER DEDUCTING TOTAL ALLOWANCES OF 60 GAL.) TO FLY 1500 STAT. AIRMILES AT 12,000 FT. ALT. MAINTAIN 1900 RPM AND 212 MPH IND. AIRSPEED WITH MIXTURE SET A. L.

LEGEND

I. A. S.: INDICATED AIRSPEED
M. P. H.: MANIFOLD PRESSURE
G. P. H.: U. S. GAL. PER HOUR
T. A. S.: TRUE AIRSPEED
S. L.: SEA LEVEL
F. T.: FULL THROTTLE
F. R.: FULL RICH
A. R.: AUTO-RICH
A. L.: AUTO-LEAN
C. L.: CRUISE LEAN

MODEL(S)										FLIGHT OPERATION INSTRUCTION CHART										EXTERNAL LOAD ITEMS																													
P-38L										ENGINE(S): V-1710-III & V-1710-113										TANK SUPPORTS ONLY																													
ENGINE(S): V-1710-III & V-1710-113										CHART WEIGHT LIMITS: 17,400 TO 13,500 POUNDS										NOTES: Column I is for emergency high speed cruising only. Columns II, III, IV and V give progressive increase in range at a sacrifice in speed. Manifold pressure (M. P.), gallons per hour (G. P. H.) and true airspeed (T. A. S.) are approximate values for reference. For efficiency maintain indicated airspeed (I. A. S.) hourly. Adjust RPM slightly if necessary to avoid exceeding manifold pressure more than 3 in. Hg.																													
INSTRUCTIONS FOR USING CHART: Select figure in FUEL column equal to or less than amount of fuel to be used for cruising. Move horizontally to left or right and select RANGE value equal to or greater than the statute or nautical air miles to be flown. Vertically below and opposite desired cruising altitude (ALT.) read optimum R. P. M., I. A. S. and MIXTURE setting required.																																																	
LIMITS										RANGE IN AIR MILES										RANGE IN AIR MILES																													
M. P. H. G. P. H. T. M. P. H. G. P. H. T. M. P. H. G. P. H. T.										M. P. H. G. P. H. T. M. P. H. G. P. H. T. M. P. H. G. P. H. T.										M. P. H. G. P. H. T. M. P. H. G. P. H. T. M. P. H. G. P. H. T.																													
WAR										MAX										MILITARY (LOWER)										NORMAL																			
3000										3000										2800																													
54										44																																							
NONE																																																	
I										II										III										IV										V									
RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES									
STATUTE										STATUTE										STATUTE										STATUTE										STATUTE									
AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000									
FUEL										FUEL										FUEL										FUEL										FUEL									
U. S.										U. S.										U. S.										U. S.										U. S.									
GAL.										GAL.										GAL.										GAL.										GAL.									
410										410										410										410										410									
360										360										360										360										360									
340										340										340										340										340									
310										310										310										310										310									
280										280										280										280										280									
250										250										250										250										250									
220										220										220										220										220									
190										190										190										190										190									
170										170										170										170										170									
150										150										150										150										150									
130										130										130										130										130									
110										110										110										110										110									
90										90										90										90										90									
80										80										80										80										80									
70										70										70										70										70									
60										60										60										60										60									
50										50										50										50										50									
40										40										40										40										40									
30										30										30										30										30									
20										20										20										20										20									
10										10										10										10										10									
0										0										0										0										0									
RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES									
STATUTE										STATUTE										STATUTE										STATUTE										STATUTE									
AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000									
FUEL										FUEL										FUEL										FUEL										FUEL									
U. S.										U. S.										U. S.										U. S.										U. S.									
GAL.										GAL.										GAL.										GAL.										GAL.									
410										410										410										410										410									
360										360										360										360										360									
340										340										340										340										340									
310										310										310										310										310									
280										280										280										280										280									
250										250										250										250										250									
220										220										220										220										220									
190										190										190										190										190									
170										170										170										170										170									
150										150										150										150										150									
130										130										130										130										130									
110										110										110										110										110									
90										90										90										90										90									
80										80										80										80										80									
70										70										70										70										70									
60										60										60										60										60									
50										50										50										50										50									
40										40										40										40										40									
30										30										30										30										30									
20										20										20										20										20									
10										10										10										10										10									
0										0										0										0										0									
RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES									
STATUTE										STATUTE										STATUTE										STATUTE										STATUTE									
AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000									
FUEL										FUEL										FUEL										FUEL										FUEL									
U. S.										U. S.										U. S.										U. S.										U. S.									
GAL.										GAL.										GAL.										GAL.										GAL.									
410										410										410										410										410									
360										360										360										360										360									
340										340										340										340										340									
310										310										310										310										310									
280										280										280										280										280									
250										250										250										250										250									
220										220										220										220										220									
190										190										190										190										190									
170										170										170										170										170									
150										150										150										150										150									
130										130										130										130										130									
110										110										110										110										110									
90										90										90										90										90									
80										80										80										80										80									
70										70										70										70										70									
60										60										60										60										60									
50										50										50										50										50									
40										40										40										40										40									
30										30										30										30										30									
20										20										20										20										20									
10										10										10										10										10									
0										0										0										0										0									
RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES									
STATUTE										STATUTE										STATUTE										STATUTE										STATUTE									
AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000									
FUEL										FUEL										FUEL										FUEL										FUEL									
U. S.										U. S.										U. S.										U. S.										U. S.									
GAL.										GAL.										GAL.										GAL.										GAL.									
410										410										410										410										410									
360										360										360										360										360									
340										340										340										340										340									
310										310										310										310										310									
280										280										280										280										280									
250										250										250										250										250									
220										220										220										220										220									
190										190										190										190										190									
170										170										170										170										170									
150										150										150										150										150									
130										130										130										130										130									
110										110										110										110										110									
90										90										90										90										90									
80										80										80										80										80									
70										70										70										70										70									
60										60										60										60										60									
50										50										50										50										50									
40										40										40										40										40									
30										30										30										30										30									
20										20										20										20										20									
10										10										10										10										10									
0										0										0										0										0									
RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES									
STATUTE										STATUTE										STATUTE										STATUTE										STATUTE									
AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000									
FUEL										FUEL										FUEL										FUEL										FUEL									
U. S.										U. S.										U. S.										U. S.										U. S.									
GAL.										GAL.										GAL.										GAL.										GAL.									
410										410										410										410										410									
360										360										360										360										360									
340										340										340										340										340									
310										310										310										310										310									
280										280										280										280										280									
250										250										250										250										250									
220										220										220										220										220									
190										190										190										190										190									
170										170										170										170										170									
150										150										150										150										150									
130										130										130										130										130									
110										110										110										110										110									
90										90										90										90										90									
80										80										80										80										80									
70										70										70										70										70									
60										60										60										60										60									
50										50										50										50										50									
40										40										40										40										40									
30										30										30										30										30									
20										20										20										20										20									
10										10										10										10										10									
0										0										0										0										0									
RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES									
STATUTE										STATUTE										STATUTE										STATUTE										STATUTE									
AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000									
FUEL										FUEL										FUEL										FUEL										FUEL									
U. S.										U. S.										U. S.										U. S.										U. S.									
GAL.										GAL.										GAL.										GAL.										GAL.									
410										410										410										410										410									
360										360										360										360										360									
340										340										340										340										340									
310										310										310										310										310									
280										280										280										280										280									
250										250										250										250										250									
220										220										220										220										220									
190										190										190										190										190									
170										170										170										170										170									
150										150										150										150										150									
130										130										130										130										130									
110										110										110										110										110									
90										90										90										90										90									
80										80										80										80										80									
70										70										70										70										70									
60										60										60										60										60									
50										50										50										50										50									
40										40										40										40										40									
30										30										30										30										30									
20										20										20										20										20									
10										10										10										10										10									
0										0										0										0										0									
RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES										RANGE IN AIR MILES									
STATUTE										STATUTE										STATUTE										STATUTE										STATUTE									
AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000										AT 5.1. AT 12.000									
FUEL										FUEL										FUEL										FUEL										FUEL									
U. S.										U. S.										U. S.										U. S.										U. S.									
GAL.										GAL.										GAL.										GAL.										GAL.									
410										410										410										410										410									
360										360										360										360										360									
340										340										340										340										340									
310										310										310										310										310									
280										280										280										280										280									
250																																																	

MODEL(S)
P-38 L

FLIGHT OPERATION INSTRUCTION CHART

EXTERNAL LOAD ITEMS

TWO 165 GALLON TANKS

ENGINE(S): V-1710-III & V-1710-III

CHART WEIGHT LIMITS: 19,400 TO 14,500 POUNDS

LIMITS	R.P.M.	M.P.H.	FLOWN POSITION	TIME LIMIT	TOTAL G.P.H.
WAR MAX	3000	60	AR	5	360
MILITARY POWER	3000	34	AR	15	334
NORMAL RATED	2600	44	AR	NONE	226

NOTES: Column I is for emergency high speed cruising only. Columns II, III, IV and V give progressive increase in range at a sacrifice in speed. Manifold pressure (M.P.), gallons per hour (G.P.H.) and true airspeed (T.A.S.) are approximate values for reference. For efficiency maintain indicated airspeed (I.A.S.) hourly. Adjust RPM slightly if necessary to avoid exceeding manifold pressure more than 3 in. Hg.

I						II						III						IV						V					
RANGE IN AIR MILES						RANGE IN AIR MILES						RANGE IN AIR MILES						RANGE IN AIR MILES						RANGE IN AIR MILES					
STATUTE						STATUTE						STATUTE						STATUTE						STATUTE					
AT S.L. AT 12,000						AT S.L. AT 12,000						AT S.L. AT 12,000						AT S.L. AT 12,000						AT S.L. AT 12,000					
R.P.M.	I.A.S.	M.P.H.	M.P.	G.P.H.	ALT.	R.P.M.	I.A.S.	M.P.H.	M.P.	G.P.H.	ALT.	R.P.M.	I.A.S.	M.P.H.	M.P.	G.P.H.	ALT.	R.P.M.	I.A.S.	M.P.H.	M.P.	G.P.H.	ALT.	R.P.M.	I.A.S.	M.P.H.	M.P.	G.P.H.	ALT.
690	770	590	660	620	630	680	710	550	620	580	590	740	830	600	670	630	640	790	880	650	720	680	690	840	930	700	770	730	740
640	710	550	620	580	590	640	670	510	580	540	550	700	790	560	630	590	600	750	840	610	680	640	650	800	890	660	730	690	700
580	640	500	560	520	530	580	610	470	540	500	510	660	750	480	550	510	520	670	760	520	590	550	560	710	800	620	690	650	660
520	580	450	500	460	470	520	550	430	480	440	450	600	690	440	510	470	480	610	700	480	550	510	520	650	740	580	650	610	620
450	510	390	440	400	410	450	480	370	420	380	390	530	620	400	470	430	440	540	630	460	530	490	500	580	670	540	610	570	580
390	440	340	380	340	350	390	420	320	360	320	330	460	550	380	450	410	420	470	560	420	490	450	460	540	630	490	560	520	530
330	370	290	320	280	290	330	360	270	300	270	280	400	490	360	430	390	400	450	540	400	470	430	440	520	610	470	540	500	510
270	310	240	260	220	230	270	300	220	240	210	220	340	430	320	390	350	360	410	500	380	450	410	420	500	590	450	520	480	490
210	240	180	200	160	170	210	240	180	200	170	180	280	370	280	350	310	320	370	460	340	410	370	380	460	550	430	500	460	470
150	170	130	150	110	120	150	180	120	140	110	120	220	310	240	290	250	260	310	400	300	370	330	340	420	510	390	460	420	430
90	100	80	90	70	80	90	100	70	80	60	70	160	250	180	210	190	200	250	340	260	330	290	300	380	470	350	420	380	390

NOTES: ALLOW 60 GAL FOR WARM-UP, TAKE-OFF & INITIAL CLIMB PLUS ALLOWANCE FOR WIND, RESERVE & COMBAT AS REQD.

EXAMPLE: AT 19,400 LB. GROSS WT. WITH 740 GAL OF FUEL (AFTER DEDUCTING TOTAL ALLOWANCES OF 60 GAL) TO FLY 1500 STAT. AIRMILES AT 12,000 FT. ALT. MAINTAIN 1900 RPM AND 212 MPH IND. AIRSPEED WITH MIXTURE SET A.L.

LEGEND: I.A.S.: INDICATED AIRSPEED M.P.H.: MANIFOLD PRESSURE G.P.H.: U.S. GAL PER HOUR T.A.S.: TRUE AIRSPEED S.L.: SEA LEVEL

RED FIGURES ARE PRELIMINARY. SUBJECT TO REVISION AFTER FLIGHT CHECK

EXTERNAL LOAD ITEMS

FLIGHT OPERATION INSTRUCTION CHART

MODEL(S)

P-38 L

TWO 300 GALLON TANKS

ENGINE(S): V-1710-III 8 V-1710-III 3

CHART WEIGHT LIMITS: 21,400 TO 14,500 POUNDS

LIMITS	B. P. M.	M. P. (IN. HG.)	BLOWER POSITION	MIXTURE POSITION	TIME LIMIT	TOTAL G. P. H.
WAR MAX.	3000	60		AR	5	360
MILITARY POWER	3000	54		AR	15	334
NORMAL RATED	2500	44		AR	NONE	226

INSTRUCTIONS FOR USING CHART: Select figure in FUEL column equal to or less than amount of fuel to be used for cruising. Move horizontally to left or right and select RANGE value equal to or greater than the statute or nautical air miles to be flown. Vertically below and opposite desired cruising altitude (ALT.) read optimum R. P. M., I. A. S. and MIXTURE setting required.

NOTES: Column I is for emergency high speed cruising only. Columns II, III, IV and V give progressive increase in range at a sacrifice in speed. Manifold pressure (M. P.), gallons per hour (G. P. H.) and true airspeed (T. A. S.) are approximate values for reference. For efficiency maintain indicated airspeed (I. A. S.) hourly. Adjust RPM slightly if necessary to avoid exceeding manifold pressure more than 3 in. Hg.

I							II				III				IV				V			
RANGE IN AIR MILES							RANGE IN AIR MILES				RANGE IN AIR MILES				RANGE IN AIR MILES				RANGE IN AIR MILES			
STATUTE							STATUTE				STATUTE				STATUTE				STATUTE			
NAUTICAL							NAUTICAL				NAUTICAL				NAUTICAL				NAUTICAL			
U. S. GAL.							U. S. GAL.				U. S. GAL.				U. S. GAL.				U. S. GAL.			
1010							1080				1350				1870				1630			
950							1050				1310				1810				1570			
920																						
830							950				1180				1640				1420			
740							840				1050				1460				1270			
650							740				920				1280				1110			
560							640				800				1100				960			
470							540				670				930				800			
380							430				540				750				650			
290							330				410				570				500			
200							230				280				390				340			
110							130				160				220				190			

Use Column 2

MAXIMUM CONTINUOUS										OPERATING DATA										OPERATING DATA										MAXIMUM RANGE																													
I. A. S. M.P.H.					M. P. In. Hg.					T. A. S. M.P.H.					R. P. M.					I. A. S. M.P.H.					M. P. In. Hg.					T. A. S. M.P.H.					R. P. M.					I. A. S. M.P.H.					M. P. In. Hg.					T. A. S. M.P.H.									
ALT. Feet					ALT. Feet					ALT. Feet					ALT. Feet					ALT. Feet					ALT. Feet					ALT. Feet					ALT. Feet					ALT. Feet					ALT. Feet					ALT. Feet									
30000					2550					2550					2550					2550					2550					2550					2550					2550					2550					2550					2550				
25000					2500					2500					2500					2500					2500					2500					2500					2500					2500					2500					2500				
20000					2000					2000					2000					2000					2000					2000					2000					2000					2000					2000					2000				
15000					1500					1500					1500					1500					1500					1500					1500					1500					1500					1500					1500				
12000					1200					1200					1200					1200					1200					1200					1200					1200					1200					1200					1200				
9000					900					900					900					900					900					900					900					900					900					900					900				
6000					600					600					600					600					600					600					600					600					600					600					600				
3000					300					300					300					300					300					300					300					300					300					300					300				
S. L.					S. L.					S. L.					S. L.					S. L.					S. L.					S. L.					S. L.					S. L.					S. L.					S. L.					S. L.				

Max. Permissible IAS with 300 gallon tanks-210 MPH.

LEGEND

I. A. S.: INDICATED AIRSPEED
M. P.: MANIFOLD PRESSURE
G. P. H.: GALLONS PER HOUR
T. A. S.: TRUE AIRSPEED
S. L.: SEA LEVEL

EXAMPLE

AT 21,400 LB. GROSS WT. WITH 1010 GAL. OF FUEL
(AFTER DEDUCTING TOTAL ALLOWANCES OF 60 GAL.)
TO FLY 1860 STAT. AIRMILES AT 12,000 FT. ALT.
MAINTAIN 2050 RPM AND 208 MPH IND. AIRSPEED
WITH MIXTURE SET A. L.

NOTES

○ ALLOW 60 GAL. FOR WARM-UP, TAKE-OFF & INITIAL CLIMB
PLUS ALLOWANCE FOR WIND, RESERVE & COMBAT AS REQ'D.

F. T.: FULL THROTTLE
P. R.: FULL RICH
M. P.: MANIFOLD PRESSURE
G. P. H.: GALLONS PER HOUR
T. A. S.: TRUE AIRSPEED
S. L.: SEA LEVEL

RED FIGURES ARE PRELIMINARY.
SUBJECT TO REVISION AFTER FLIGHT CHECK